TECHNICAL MANUAL

ORGANIZATIONAL MAINTENANCE SYSTEM SCHEMATICS

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

NAVY MODEL F/A-18A AND F/A-18B 161353 AND UP

N68936-01-D-0007

<u>DISTRIBUTION STATEMENT C.</u> Distribution authorized to U.S. Government agencies and their contractors to protect publications required for official use or for administrative or operational purposes only, determined on 15 October 1987. Other requests for this document shall be referred to Commanding Officer, Naval Air Technical Data and Engineering Service Command, Naval Air Station North Island, P.O. Box 357031, Building 90 Distribution, San Diego, CA 92135-7031.

<u>DESTRUCTION NOTICE</u>. For unclassified, limited documents, destroy by any method that will prevent disclosure of contents or reconstruction of the document.

PUBLISHED BY DIRECTION OF THE COMMANDER, NAVAL AIR SYSTEMS COMMAND

0801LP1018031

Change 8 - 1 June 2002

NUMERICAL INDEX OF EFFECTIVE WORK PACKAGES/PAGES

List of Current Changes

Original 0 Mar 85	Change 3 15 Apr 87	Change 5 1 Oct 88	Change 7 1 Dec 00
Change 1 1 May 86	Change 4 15 Oct 87	Change 6 15 Aug 92	Change 8 1 Jun 02
Change 2 1 Sep 86	_		_

Only those work packages/pages assigned to the manual are listed in this index. Insert Change 8, dated 1 June 2002. Dispose of superseded work packages/pages. Superseded classified work packages/pages shall be destroyed in accordance with applicable security regulations. If changed pages are issued to a work package, insert the changed pages in the applicable work package. The portion of text affected in a change or revision is indicated by change bars or the change symbol "R" in the outer margin of each column of text. Changes to illustrations are indicated by pointing hands, change bars, or MAJOR CHANGE symbols. Changes to diagrams may be indicated by shaded borders.

WP Number	Title	WP Number	Title
Title		016 00	Schematic-Landing Gear and
Page A	Numerical Index of Effective Work		Related Systems Interface
	Packages/Pages	016 01	Schematic-Landing Gear and
TPDR-1	List of Technical Publications		Related Systems Interface (After
	Deficiency Reports		F/A-18 AFC 253 or F/A-18 AFC
001 00	Alphabetical Index		292)
002 00	Introduction	017 00	Schematic-Electrical and Hydraulic
003 00	Component Locator		Systems Interface
004 00	Schematic-Interconnect	017 01	Schematic-Electrical and Hydraulic
004 01	Schematic-Mission Data Loader		Systems Interface (After F/A-18
	(After F/A-18 AFC 253 or F/A-18		AFC 253 or F/A-18 AFC 292)
	AFC 292)	018 00	Schematic-Secondary Power
005 00	Schematic-Power		System Interface
006 00	Schematic-Fluids Test	018 01	Schematic-Secondary Power
009 00	Schematic-Left Engine Interface		System Interface (After F/A-18
009 01	Schematic-Left Engine Interface		AFC 253 or F/A-18 AFC 292)
	(After F/A-18 AFC 253 or F/A-18	019 00	Schematic-Canopy, Wingfold;
	AFC 292)		Boarding Ladder; Pitot Static, Gun,
010 00	Schematic-Right Engine Interface		Anti-Icing and Air Induction
010 01	Schematic-Right Engine Interface		Systems Interface
	(After F/A-18 AFC 253 or F/A-18	019 01	Schematic-Canopy, Wingfold;
	AFC 292)		Boarding Ladder; Pitot Static, Gun,
011 00	Schematic-Fuel System Interface		Anti-Icing and Air Induction
011 01	Schematic-Fuel System Interface		Systems Interface (After F/A-18
	(After F/A-18 AFC 253 or F/A-18		AFC 253 or F/A-18 AFC 292)
	AFC 292)	020 00	Schematic-Maintenance Code Clear
012 00	Schematic-Built-In Test		and Inhibit
012 01	Schematic-Built-In Test (After	021 00	Schematic-Mission Data Loader
	F/A-18 AFC 253 or F/A-18 AFC		Mission Initialization Functional
	292)		(After F/A-18 AFC 253 or F/A-18
013 00	Schematic-Fatigue Strain Data		AFC 292)
014 00	Schematic-Record Function	022 00	Schematic-Mission Data Loader
015 00	Schematic-Environmental Control		Built-In Test (After F/A-18 AFC
	Systems Interface		253 or F/A-18 AFC 292)
015 01	Schematic-Environmental Control		
	Systems Interface (After F/A-18		
	AFC 253 or F/A-18 AFC 292)		

A1-F18AC-580-500

Change 8 - 1 June 2002

Total number of pages in this manual is 276 consisting of the following:

WP/Page Change	WP/Page Change	WP/Page Change	WP/Page Change
Number Number	Number Number	Number Number	Number Number
Title 8 A - B 8 C Blank 8 TPDR-1 8 TPDR-2 Blank 8 001 00 1 7 2 Blank 7 002 00 1 - 7 8 8 Blank 8 003 00 1 - 3 7 4 6 5 7 6 2 7 - 8 0 9 1 10 - 21 0 22 1 10 - 21 0 22 1 23 6 24 0 25 6 24 0 25 6 24 0 25 6 24 0 25 6 24 0 25 6 24 0 25 6 26 7 27 6 28 0 29 7 30 Blank 7 004 00 1 - 2 7 3 - 5 5 6 7 7 - 9 5 10 7	13	012 00 1 - 7 7 8 Blank 7 012 01 1 - 7 7 8 Blank 7 013 00 1 - 4 2 Blank 4 013 01 1 - 7 4 8 Blank 4 013 02 1 - 5 4 6 Blank 4 013 03 1 - 6 4 014 00 1 - 5 6 6 Blank 6 015 00 1 - 9 7 10 Blank 7 015 01 1 - 9 7 10 Blank 7 016 00 1 - 6 7 016 01 1 - 6 7	017 00 1 - 8

Page B/(C blank)

TPDR-1

Change 8 - 1 June 2002 (TPDR-2 blank)

LIST OF TECHNICAL PUBLICATION DEFICIENCY REPORTS INCORPORATED ORGANIZATIONAL MAINTENANCE SYSTEM SCHEMATICS

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes TPDR WP, dated 1 December 2000.

1. The TPDRs listed below have been incorporated in this issue.

IDENTIFICATION NUMBER/ QA SEQUENCE NUMBER	LOCATION
NO	ONE

Page 1/(2 blank)

ALPHABETICAL INDEX

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes WP001 00, dated 1 October 1988.

Title	WP Number
Component Locator	003 00
Introduction	002 00
Content	002 00
Digital Display Indicators	002 00
Effectivities	002 00
Manual Issue Date	002 00
Record of Applicable Technical Directives	002 00
Schematic Highlights	002 00
Technical Directives	002 00
Technical Publications Deficiency Report (TPDR)	002 00
Schematic -	
Built-In Test	012 00
Built-In Test (After F/A-18 AFC 253 or F/A-18 AFC 292)	012 01
Canopy, Wingfold; Boarding Ladder; Pitot Static, Gun, Anti-Icing and Air	
Induction Systems Interface	019 00
Canopy, Wingfold; Boarding Ladder; Pitot Static, Gun, Anti-Icing and Air	
Induction Systems Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	019 01
Electrical and Hydraulic Systems Interface	017 00
Electrical and Hydraulic Systems Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	017 01
Environmental Control Systems Interface	015 00
Environmental Control Systems Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	015 01
Fatigue Strain Data	013 00
Fluids Test	006 00
Fuel System Interface	011 00
Fuel System Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	011 01
Interconnect	004 00
Landing Gear and Related Systems Interface	016 00
Landing Gear and Related Systems Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	016 01
Left Engine Interface	009 00
Left Engine Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	009 01
Maintenance Code Clear and Inhibit	020 00
Mission Data Loader (After F/A-18 AFC 253 or F/A-18 AFC 292)	004 01
Mission Data Loader Built-In Test (After F/A-18 AFC 253 or F/A-18 AFC 292)	022 00
Mission Data Loader Mission Initialization Functional (After F/A-18 AFC 253	
or F/A-18 AFC 292)	021 00
Power	005 00
Record Function	014 00
Right Engine Interface	010 00
Right Engine Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	010 01
Secondary Power System Interface	018 00
Secondary Power System Interface (After F/A-18 AFC 253 or F/A-18 AFC 292)	018 01

INTRODUCTION

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes WP002 00, dated 1 October 1988.

1. PURPOSE.

2. This manual has system schematics to give information about the system and allow signal tracing through the system. The system schematics support on-aircraft maintenance of mechanical, pneudraulic, electrical, and electronic functions. These functions are integrated on the schematics for ease of trouble-shooting a complete system.

3. CONTENT.

- 4. Each system is supported by schematics and a component locator.
- 5. COMPONENT LOCATOR. The component locator shows aircraft location, nomenclature and reference designation number of each system component. The illustration shows the technicians view when possible.
- 6. SCHEMATICS. Simplified schematics, and detailed schematics provide direct support for testing and troubleshooting. All schematics are shown with electrical power off, switches in off position, and relays in deenergized position unless noted on schematic.
- 7. Simplified Schematics. Simplified schematics consist primarily of blocks connected by single lines with limited use of symbols and pictorial drawings of units. These schematics simplify system functions as much as possible.
- 8. **Detailed Schematics.** Detailed schematics integrate applicable electrical, pneudraulic and mechanical functions of the system. Detailed schematics show component location, connector pin letters and numbers, in line connectors, test points, and

enough data to trace signals through the components within the system. Operational information next to components provides more data as required.

9 SCHEMATIC HIGHLIGHTS.

10. For schematic highlights see figure 1.

11 MANUAL ISSUE DATE.

12. The date on the title page is the copy freeze date. No additions, deletions, or changes are made after the manual issue date except last minute safety of flight or required maintenance changes. Data collected after the manual issue date will be included in later changes or revisions of the manual.

13. EFFECTIVITIES.

14. Effectivity notes on manual title pages, work package title pages, and within a work package indicate the aircraft or software program to which the data applies. If no effectivity note appears on the work package title page, the work package has the same effectivity as shown on the manual title page. The effectivity notes may use:

NOTE

Aircraft with model designator F/A-18B are the same type and model as TF/A-18A.

- a. Type, model, and series
- b. Bureau number (tail number)
- c. Combination of type, model, series, and bureau numbers

d. Part number or serial number

f. Configuration/identification number

e. Technical directive number

15. The table below shows examples of effectivity notes and their meanings:

Effectivity Note Examples

Effectivity Note	Definition	
160777 AND UP	Applicable to all F/A-18A, F/A-18B, F/A-18C and F/A-18D for bureau numbers listed.	
F/A-18A, F/A-18B	Applicable to all F/A-18A and F/A-18B.	
F/A-18C, F/A-18D	Applicable to all F/A-18C and F/A-18D.	
F/A-18A	Applicable to all F/A-18A, but not F/A-18B, F/A-18C and F/A-18D.	
F/A-18B	Applicable to all F/A-18B, but not F/A-18A, F/A-18C, and F/A-18D.	
F/A-18C	Applicable to all F/A-18C, but not F/A-18A, F/A-18B, and F/A-18D.	
F/A-18D	Applicable to all F/A-18D, but not F/A-18A, F/A-18B, and F/A-18C.	
F/A-18A, F/A-18C	Applicable to all F/A-18A and F/A-18C, but not to F/A-18B and F/A-18D.	
F/A-18B, F/A-18D	Applicable to all F/A-18B and F/A-18D, but not to F/A-18A and F/A-18C.	
F/A-18A 160775, 160777 THRU 160782	Only applicable to some bureau numbers of F/A-18A. Not applicable to any F/A-18B, even if an F/A-18B bureau number is within the numbers listed.	
F/A-18C 163427, 163430 THRU 163456	Only applicable to some bureau numbers of F/A-18C. Not applicable to any F/A-18D, even if an F/A-18D bureau number is within the numbers listed.	
F/A-18B 160784 AND UP	Only applicable to some bureau numbers of F/A-18B. Not applicable to any F/A-18A, even if an F/A-18A bureau number is within the numbers listed.	
F/A-18D 163434 THRU 163457	Only applicable to some bureau numbers of F/A-18D. Not applicable to any F/A-18C, even if an F/A-18C bureau number is within the numbers listed.	
160775 THRU 160785 BEFORE F18 AFC 772	Applicable to F/A-18A and F/A-18B for bureau numbers listed, before modification by technical directive.	
161213 AND UP, ALSO 160775 THRU 160785 AFTER F18 AFC 772	Applicable to aircraft modified during production; also applicable when affected aircraft have been modified by technical directive.	
160775 THRU 160785; WHEN NO. 2 CONTROL PANEL P/N XXXX-X IS INSTALLED	Applicable to F/A-18A and F/A-18B for bureau numbers listed if panel P/N XXXX-X is installed.(Configuration before AVC)	

Effectivity Note Examples (Continued)

Effectivity Note	Definition
161213 AND UP; ALSO 160775 THRU 160785; WHEN NO. 2 CONTROL PANEL P/N XXXX-Y (AVC-102) IS INSTALLED	Applicable to aircraft modified during production; also applicable to aircraft components modified to the production configuration by technical directive. (Configuration after AVC)
P/N MBEU65101-9, MBEU65101-10 & MBEU65105-3	Applicable to assemblies which are interchangeable between aircraft.
ENGINE No. 215101 THRU 215109	Applicable to assemblies which are interchangeable between aircraft, but configurations can not be identified by part number.
CONFIG/IDENT NUMBER 84A	The CONFIG/IDENT Number is the program load identification number which identifies the software program loaded in specific programmable units. Refer to A1-F18AC-SCM-000 for CONFIG/IDENT Number tables.

16 TECHNICAL DIRECTIVES.

- 17. Technical directives are documents which direct the accomplishment, and recording of a retrofit configuration or inspection to delivered aircraft, or aircraft components.
- 18. AIRFRAME (AFC) OR SOFTWARE CONFIGURATION (ASC) CHANGES. AFC and ASC effectivities are written the same, except only the AFTER configuration of an ASC is shown in a manual. See AFC effectivity example in Effectivity Note Example table.
- 19. AIRCRAFT COMPONENT CHANGES. Technical directives which change configuration of aircraft components, i.e. AAC, ACC, AVC, AYC, and PPC will list part numbers in the effectivities. See AVC effectivity examples in Effectivity Note Example table.

20. RECORD OF APPLICABLE TECHNICAL DIRECTIVES.

21. The technical directives affecting this manual are listed in the Record of Applicable Technical Directives of each affected work package. Because an ASC directs all aircraft be modified within 30 days, ASC's are not listed. When all affected aircraft are modified, the before configuration is removed from the manual, and the technical directive entry is removed from the Record of Applicable Technical Directives.

22. TECHNICAL PUBLICATIONS DEFI-CIENCY REPORT (TPDR).

23. The TPDR (OPNAV FORM 4790/36) is the form for reporting errors and suspected omissions in the technical manuals. Reporting procedures are in OPNAVINST 4790.2 SERIES.

24 DIGITAL DISPLAY INDICATORS.

25. Digital Display Indicators IP-1317() used on aircrafts 161353 THRU 163782 and Digital Display Indicators IP-1556/A used on aircrafts 163895 AND UP may be referred to as Digital Display Indicators. Full Navy (AN) standard nomenclature will be used in the Illustrated Parts Breakdown (IPB).

26. REQUISITION AND AUTOMATIC DISTRIBUTION OF NAVAIR TECHNICAL MANUALS.

- Procedures to be used by Naval activities and other Department of Defense activities requiring NAVAIR technical manuals are defined in NAVAIR 00-25-100 and NAVAIRINST 5605.5A.
- 28. To automatically receive future changes and revisions to NAVAIR technical manuals, an activity must be established on the Automatic Distribution Requirements List (ADRL) maintained by the Naval Air Technical Data and Engineering Service Command (NATEC). To become established on the ADRL, notify your activity central technical publications librarian. If your activity does not have a library, you may establish your automatic distribution by contacting the Commanding Officer, NA-

A1-F18AC-580-500 Change 8

102 00 Page 4

TEC, Attn: Distribution, NAS North Island, Bldg. 90, P.O. Box 357031, San Diego, CA 92135-7031. Annual reconfirmation of these requirements are necessary to remain on automatic distribution. Please use your NATEC assigned account number whenever referring to automatic distribution requirements.

29. If additional or replacement copies of this manual are required with no attendant changes in the ADRL, they may be ordered by submitting a

MILSTRIP requisition in accordance with NAVSUP 485 to Routing Identifier Code "NFZ". MILSTRIP requisitions can be submitted through your supply office, Navy message, or SALTS to DAAS (Defense Automated Address System), or through the DAAS or NAVSUP web sites. For assistance with a MILSTRIP requisition, contact the Naval Inventory Control Point (NAVICP) Publications and Forms Customer Service at DSN 442-2626 or (215) 697-2626, Monday through Friday, 0700 to 1600 Eastern Time.

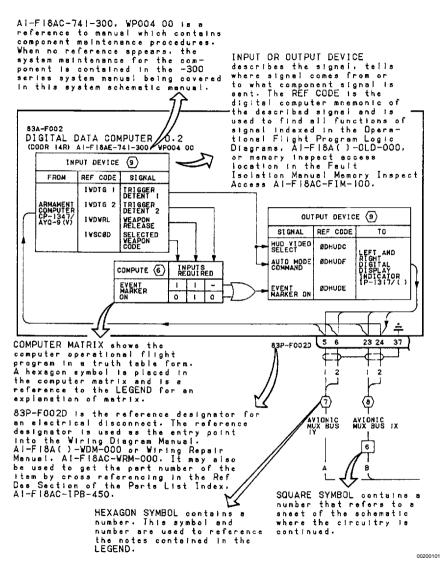


Figure 1. Schematic Highlights (Sheet 1)

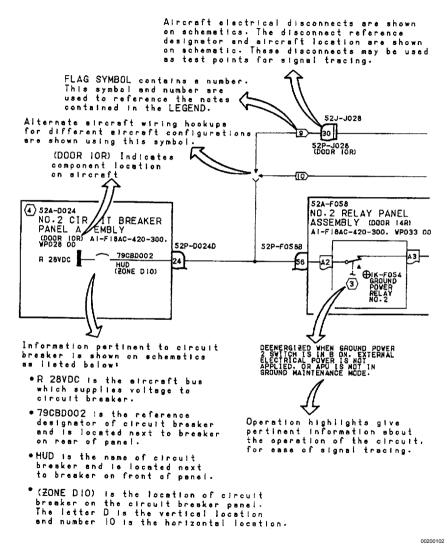


Figure 1. Schematic Highlights (Sheet 2)

Page 7/(8 blank)

The legend contains all notes pertinent to the schematic as listed below:

- NUMBER listed with no symbol is general information about the schematic.
- NONSTANDARD SYMBOLS appearing on schematic are shown or referenced with an explanation.
- ABBREVIATIONS appearing on schematic are shown or referenced with an explanation.
- HEXAGON SYMBOL refers to another schematic or manual for continuation of a circuit or an explanation of data contained on schematic.
- FLAG SYMBOL indicates limited aircraft application.



1. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON THE SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 (2) SHORTS BETYEEN SURROUNDING PINS ON CONNECTORS.
 (3) SHORTS BETYEEN SHIELD AND CONDUCTORS.
 (4) SHIELD CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
 - TIPE TIPE S RELAY USED TO SWITCH TO LOW LEVEL CURRENT. SEE NOTE 1.
- (3) GROUND POWER SWITCHING SCHEMATIC. AI-F18AC-420-500. WP005 00-
- 4) POWER DISTRIBUTION SCHEMATIC. A1-F18AC-420-500. WP004 00.
- S EXPLANATION OF MATRIX:
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
 - C. THE SIGNAL OUTPUT IS READ HORIZONTALLY, EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED:
- (6) AVIONIC MUX CHANNEL I SCHEMATIC. AI-FIBAC-741-500. WPOOB 00.
- AVIONIC MUX CHANNEL 2 SCHEMATIC. AI-FIBAC-741-500, WPOC3 OD-
- FOR LOGIC DIAGRAMS RELATING TO REF CODE. REFER TO A1-F18AC-CLD-COO. REMORY HISPECT ACCESS LOCATION RELATING TO REF CODE. REFER TO A1-F18AC-FIM-100.
- 9)F/A-184.
- 10)F/A-18B.

00200103

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

COMPONENT LOCATOR

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

Reference Material

None

Alphabetical Index

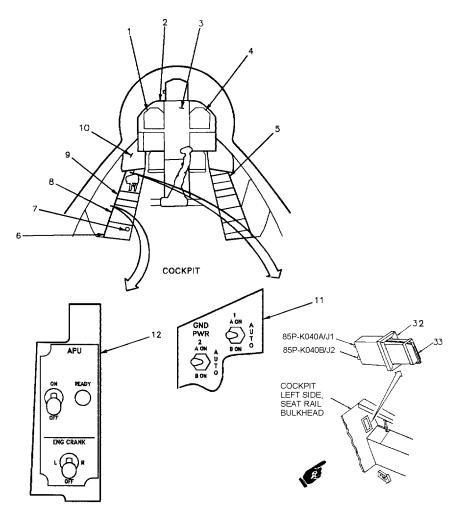
Subject	Page No.
Component Locator, Figure 1	3

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 IAFC 056	27 Mar 85	Fuel System Components Replacement and System Inspection (ECP MDA-F18-00158R1 and ECP MDA-F/A-18-00160)	1 May 86	
F/A-18 AFC 27	13 Jul 90	Leading Edge Flap/Control Stick Changes, Incorporation of (ECP MDA-F/A-18-00044C2)	1 May 86	
F/A-18 AFC 49	28 Feb 90	Sealed Lead Acid Battery, Addition of (ECP MDA-F/A-18-00074)	1 Sep 86	
F/A-18 AFC 48	28 Feb 90	Automatic AC Bus Isolation, Incorporation of (ECP MDA-F/A-18-00121R1)	1 Sep 86	

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 70	31 Dec 89	Motive Flow Fuel Boost Pump Pressure Switch Installation of (ECP MDA-F/A-18-00158R2)	15 Oct 87	
F/A-18 AFC 90	2 Feb 90	GFE Battery Relay Control Unit, Incorporation of (ECP MDA-F/A-18-00165R1)	1 Oct 88	
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-



00300101

Figure 1. Component Locator (Sheet 1)

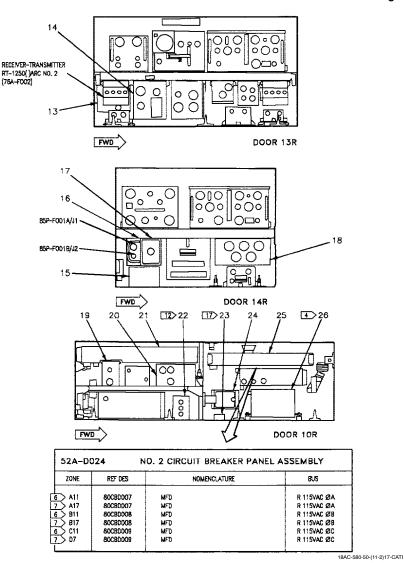
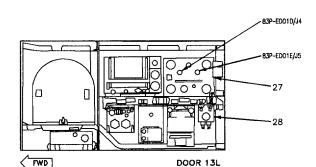
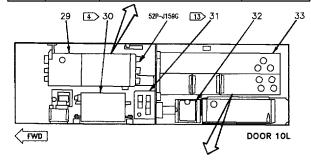


Figure 1. Component Locator (Sheet 2)



ZONE	REF DES	NOMENCLATURE	BUS
D2	85CBCD04	MSDRS	4 MAINT 24/28VC 5 U BATT/MAINT 24/28VDC
D12	80CBC006	MMD	L 115VAC ØC
E12	80CBC005	MMD	L 115VAC ØB
F12	80CBC004	MMD	L 115VAC ØA

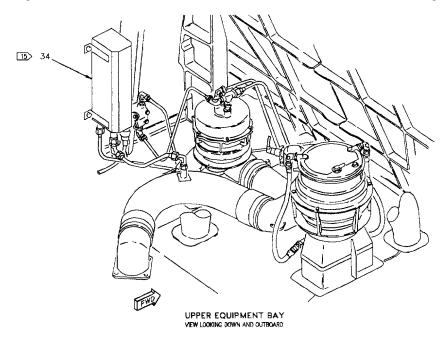


52A-C057 NO. 7 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY					
ZONE	REF DES	NOMENCLATURE	BUS		
A20	83CBCC06	MISSION COMP NO 1	L 115VAC ØA		
B20	83CBC007	MISSION COMP NO 1	L 115VAC Ø		
C20	83CBCCOB	MISSION COMP NO 1	L 115VAC Ø0		

00300103

Figure 1. Component Locator (Sheet 3)

Change 2



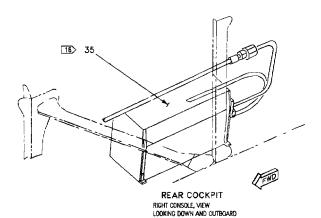


Figure 1. Component Locator (Sheet 4)

18AC-580-50-(11-4)17-SCAN

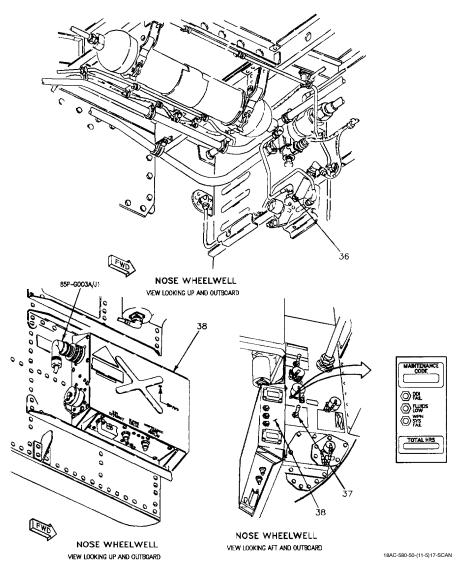


Figure 1. Component Locator (Sheet 5)

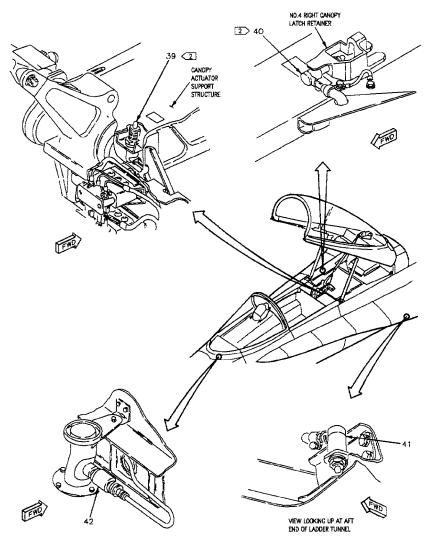


Figure 1. Component Locator (Sheet 6)

18AC-580-50-(11-6)17-SCAN

Change 1

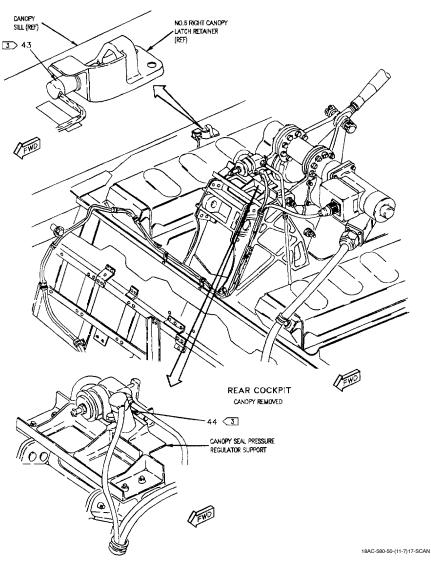


Figure 1. Component Locator (Sheet 7)

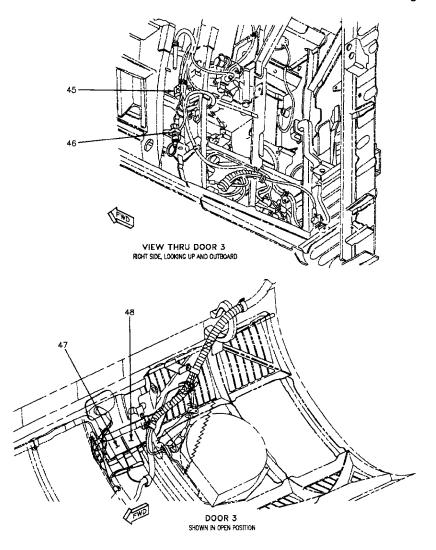
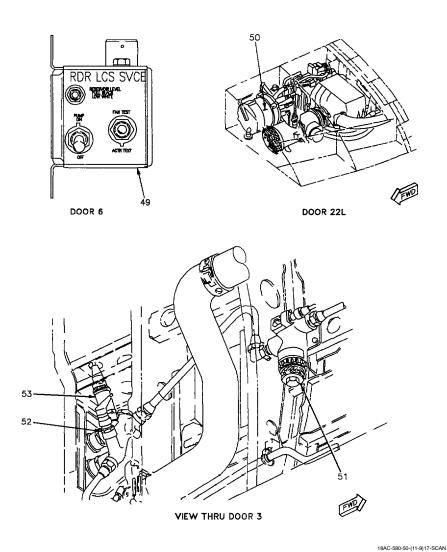


Figure 1. Component Locator (Sheet 8)

18AC-580-50-(11-8)17-CATI



10A0-300-30-(11-8)17

Figure 1. Component Locator (Sheet 9)

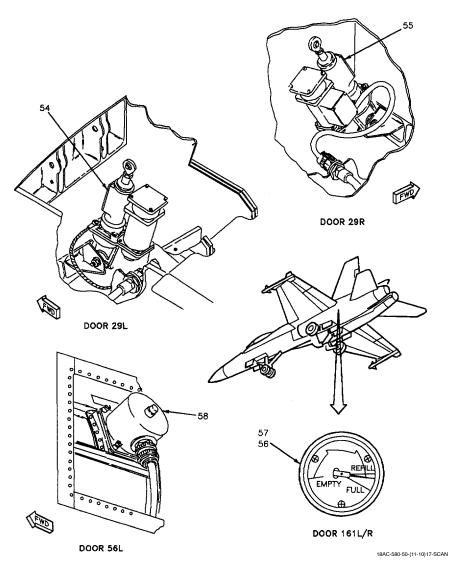
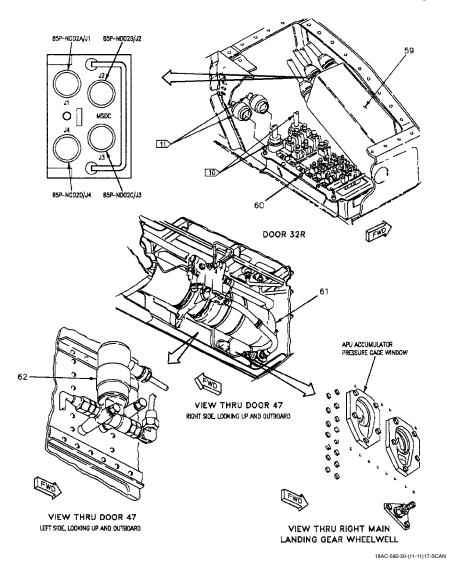


Figure 1. Component Locator (Sheet 10)



. . . .

Figure 1. Component Locator (Sheet 11)

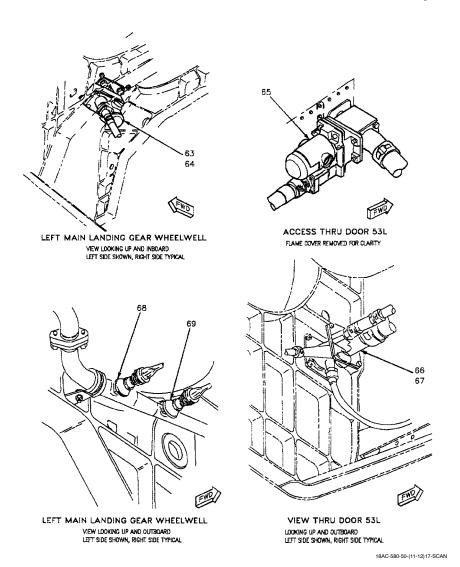


Figure 1. Component Locator (Sheet 12)

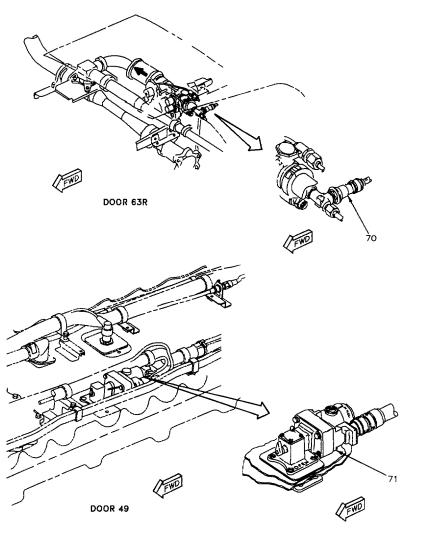
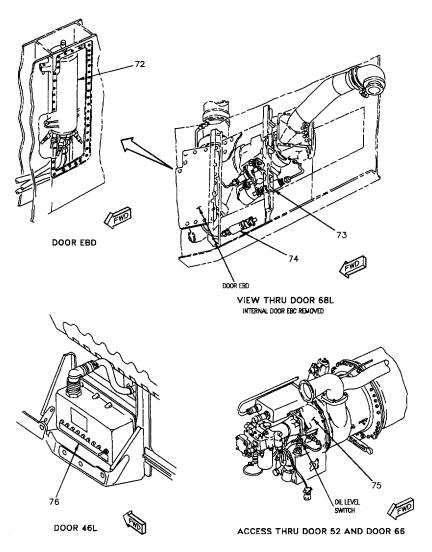


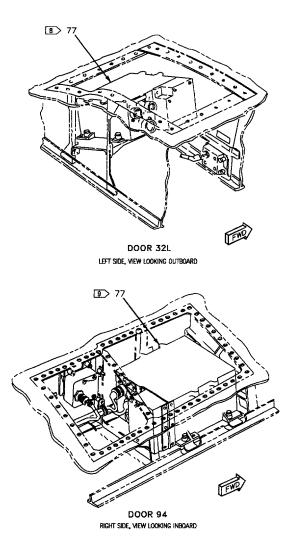
Figure 1. Component Locator (Sheet 13)

18AC-580-50-(11-13)17-SCAN



18AC-580-50-(11-14)17-SCAN

Figure 1. Component Locator (Sheet 14)



18AC-580-50-(11-15)17-SCAN

Figure 1. Component Locator (Sheet 15)

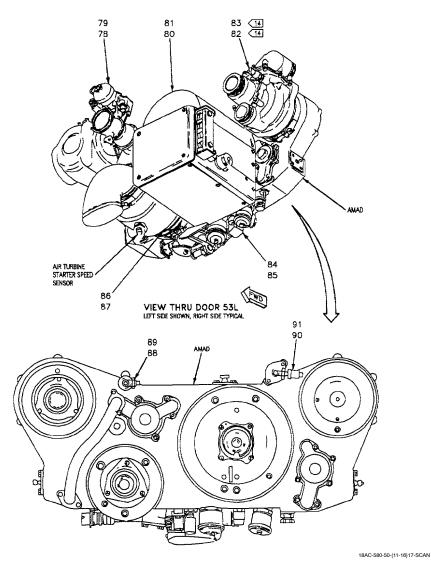


Figure 1. Component Locator (Sheet 16)

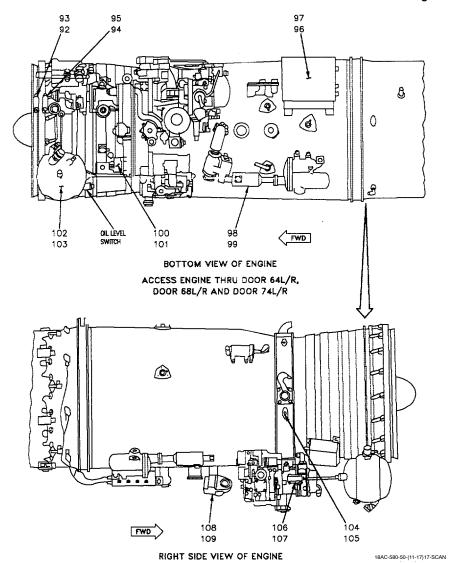


Figure 1. Component Locator (Sheet 17)

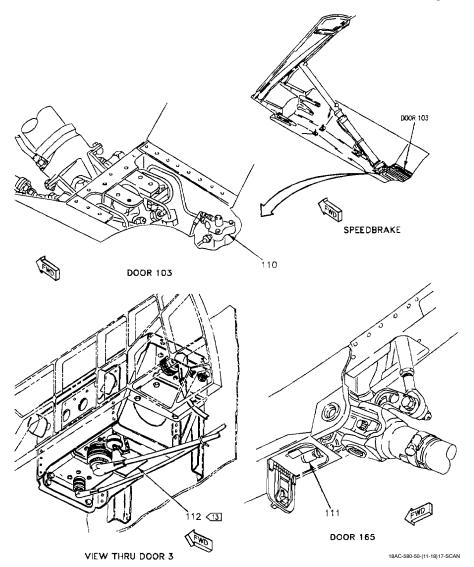
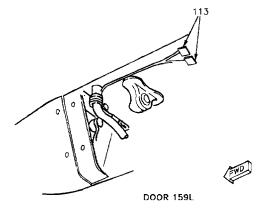


Figure 1. Component Locator (Sheet 18)



LEFT WING TIP COMMAND SIGNAL ENCODER-DECODER KY-851/AYQ-9(V) REMOVED FOR CLARITY

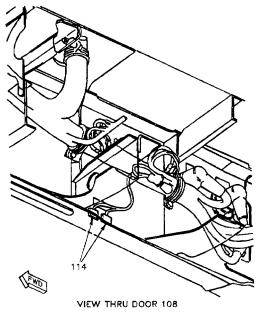
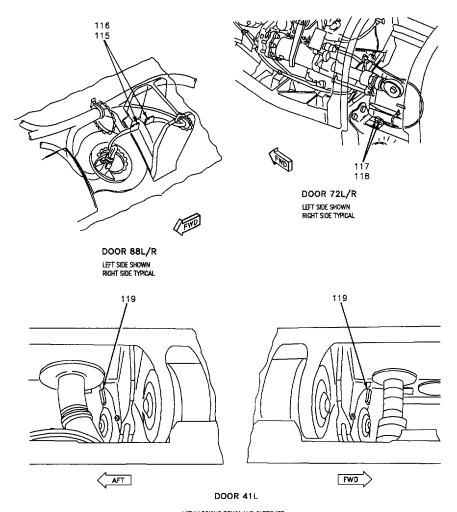


Figure 1. Component Locator (Sheet 19)

18AC-580-50-(11-19)17-SCAN



VIEW LOOKING DOWN AND OUTBOARD

18AC-580-50-(11-20)17-SCAN

Figure 1. Component Locator (Sheet 20)

NOMENCLATURE	INDEX NO.	REF DES
ACS TEMPERATURE FLOW CONTROLLER	20	22A-D002
AIR PRESSURE SWITCH	70	5S-T106
APU	75	2A-P015
APU ACCUMULATOR AND START VALVE ASSEMBLY	61	2L-P011
APU CONTROL PANEL APU CONTROL SWITCH ENG CRANK SWITCH	12	52A-H079 2S-H003 3S-H003
APU FUEL SHUTOFF VALVE	62	2L-P012
ARRESTING HOOK UP SWITCH	111	19S-S006
17 BATTERY RELAY CONTROL UNIT	23	1A-D155
BLEED AIR LEAK DETECTION WARNING SYSTEM CONTROL UNIT	76	24A-P011
BOARDING LADDER STOWED SWITCH	41	20S-M009
15 CABIN EXIT AIR REGULATOR CONTROLLER	34	22A-K170
16 CABIN EXIT AIR REGULATOR CONTROLLER	35	22A-L170
CANOPY LOCKED SWITCH	44	3 20S-E007
CANOPY LOCKED SWITCH	39	2 20S-L007
CANOPY POSITION SWITCH	43	3 20S-F008
CANOPY POSITION SWITCH	40	2 20S-L008
CONTROL CONVERTER C-10382A	14	82A-F001
DIGITAL DATA COMPUTER NO. 1	27	83A-E001
DIGITAL DISPLAY INDICATOR ID-2150/ASM-612	38	85A-G003
DRAG BRACE SUPPORT STRAIN GAGE	114	85M-F019
ECS PANEL ASSEMBLY	5	52A-J078

Figure 1. Component Locator (Sheet 21)

NOMENCLATURE	INDEX NO.	REF DES
ELECTRICAL BORESIGHT COMPENSATION ASSEMBLY	13	85A-F007
ELECTRONIC CONTROL UNIT	77	8 2A-M010 9 2A-N010
ELECTRONIC EQUIPMENT CONTROL C-10380/ASQ	3	79A-J006
4 EMERGENCY BATTERY AND CHARGER UNIT	30	1A-C072
EXTERNAL FUEL SYSTEM AIR PRESSURE SWITCH	69	5A-P151
EXTERNAL FUEL SYSTEM AIR PRESSURE SWITCH	68	5A-P152
FIRE EXTINGUISHER TANK	72	4SQT109
FLUID LEVEL INDICATOR AND FLEXIBLE CABLE	57	10M-P010
FLUID LEVEL INDICATOR AND FLEXIBLE CABLE	56	10M-R011
FUEL CROSSFEED SHUTOFF VALVE	65	5B-P071
FUEL DUMP VALVE	71	5B-P069
FUEL FEED LINE TEMPERATURE SENSOR (LH)	67	5A-P111
FUEL FEED LINE TEMPERATURE SENSOR (RH)	66	5A-R112
FUEL QUANTITY GAGING INTERMEDIATE DEVICE	15	5A-F014
FUEL SYSTEM CONTROL PANEL	9	5A-H027
GND PWR CONTROL PANEL ASSEMBLY	11	1A-H004
GUN GAS PURGE PRESSURE SWITCH 1	46	61S-B184
GUN GAS PURGE PRESSURE SWITCH 2	45	61S-B185
INLET ICE DETECTOR	58	9A-P005
INTERCOMMUNICATION AMPLIFIER-CONTROL AM-6979/A OR AM-7360/A	8	76A-H009

Figure 1. Component Locator (Sheet 22)

	NOMENCLATURE	INDEX NO.	REF DES
	LANDING GEAR CONTROL UNIT	12 22 13 112	12 12A-D004 13 12A-A004
	LEFT AIR TURBINE STARTER	86	3MAP521
	LEFT AIR TURBINE STARTER CONTROL VALVE	79	3L-P006
	LEFT AMAD LEFT AMAD OIL LEVEL SWITCH LEFT AMAD OIL PRESSURE SWITCH LEFT AMAD OIL TEMPERATURE THERMOSTAT LEFT CURRENT FLOW SENSOR	84 90 88	3MAP515 3S-P059 3S-P055 3S-P064
	LEFT DIGITAL DISPLAY INDICATOR IP-1317()	1	80A-H001
	LEFT ENGINE ALTERNATOR ANTI ICING VALVE COMPRESSOR DISCHARGE PRESSURE	100 108 106	3MAS551 3G-S607 3VAS641
	TRANSMITTER ELECTRICAL CONTROL ASSEMBLY FUEL FLOW TRANSMITTER OIL PRESSURE TRANSMITTER OIL TANK TURBINE DISCHARGE PRESSURE TRANSMITTER VIBRATION ACCELEROMETER	96 98 94 102 92	3TRS685 3Z-S605 3TRS647 3TRS649 3CAS595 3TRS651
	LEFT ENGINE FUEL SHUTOFF VALVE	63	5B-P072
14	LEFT FUEL BOOST PRESSURE SWITCH	82	5S-P113
	LEFT GENERATOR CONVERTER UNIT	81	1A-P001
	LEFT HORIZONTAL STABILATOR STRAIN GAGE	117	85M-S013
	LEFT INLET BLEED AIR DOOR ACTUATOR	54	3B-M028
	LEFT POWER CONTACTOR	32	1K-C007
	LEFT VERTICAL STABILIZER STRAIN GAGE	115	85M-S011
	LEFT WING FOLD STRAIN GAGE	113	85M-U021
		l	

Figure 1. Component Locator (Sheet 23)

	NOMENCLATURE	INDEX NO.	REF DES
	LEFT WING ROOT STRAIN GAGE	119	85M-U020
	LH ADVISORY AND THREAT WARNING INDICATOR PANEL	2	52A-H073
	LH VERTICAL CONSOLE CONTROL PANEL	10	52A-H077
	LIQUID OXYGEN QUANTITY INDICATOR GMU-751A	7	15M-H002
	MAGNETIC TAPE CARTRIDGE MX-9972/ASM-612	17	85A-F501
18	MISSION DATA LOADER AN/ASQ-215	12B	85A-K503
18	MISSION DATA LOADER CP-2092(P)/A	12A	85A-K040
17	MMP ENABLE/BRCU SWITCH	37	1S-G160
	NO. 2 CIRCUIT BREAKER PANEL ASSEMBLY	25	52A-D024
	NO. 4 CIRCUIT BREAKER PANEL ASSEMBLY	21	52A-D026
	NO. 7 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	33	52A-C057
	NO. 8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	29	52A-C159
	NO. 2 RELAY PANEL ASSEMBLY	18	52A-F058
	NO. 3 RELAY PANEL ASSEMBLY	28	52A-E059
	NO. 4 RELAY PANEL ASSEMBLY	60	52A-N118
13	NO. 9 RELAY PANEL ASSEMBLY	31	52A-C161
	PILOT SERVICES CONTROL PANEL ASSEMBLY	6	52A-H083
	PRIMARY BLEED AIR OVERPRESSURE SWITCH	73	22S-S019
	RADAR LIQUID COOLING CENTRIFUGAL PUMP UNIT	50	22B-M086
	RADAR LIQUID COOLING FLUID PRESSURE FILTER	51	22FLA089
	RADAR LIQUID COOLING HIGH TEMPERATURE WARNING SENSOR	52	22A-A088
	RADAR LIQUID COOLING LOW PRESSURE SENSOR	53	22A-A087
	Figure 1 Component Locate	or (Chaot 24)	'

Figure 1. Component Locator (Sheet 24)

NOMENCLATURE	INDEX NO.	REF DES
RDR LCS SVCE PANEL ASSEMBLY	49	22A-A090
RIGHT AIR TURBINE STARTER	87	3MAR522
RIGHT AIR TURBINE STARTER CONTROL VALVE	78	3L-R007
RIGHT AMAD RIGHT AMAD OIL LEVEL SWITCH RIGHT AMAD OIL PRESSURE SWITCH RIGHT AMAD OIL TEMPERATURE THERMOSTAT	85 91 89	3MAR516 3S-R060 3S-R056 3S-R065
RIGHT CURRENT FLOW SENSOR	47	28A-B016
RIGHT DIGITAL DISPLAY INDICATOR IP-1317()	4	80A-J002
RIGHT ENGINE ALTERNATOR ANTI ICING VALVE COMPRESSOR DISCHARGE PRESSURE TRANSMITTER ELECTRICAL CONTROL ASSEMBLY FUEL FLOW TRANSMITTER OIL PRESSURE TRANSMITTER OIL TANK TURBINE DISCHARGE PRESSURE TRANSMITTER VIBRATION ACCELEROMETER	101 109 107 97 99 95 103 93	3MAT552 3G-T608 3VAT642 3TRT686 3Z-T606 3TRT648 3TRT650 3CAT596 3TRT652 3TRT690
RIGHT ENGINE FUEL SHUTOFF VALVE	64	5B-R070
14 RIGHT FUEL BOOST PRESSURE SWITCH	83	5S-R114
RIGHT GENERATOR CONVERTER UNIT	80	1A-R002
RIGHT HORIZONTAL STABILATOR STRAIN GAGE	118	85M-T012
RIGHT INLET BLEED AIR DOOR ACTUATOR	55	3B-N033
RIGHT POWER CONTACTOR	24	1K-D008
RIGHT VERTICAL STABILIZER STRAIN GAGE	116	85M-T010
SECONDARY BLEED AIR OVERPRESSURE SWITCH	74	22S-S018
SIGNAL DATA CONVERTER CV-3493/ASM-612	59	85A-N002

Figure 1. Component Locator (Sheet 23)

NOMENCLATURE	INDEX NO.	REF DES
SIGNAL DATA RECORDER RO-508/ASM-612	16	85A-F001
SKID CONTROL BOX ASSEMBLY	19	13A-D003
TEMPERATURE COMPENSATED PRESSURE SWITCH	110	19S-T012
TEMPERATURE COMPENSATED PRESSURE SWITCH	36	10S-G009
UTILITY BATTERY AND CHARGER UNIT	26	1A-D035
4 WINDSHIELD OVERHEAT TEMPERATURE SENSOR	42	23A-B003
LEGEND		
1. AIRCRAFT CONNECTOR LOCATIONS ARE SHOWN IN A	1-F18A()-WDM-0	000.
2 F/A-18A.		
3 F/A- 18B.		
4 161353 THRU 161528 BEFORE F/A-18 AFC 49.		
5 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A	-18 AFC 49.	
6 161353 THRU 161359.		
7 161360 AND UP.		
8 161353 THRU 161519 BEFORE F/A-18 AFC 27.		
9 161520 AND UP; ALSO 161353 THRU 161519 AFTER F/A	-18 AFC 27.	
10 161353 THRU 161727.		
11 161728 AND UP.		
12 161353 THRU 161987 BEFORE F/A-18 AFC 48.		
13 162394 AND UP; ALSO 161353 THRU 161987 AFTER F/A	-18 AFC 48.	
163119 AND UP; ALSO 161353 THRU 161924 BEFORE F/A-18 IAFC 056, OR 161353 THRU 163118 AFTER F/A-18 AFC 70.		
15 F/A-18A 163092 AND UP.		

Figure 1. Component Locator (Sheet 24)

A1-F18AC-580-500

Change 7

003 00 Page 29/(30 blank)

NOMENCLATURE	INDEX NO.	REF DES
16 F/A-18B 163104 AND UP.		
163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.		
18 F/A-18A 162394 THRU 163175 AFTER F/A-18 AFC 253 O	R F/A-18 AFC 292	

Figure 1. Component Locator (Sheet 27)

Page No.

3

Subject

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - INTERCONNECT

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

Reference Material

None

Alphabetical Index

Interconnect Schematic, Figure 1

Record of Applicable Technical Directives					
Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks	
F/A-18 IAFC 056	27 Mar 85	Fuel System Components Replacement and System Inspection (ECP MDA-F/A-18- 00158R1 and ECP MDA-F/A-18-00160)	15 Oct 83		
F/A-18 AFC	-	Air Turbine Starter/Airframe Mounted	1 Mar 85	ECP	

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 IAFC 056	27 Mar 85	Fuel System Components Replacement and System Inspection (ECP MDA-F/A-18- 00158R1 and ECP MDA-F/A-18-00160)	15 Oct 83	
F/A-18 AFC 26	-	Air Turbine Starter/Airframe Mounted Accessory Drive Design Changes (ECP MDA-F/A-18-00068)	1 Mar 85	ECP coverage only
F/A-18 AFC 27	-	Improvement of Leading Edge Flap Design (ECP MDA-F/A-18-00044)	1 May 86	ECP coverage only

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 49	-	Addition of Sealed Lead Acid Battery (ECP MDA- F/A-18-00074)	1 Sep 86	ECP coverage only
F/A-18 AFC 48	-	Alternating Current Bus Isolation (ECP MDA- F/A-18-00121)	1 Sep 86	ECP coverage only
F/A-1B AFC 39	-	No 1 Fuel Tank Interconnect Valve Replacement and Fuel Sequencing Modification (ECP MDA-F/A-18-00072C1)	15 Apr 87	ECP coverage only
F/A-18 AFC 70	-	Motive Flow Fuel Boost Pump Pressure Switch Installation of (ECP MDA- F/A-18-00158R2 and ECP MDA- F/A-18-00160)	15 Oct 87	ECP coverage only
F/A-18 AFC 90	-	Incorporation of GFE Battery Relay Control Unit (ECP MDA-F/A-18-00165R1)	1 Oct 88	ECP coverage only
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

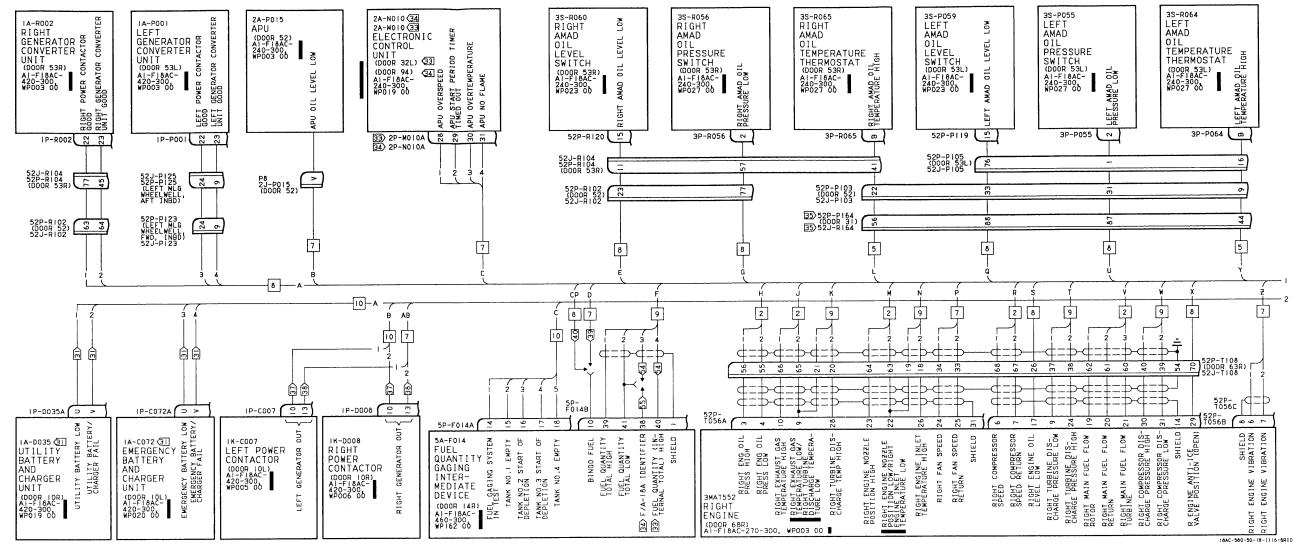


Figure 1. Figure 1. Interconnect Schematic (Sheet 1)

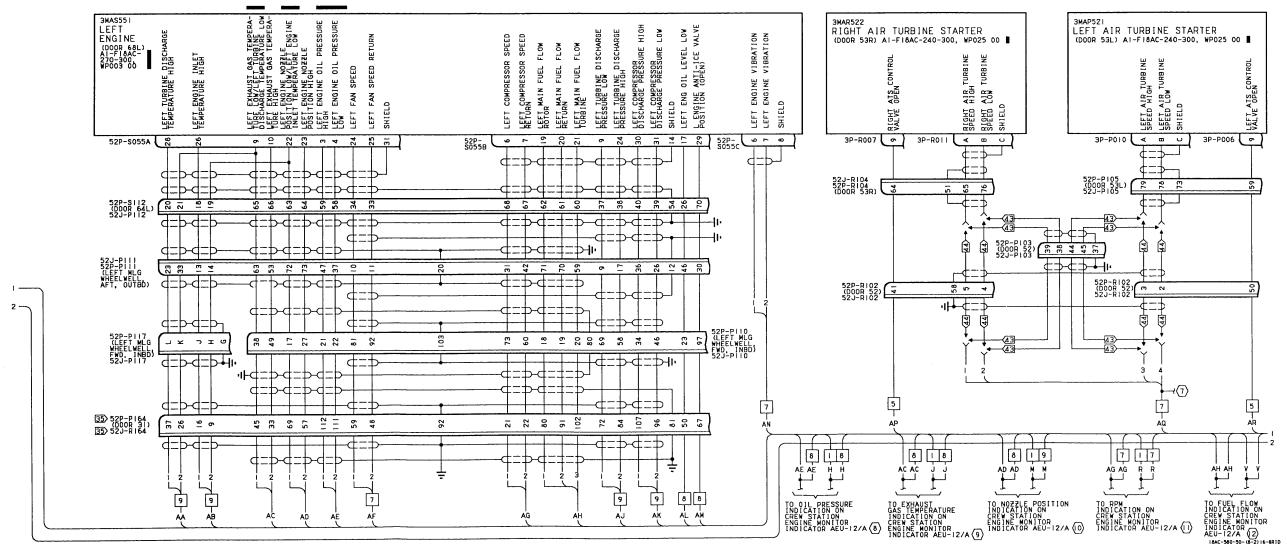


Figure 1. Figure 1. Interconnect Schematic (Sheet 2)

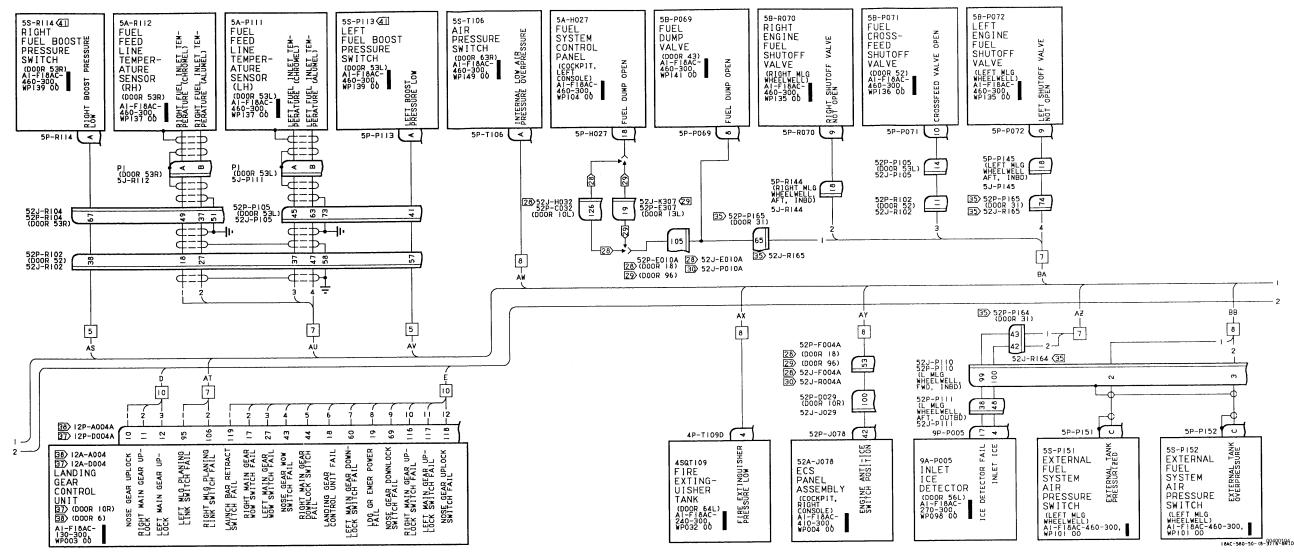


Figure 1.

Figure 1. Interconnect Schematic (Sheet 3)

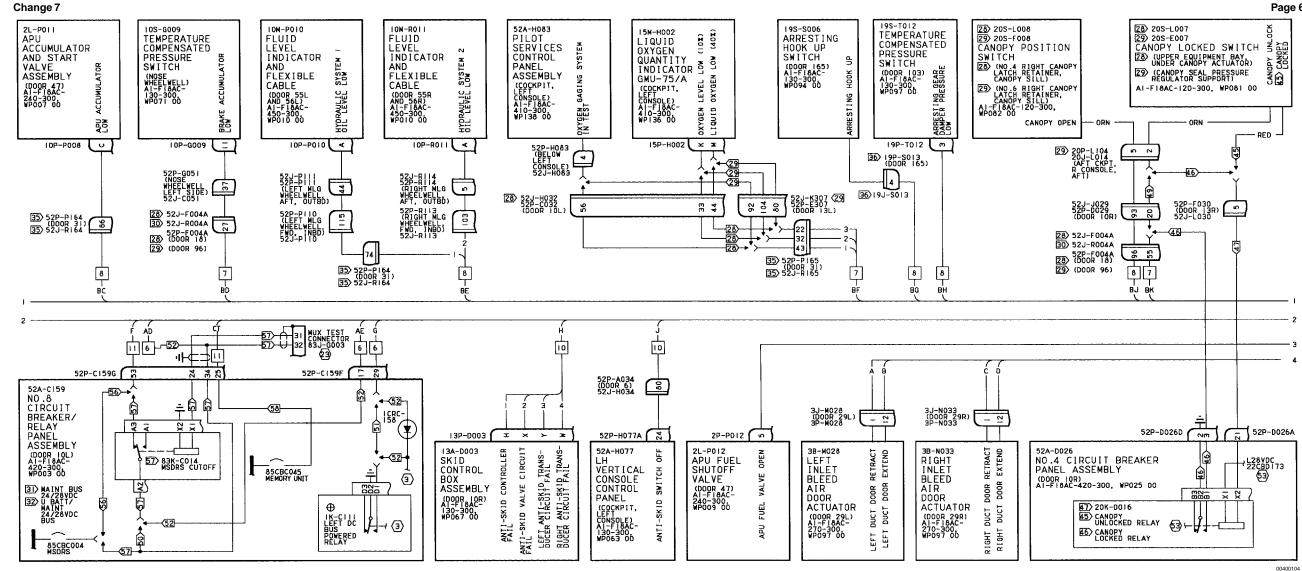


Figure 1. Interconnect Schematic (Sheet 4)

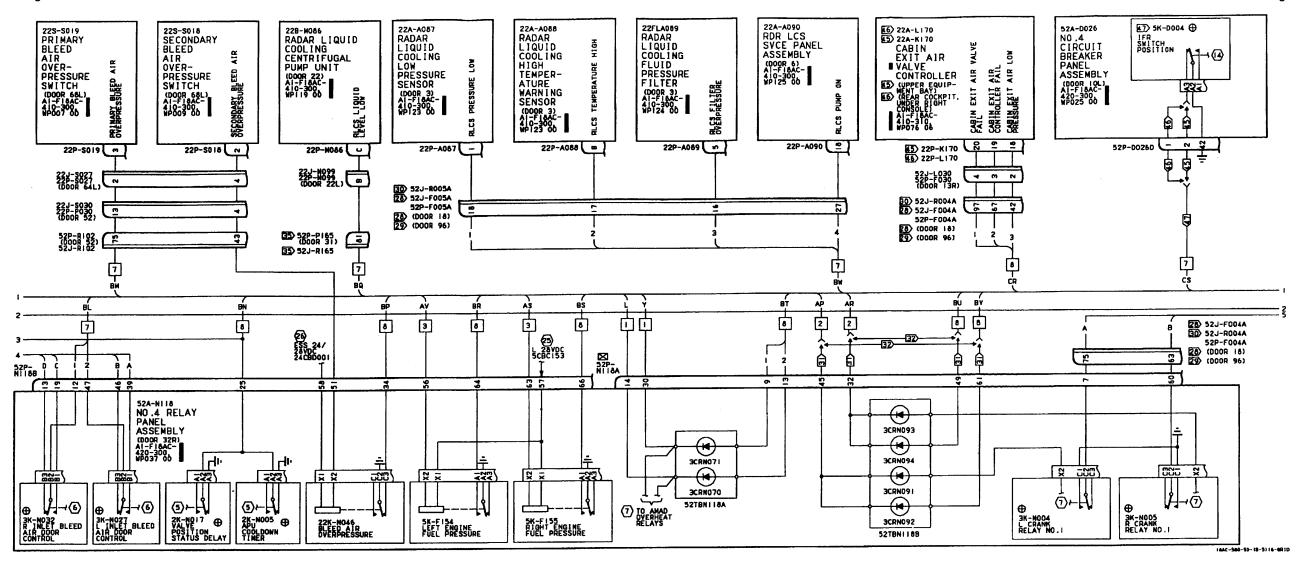


Figure 1.

Figure 1. Interconnect Schematic (Sheet 5)

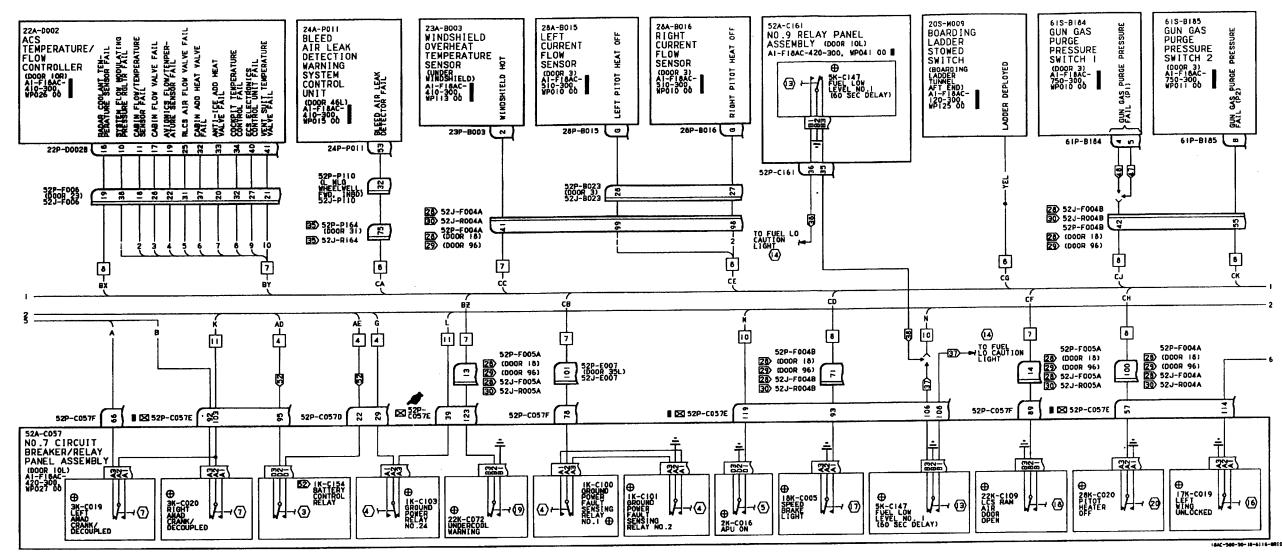


Figure 1.

Figure 1. Interconnect Schematic (Sheet 6)

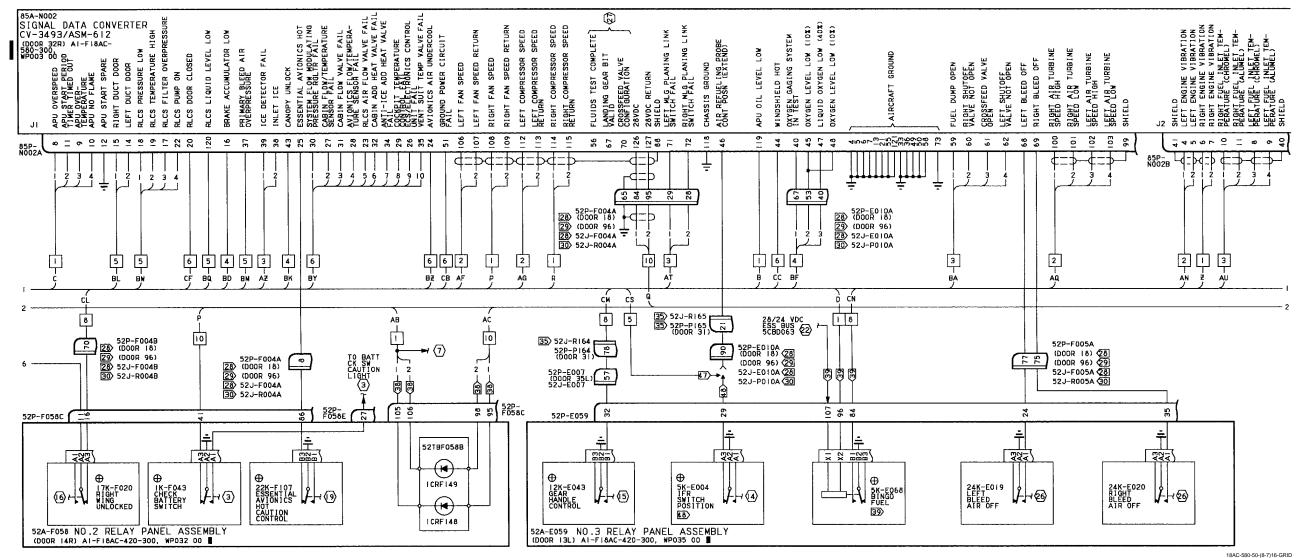


Figure 1. Interconnect Schematic (Sheet 7)

Figure 1.

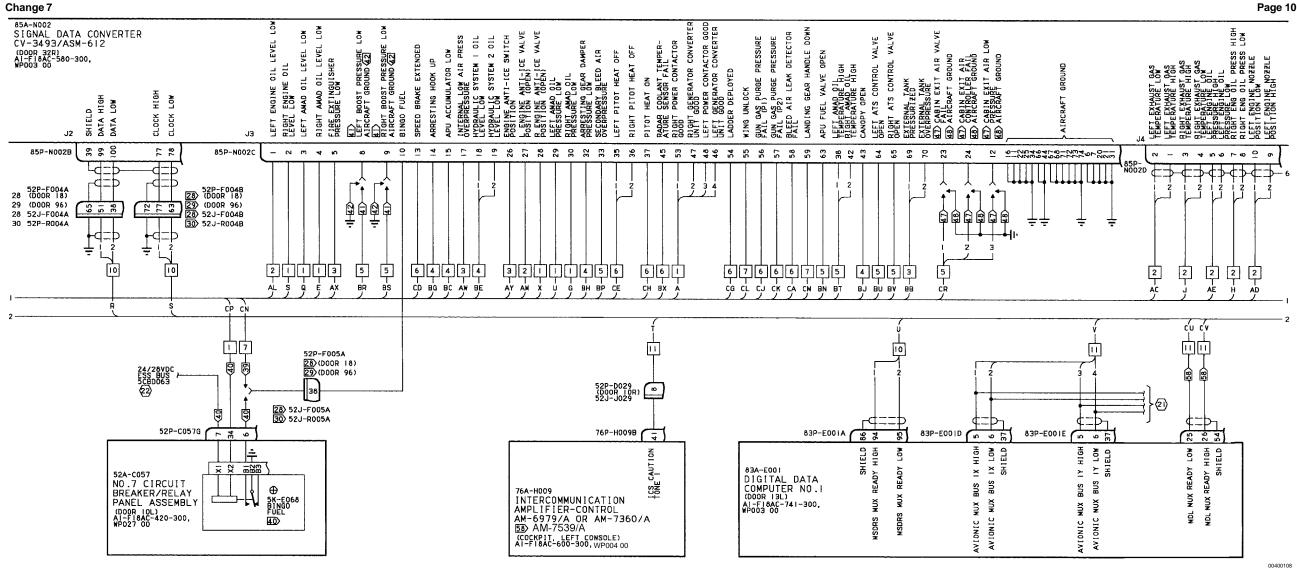
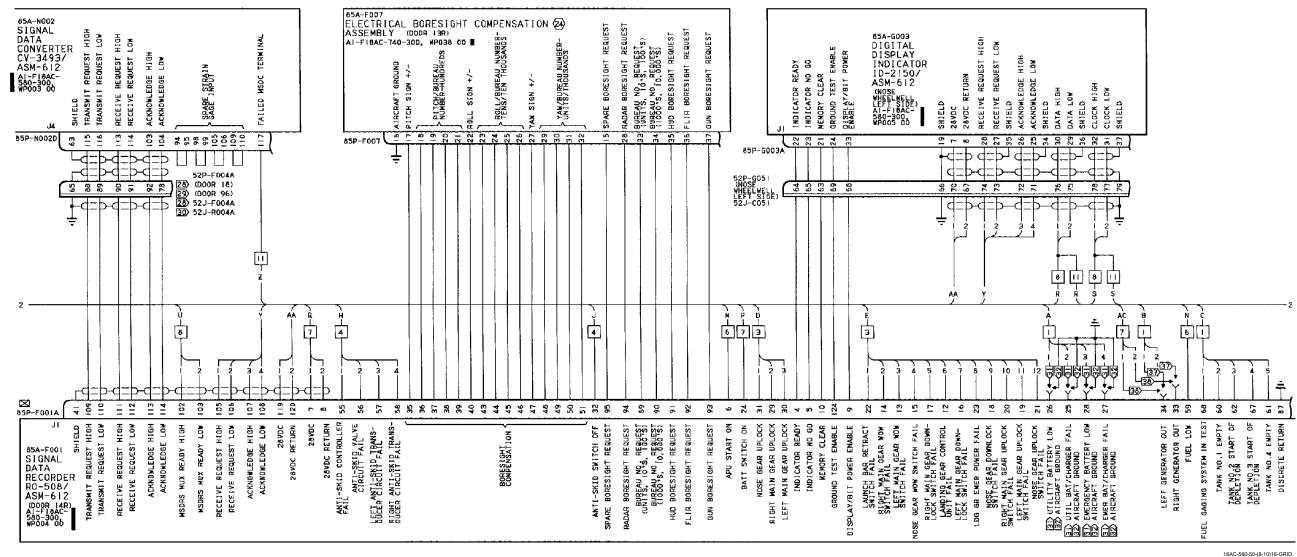


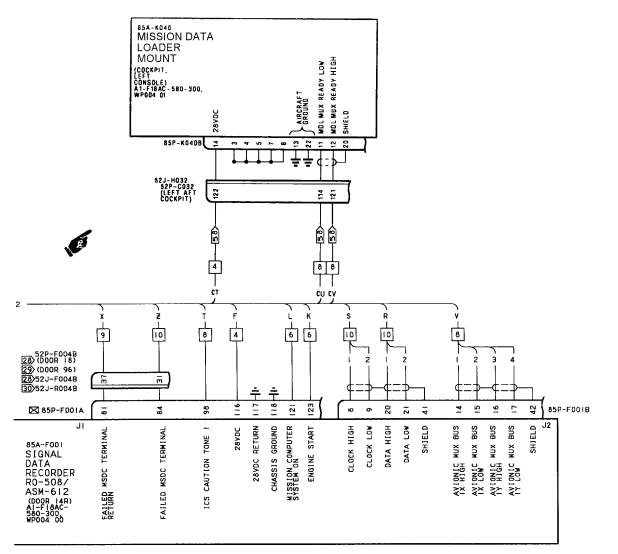
Figure 1. Interconnect Schematic (Sheet 8)

Figure 1. Interconnect Schematic (Sheet 9)



Change 7

Figure 1.



LEGEND

- I. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FISA()-WDM-000.
- WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ED) IS REMOVED FOR TROBLE-SHOOTING A IDENTIFY RELAY AND SOCKET FOR CORRECT AND SOCKET FOR LOW LOW LOW LOW LOW LAW LAY LATER AND SOCKET FOR LOW LOW LOW LOW LAW LAY IS DEFECTIVE, REPLACE WITH NEW RELAY. в.
- DONNACTS FOR CONTYPUL DEVICES (SYLICHES/RELAY CONTACTS) FOR CONTYPUL DEVICES (SYLICHES/RELAY CONTACTS FOR THAT DO NOT THE RXI SCALE INTERESTRETAY CONTACTS MAY USE С.
- D. WHEN TESTING FOR CONTINUITY, TEST FOR:

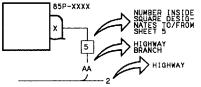
 (1) SHORTS TO GROUND

 (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS

 (3) SHORTS BETWEEN SHIELD AND CONDUCTORS

 (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF 24YDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY 🖾). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS JUMPER THIS FOR CONTINUITY.
- 2. NON STANDARD SYMBOLS.
- DENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE !.

HIGHWAY EXAMPLES



IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE

- 3) DC POWER SYSTEM SCHEMATIC, AI-FI8AC-420-500, WP004 00.
- 4 GROUND POWER SWITCHING SCHEMATIC, AI-FIBAC-420-500, WP005 00. 5 APD START SYSTEM SCHEMATIC. AI-FIBAC-240-500, WP004 00.

- 240-500, WPOUA OUR

 6 INLET BLEED AIR DOOR SYSTEM SCHEMATIC,
 AI-FIBAC-270-500, WPO09 00.

 7 ENGINE START AND GROUND MAINTENANCE
 WPO05 00.

 8 OIL PRESSURE INDICATING SYSTEM SCHEMATIC,
 AI-FIBAC-270-500, WPO07 00.
- 9 EGT INDICATING SYSTEM SCHEMATIC, AI-F18AC-270-500, WP007 00.
- (D) NOZZLE POSITION INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00.
- (I) RPM INDICATING SYSTEM SCHEMATIC, A1-F18AC-270-500, WP007 00
- (2) FUEL FLOW INDICATING SYSTEM SCHEMATIC, AI-F18AC-270-500, WP007 00.
- 13 FUEL LOW LEVEL WARNING SYSTEM
- (4) IN FLIGHT REFUELING SYSTEM SCHEMATIC, AI-FISAC-460-500, WP005 00.
- (5) LANDING GEAR CONTROLLED RELAYS SCHEMATIC, AI-F18AC-130-500. WP006 00.

- (6) WINGFOLD SYSTEM SCHEMATIC, AI-FIBAC-570-500, WP027 00.
- 7 SPEED BRAKE SYSTEM SCHEMATIC, AI-FIBAC-570-500, WP026 00.
- (8) RADAR LIQUID COOLING SYSTEM AI-FIBAC-410-500, WP014 00.
- (9) AVIONICS COOLING SYSTEM SCHEMATIC, AI-FIBAC-410-500, WP009 00.
- PITOT STATIC SYSTEM HEATER SCHEMATIC,
- AVIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- FUEL DUMP SYSTEM SCHEMATIC,
- DIGITAL DATA COMPUTER NO.1 AND NO.2 INTERCONNECT, A1-F18AC-741-500, WP008 00.
- ELECTRICAL BORESIGHT COMPENSATION SYSTEM SCHEMATIC, AI-FIBAC-740-500, WP066 00.
- ENGINE FUEL SUPPLY SYSTEM SCHEMATIC, AI-FIBAC-460-500, WP008 00.
- 26 BLEED AIR LEAK DETECTION SYSTEM SCHEMATIC, AI-F18AC-410-500, WP006 00.
- (27) INPUT ALWAYS OPEN
- 28) F/A-18A. 29) F/A-18B.
- 30 F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 31) 161353 THRU 161528 BEFORE F/A-18 AFC 49.
- 161702 AND UP, ALSO 161353 THRU 161528 AFTER F/A-18 AFC 49.
- 161353 THRU 161519 BEFORE F/A-18 AFC 27.
- 34 161520 AND UP: ALSO 161353 THRU 161519 AFTER F/A-18 AFC 27.
- 162445 AND UP
- 161522 AND UP.
- 37) 161353 THRU 161987 BEFORE F/A-18 AFC 48.
- 38 162394 AND UP, ALSO 161353 THRU 161987 AFTER F/A-18 AFC 48.
- 39 161353 THRU 161761.
- 40 161924 AND UP.
- . 163119 AND UP: ALSO 161353 THRU 161924 BEFORE F/A-18 IAFC 056, OR 161353 THRU 163118 AFTER F/A-18 AFC 70.
- 42) 161353 THRU 161924 AFTER F/A-18 IAFC 056, OR 161353 THRU 163118 BEFORE F/A-18 AFC 70.
- 43 161353 THRU 161528 BEFORE F/A-18 AFC 26.
 44 161702 AND UP: ALSO 161353 THRU 161528 AFTER F/A-18 AFC 26.
- 45) F/A-18A 163092 AND UP 46) F/A-18B 163104 AND UP
- 47> 163092 AND UP
- 161353 THRU 162909
- 49) F/A-18A; F/A-18B 161354 THRU 162885
- 161702 THRU 163118 BEFORE F/A-18 AFC 90.
- 51) 161353 THRU 163118 BEFORE F/A-18 AFC 90
- 52) 163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.
- (3) CANOPY SYSTEMS SCHEMATICS, AI-FIBAC-120-500, WP006 00 AND WP007 00.
- 54) 161353 THRU 161519 BEFORE F/A-18 AFC 39.
 55) 161520 AND UP: ALSO 161353 THRU 161519 AL
 56) 161353 THRU 161528 BEFORE F/A-18 AFC 90. 161520 AND UP: ALSO 161353 THRU 161519 AFTER F/A-18 AFC 39.
- 57 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 90
- F/A-18A 162394 THRU 163175 AFTER F/A-18 AFC 253 OR F/A-18 AFC 292.

00400111

Figure 1. Interconnect Schematic (Sheet 11)

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - MISSION DATA LOADER

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Mission Data Loader	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

Change 7 85A-K040 85A-K503 MISSION DATA LOADER (COCKPIT, LEFT SIDE, SEAT RAIL BULKHEAD) A1-F18AC-580-300, MISSION DATA LOADER MOUNT (COCKPIT, LEFT SIDE, SEAT RAIL BULKHEAD) A1-F18AC-580-300, WP007 00 WP007 00 85P-K040A J3 AVIONIC MUX CHANNEL 4 (3) 83A-E001 52P-C032 (D00R 10L) 52J-H032 DIGITAL DATA COMPUTER NO. 1 (DOOR 13L) A1-F18AC-741-300, WP003 00 85P-K040B J4 P 2 83P-E001C MOL MUX READY LOW ii 1.1 1.1 MOL MUX READY HIGH 121 52A-C159 NO. 8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY (DOOR 10L) A1-F18AC-420-300, WP030 00 \boxtimes 52P-C159G 52J-C159G $\langle 2 \rangle$ 85CBC045 L 28VDC L 28VDC 122 MEMORY UNIT (ZONÉ A7) 76A-H009 76P-H009A INTERCOMMUNICATION AMPLIFIER-CONTROL ERASE DATA (COCKPIT, LEFT CONSOLE) A1-F18AC-600-300, WP012 00

04010101

Figure 1. Mission Data Loader Interconnect Schematic (Sheet 1)

LEGEND

1. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.

- D. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \square). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.
- (3) AVIONIC MUX CHANNEL 4 SCHEMATIC, A1-F18AC-741-500, WP017 00.

Subject

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - POWER

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes WP005 00, dated 15 August 1992.

Reference Material

None

Alphabetical Index

	Subject	Page No.
Dowar Sahamatia	P' 4	

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 49	28 Feb 90	Sealed Lead Acid Battery, Addition of (ECP MDA-F/A-18-00074)	1 Sep 86	
F/A-18 AFC 90	2 Feb 90	GFE Battery Relay Control Unit, Incorporation of (ECP MDA-F/A-18-00165R1)	1 Oct 88	
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

Change 7

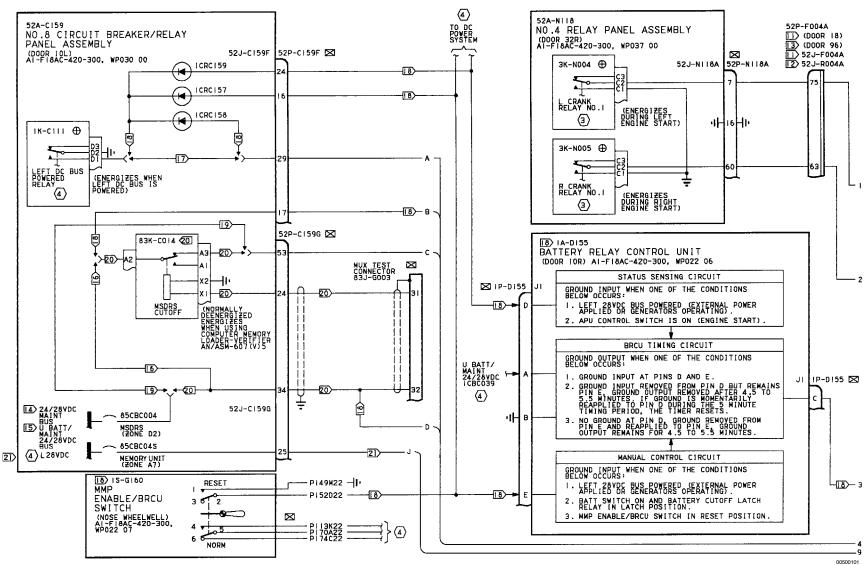


Figure 1.

Figure 1. Power Schematic (Sheet 1)

Figure 1.

Figure 1.

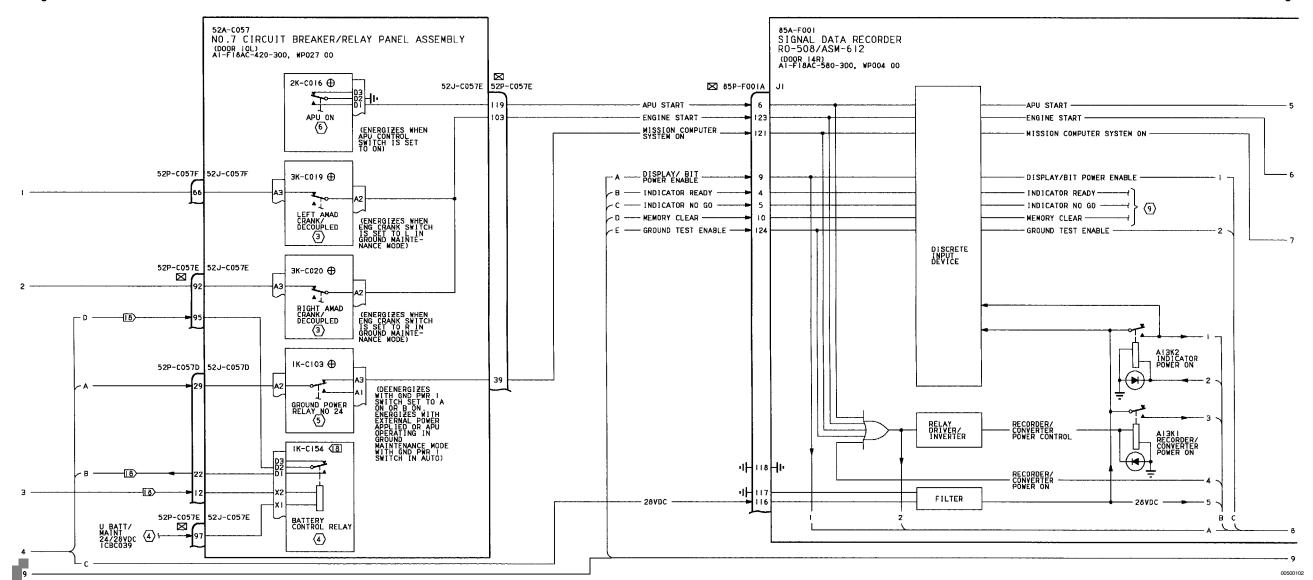


Figure 1. Power Schematic (Sheet 2)

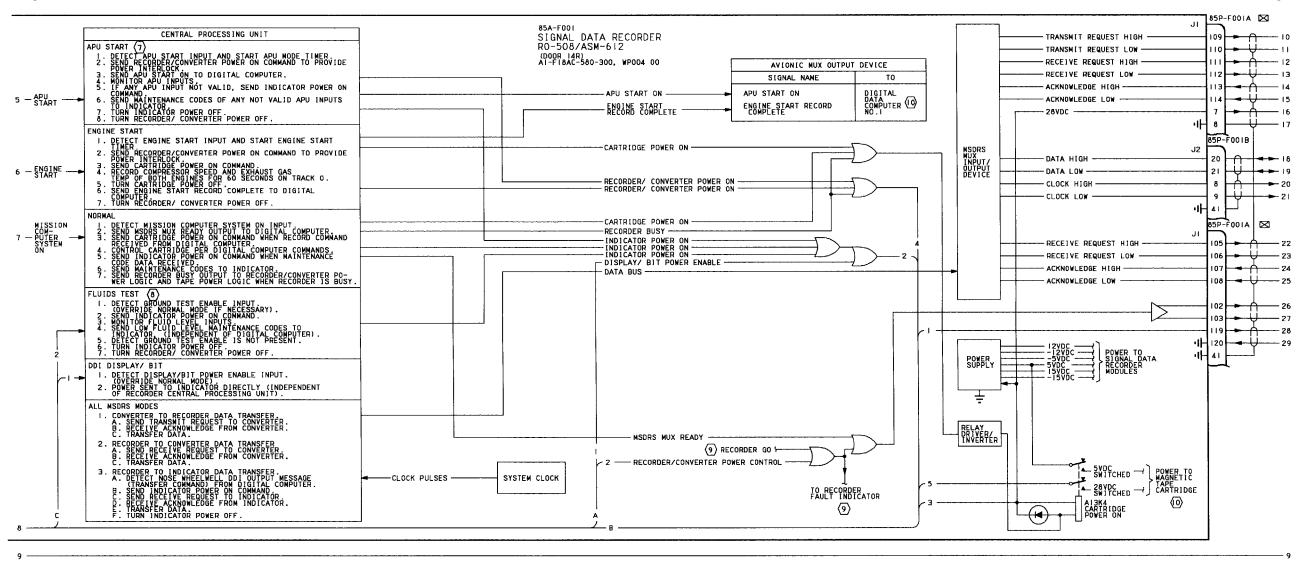
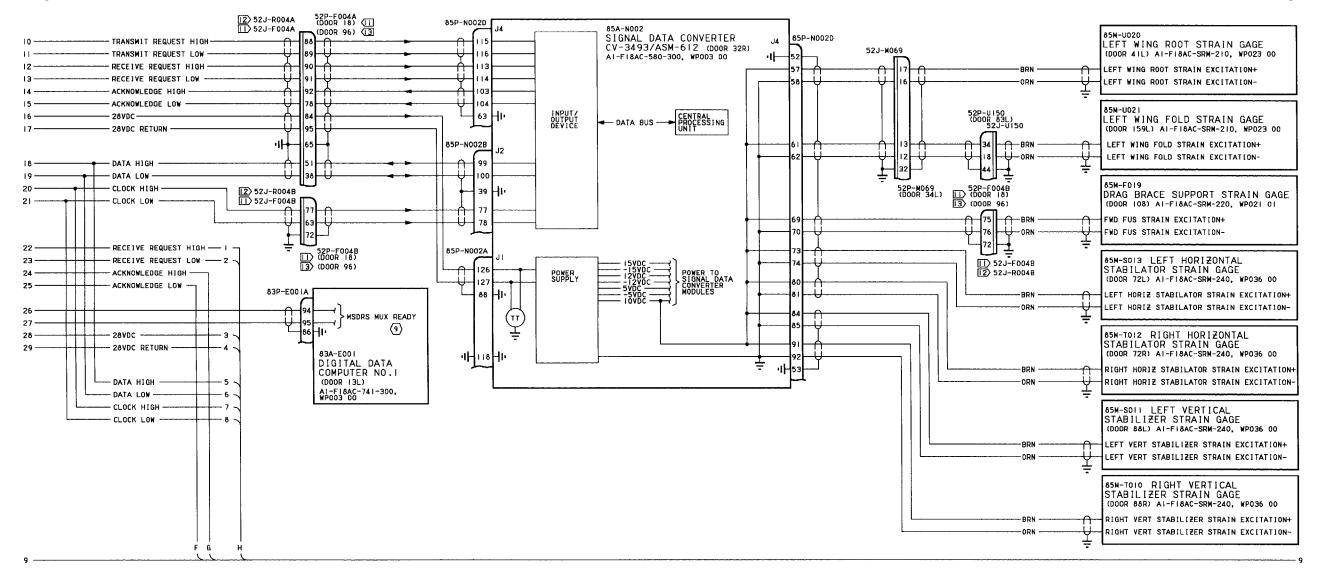


Figure 1. Figure 1. Power Schematic (Sheet 3)

Change 7

Figure 1.



00500104

Figure 1. Power Schematic (Sheet 4)

Figure 1.

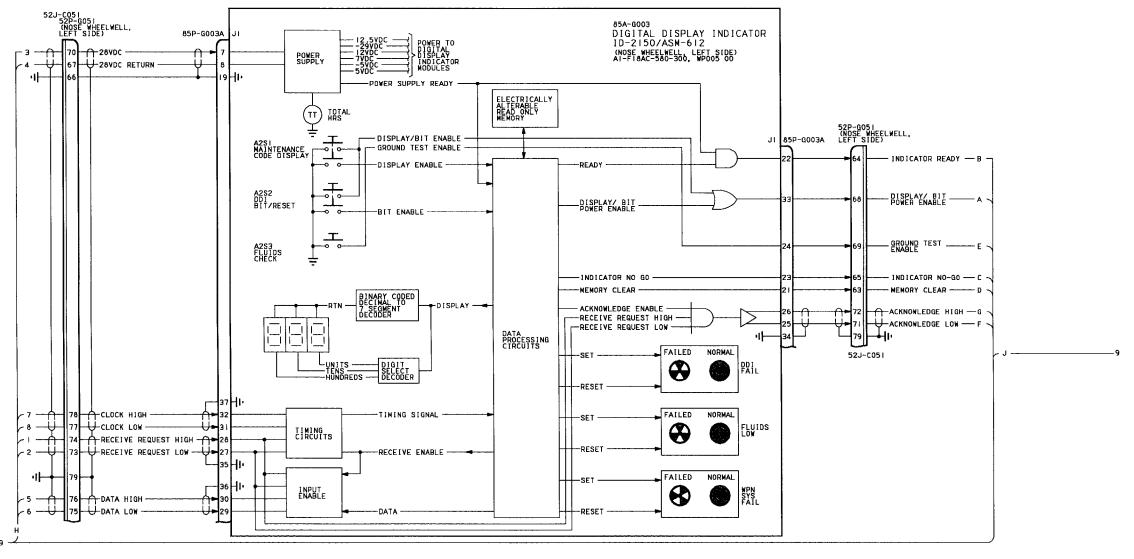


Figure 1. Power Schematic (Sheet 5)

52P-C032 (D00R 10L) 57J-H032

L 28VDC

21 854-K040

MOUNT

85P-K04GB

1.7

MISSION DATA LOADER

(COCKPIT, LEFT SIDE, SEAT RAIL BULKHEAD) A1-F18AC-580-300, WP005 00

MISSION DATA LOADER (COCKPIT, LEFT SIDE, SEAT RAIL BULKHEAD) A1-F18AC-SBO-300, WP005 00

POWER SUPPLY

LOSS OF PRIME
POWER OUTPUT WHEN
WHEN 28VOC INPUT
FALLS BELOW PRESET VALUE

POWER ON RESET CIRCUIT ALLOWS DATA PRUCESSING TO BEGIN AT LOGICAL POINT WHEN POWER FIRST APPLIED OR REAPPLIED AFTER POWER LOSS

PROCESSOR

OUTPUT MOL MUX
READY HIGH WHEN
COMMUNICATION
WITH DIGITAL DATA
COMPUTER NO.1
ESTABLISHED

GUTPUT MOL MUX READY LOW

12VDC - MDL -12VDC - OPERATING 5.1VDC - VOLTAGES

Change 7

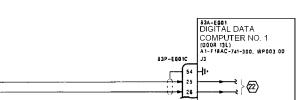
LEGEND

I. CONTINUITY TESTS:

- MHEN A LOW LEVEL CURRENT SWITCHING RELAY (10 M) THE SHOOTING LIBERT OF THE SHOOTING LIBERT
- DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINITY WITH MULTIMETER ON TRI SCALE, PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (I) SHORTS TO GROUND.
- (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3) SHORTS BETWEEN SHIELD AND CONDUCTORS
- (4) SHIELD CONTINUITY.
- WHEN ELECTRICAL POWER IS OFF 24 PDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY 185). MAKE SURE MULTIMETER LEADS/JUMPER MIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTIDUITY.
- 2. NONSTANDARD SYMBOLS:
 - ⊕ IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE I.
 - ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE |.

- (3) ENGINE START AND GROUND MAINTENANCE MODE INTERFACE SCHEMATIC. AI-FIBAC-240-500. WP005 00.
- DC POWER SYSTEM SCHEMATIC, AI-F18AC-420-500, WP004 00.
- (5) GROUND POWER SWITCHING SCHEMATIC, AI-FIBAC-420-500, WP005 00.
- (6) APU CONTROL SYSTEM SCHEMATIC, A1-F18AC-240-500, WP004 00.
- (7) SECONDARY POWER SYSTEM INTERFACE SCHEMATIC, WP018 00.
- (8) FLUIDS TEST SCHEMATIC, WP006 00.
- (9) BUILT-IN TEST SCHEMATIC, WP012 00.
- RECORD FUNCTION SCHEMATIC, WP014 00
- F/A-18A .
- [2] F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 3 F/A-18B.
- 13 161353 THRU 161528 BEFORE F/A-18 AFC 49.
 15 161702 AND UP, ALSO 161353 THRU 161528 AFTER F/A-18 AFC 49.

- [6] 161702 THRU 163118 BEFORE F/A-18 AFC 90.
 [7] 161353 THRU 163118 BEFORE F/A-18 AFC 90.
 [8] 163119 AND UP: ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.
- 19 161353 THRU 161528 BEFORE F/A-18 AFC 90
- 20 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 90. F/A-18A 162394 THRU 163175 AFTER F/A-18 AFC 253 OR F/A-18 AFC 292.
- 2 MISSION DATA LOADER BUILT-IN TEST SCHEMATIC, WP022 00.



Change 3 - 15 April 1997

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FLUIDS TEST

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes WP006 00, dated 1 May 1986.

Reference Material

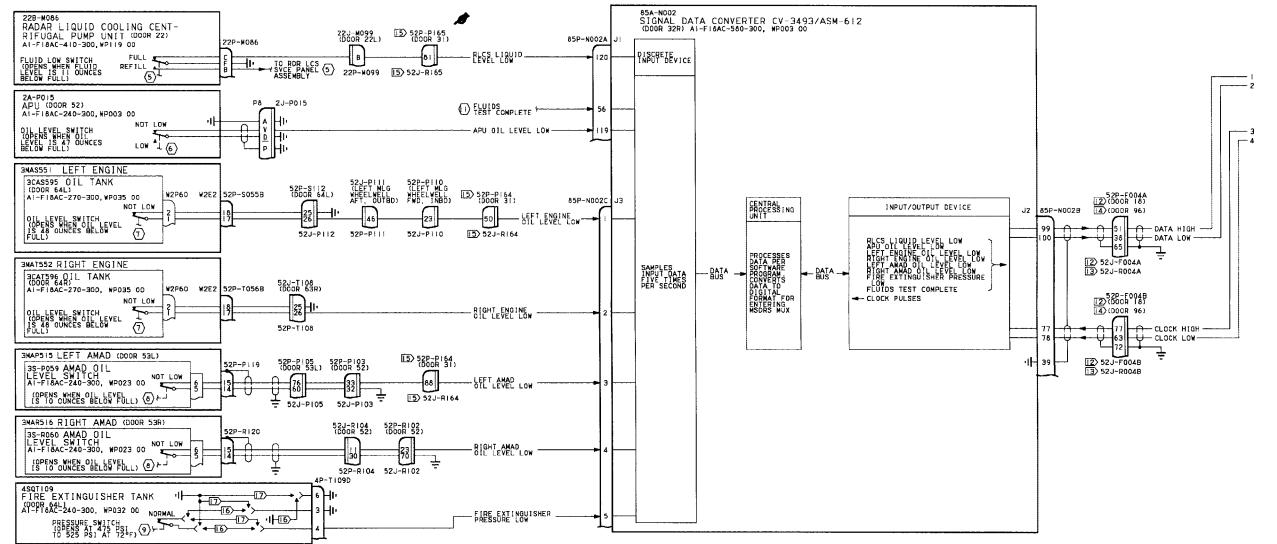
None

Alphabetical Index

Subject	Page No.
Fluids Test Schematic, Figure 1	3

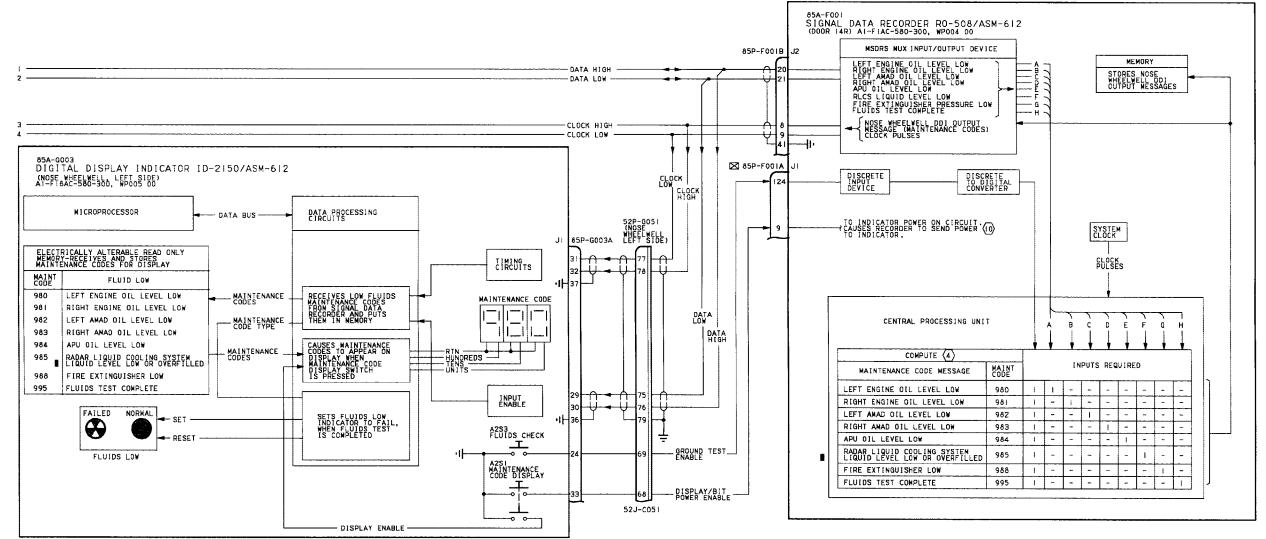
Record of Applicable Technical Directives

None



18AC-580-50-(7-1)-E-GRID

Figure 1. Fluids Test Schematic (Sheet 1)



18AC-580-50-(7-2)-E-GRID

Figure 1. Fluids Test Schematic (Sheet 2)

LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FISA()-WDM-DOD.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY (D) IS REMOYED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION, DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL REYICES (SWITCHES/RELAY CONTACTS) FOR THAT DO NOT GO THROUGH SHITCHES/RELAY CONTACTS NAY USE THE
 - D. WHEN TESTING CONTINUITY. TEST FOR:
 - (I) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E HHEN ELECTRICAL POWER IS OFF, 24 VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY AME SURE WILL HEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
 - # IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 1.
 - ☐ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.
- 3. LINE UNDER LETTER (S) INDICATES LOWER CASE PIN LETTER.
- 4 EXPLANATION OF MATRIX
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY, EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED.
 - (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE
 - (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.
- (5) RADAR LIQUID COOLING SYSTEM SCHEMATIC, A1-F18AC-410-500, WP014 00.
- (6) APU LUBRICATION SYSTEM SCHEMATIC, AI-FIBAC-240-500.WP006 00.
- (7) LUBRICATION SYSTEM SCHEMATIC, AI-FIBAC-270-500.WP004 00.
- (8) AMAD LUBRICATION SYSTEM SCHEMATIC, A!-FI8AC-240-500, WP007 00.
- (9) APU FIRE EXTINGUISHING SYSTEM SCHEMATIC, A!-FIBAC-240-500, WPO10 00.
- (0) POWER SCHENATIC, WP005 00.
- (I) INPUT IS OPEN TO INDICATE FLUIDS TEST IS COMPLETE.
- 12>F/A-18A.
- 13 F/A-188 PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- [**4** F/A−18B.
- 15 162445 AND UP.
- 16 TANK PART NUMBER 33500002 AND 33500003.
- TANK PART NUMBER 826200-107.

18AG-580-50-(7-3)E-GRID

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC LEFT ENGINE INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP009 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject		Page No	
Left Engine Interface Schematic Figure 1		2	

Record of Applicable Technical Directives

None

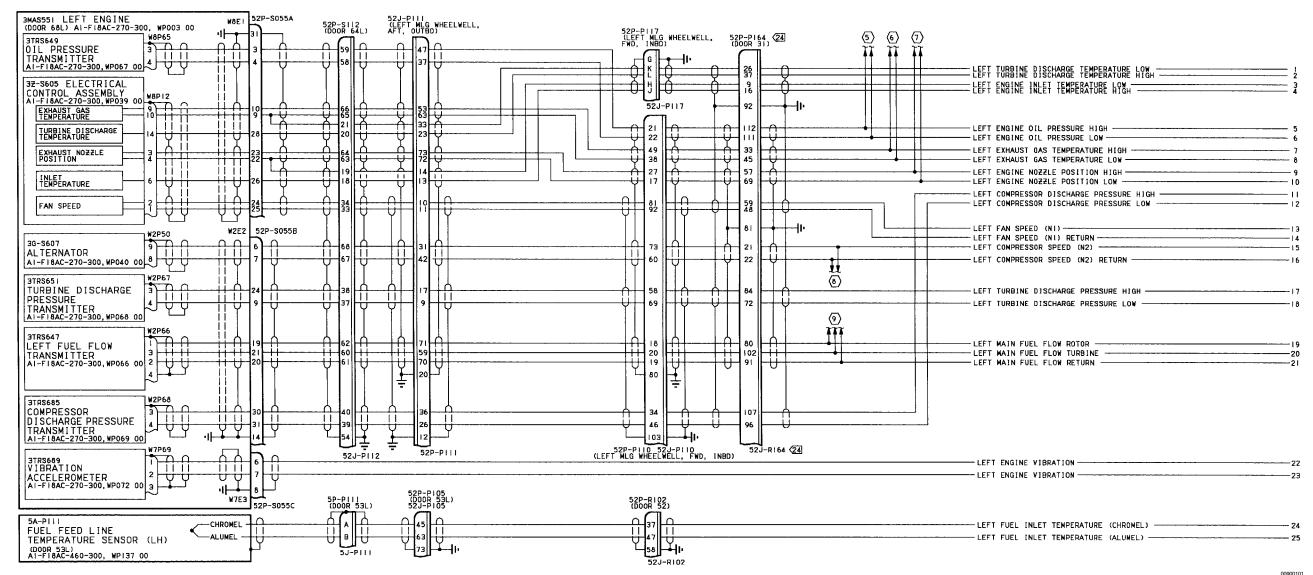


Figure 1.

Figure 1. Left Engine Interface Schematic (Sheet 1)

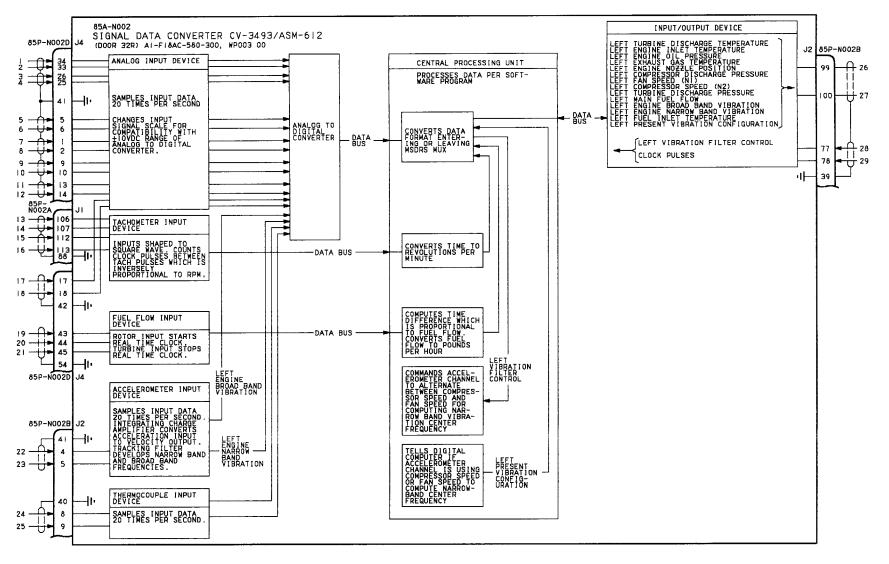


Figure 1. Figure 1. Left Engine Interface Schematic (Sheet 2)

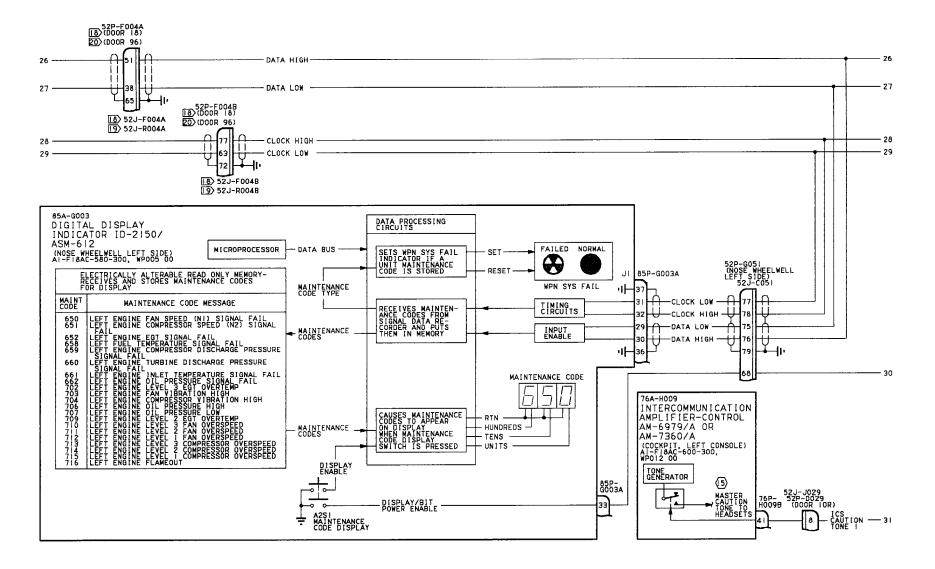


Figure 1. Left Engine Interface Schematic (Sheet 3)

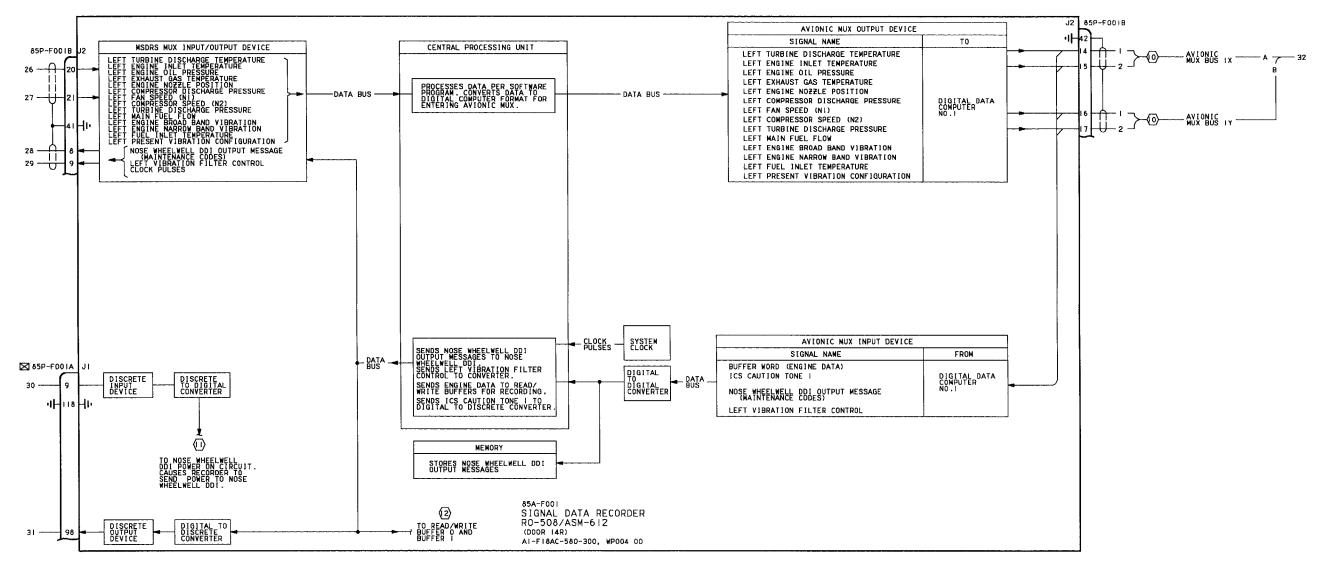


Figure 1. Figure 1. Left Engine Interface Schematic (Sheet 4)

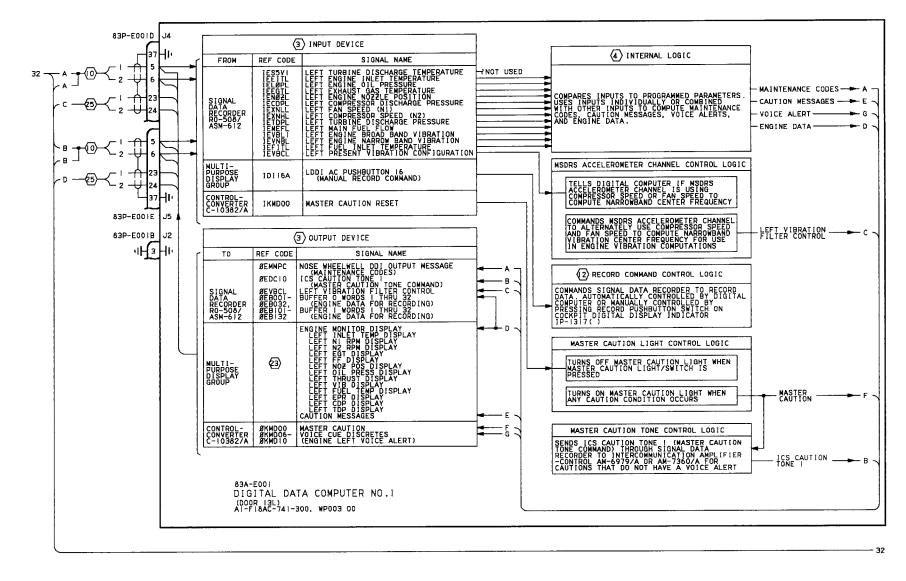
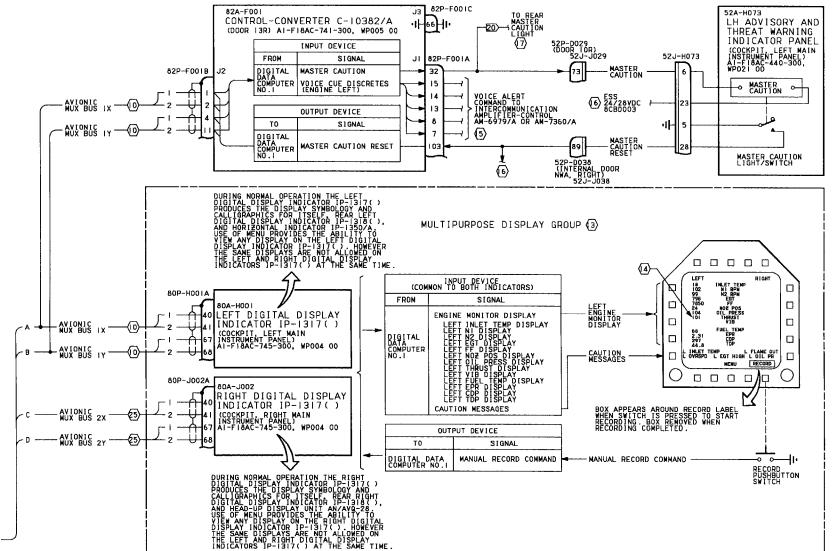


Figure 1.

Figure 1. Left Engine Interface Schematic (Sheet 5)

009 00



LEGEND

- I. CONTINUITY TEST:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-000.
 - B. MHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY (M) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION, DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE. REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON MXI SCALE PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE MXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☑). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - → IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 1.
 - ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE !.



- FOR LOGIC DIAGRAMS RELATING TO REF CODE. REFER TO AI-FIBA()-OLD-000.
- 4 ENGINE SYSTEM CAUTION AND MAINTENANCE CODES SCHEMATICS, AI-FIBAC-270-500, WPOIO 00
- (5) OIL PRESSURE INDICATING SYSTEM SCHEMATIC, AI-FI8AC-270-500, WP007 00
- O TE PRESSORE INDICATING SISTEM SCHEMATIC, ATTIONCT 270-500, MPOUT
- (6) EGT INDICATING SYSTEM SCHENATIC, AI-FI8AC-270-500, WP007 00.
- 7 NOZZLE POSITION INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00
- (8) RPM INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 DO
- (9) FUEL FLOW INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00.
- (0) AVIONIC MUX CHANNEL I SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- (I) POWER SCHEMATIC, WP005 00.
- (2) RECORD FUNCTION SCHEMATIC, WP014 00.
- (3) THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP— 1317() RIGHT DIGITAL DISPLAY INDICATOR IP—1317(). HEAD-UP DISPLAY UNIT ANALYO 28. HORIZONTAL INDICATOR IP—1350/A, AND ON FA-16B THE FEAR LEFT DIGITAL DISPLAY INDICATOR IP—1318(). AND REAR RIGHT DIGITAL DISPLAY INDICATOR IP—1318(). AND REAR CENTER DIGITAL DISPLAY INDICATOR IP—1318().
- (4) DISPLAYED ONLY DURING THRUST TEST BEFORE TAKEOFF
- (5) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC. AI-FIBAC-600-500, WPDIE 00
- (6) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00
- (17) REAR COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP007 00
- T8>F/A-18A.
- 19 F/A-188 PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN
- 20) F/A-18B.
- DELETED
- 22 DELETED.
- (3) DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ON MORE THAN ONE INDICATOR. TROUBLESHOOT USING AI-FIBA()-010-000 IMPUT REF CODES IF MALFUNCTION EXISTS ONLY ON ONE INDICATOR, TROUBLESHOOT USING SHOOT BY DOING DISPLAY TEST, AI-FIBAC-745-200, WP004 00 (F/A-18A) OR WP005 00 (F/A-18B)
- 24) 162445 AND UP
- \$\forall \text{AVIONIC MUX CHANNEL 2 SCHENATIC, AI-FIBAC-741-500, WP005 00}

00900106

Figure 1. Left Engine Interface Schematic (Sheet 6)

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC LEFT ENGINE INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292.

Reference Material

None

Alphabetical Index

Subject	Page No.
Left Engine Interface Schematic, Figure 1	 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

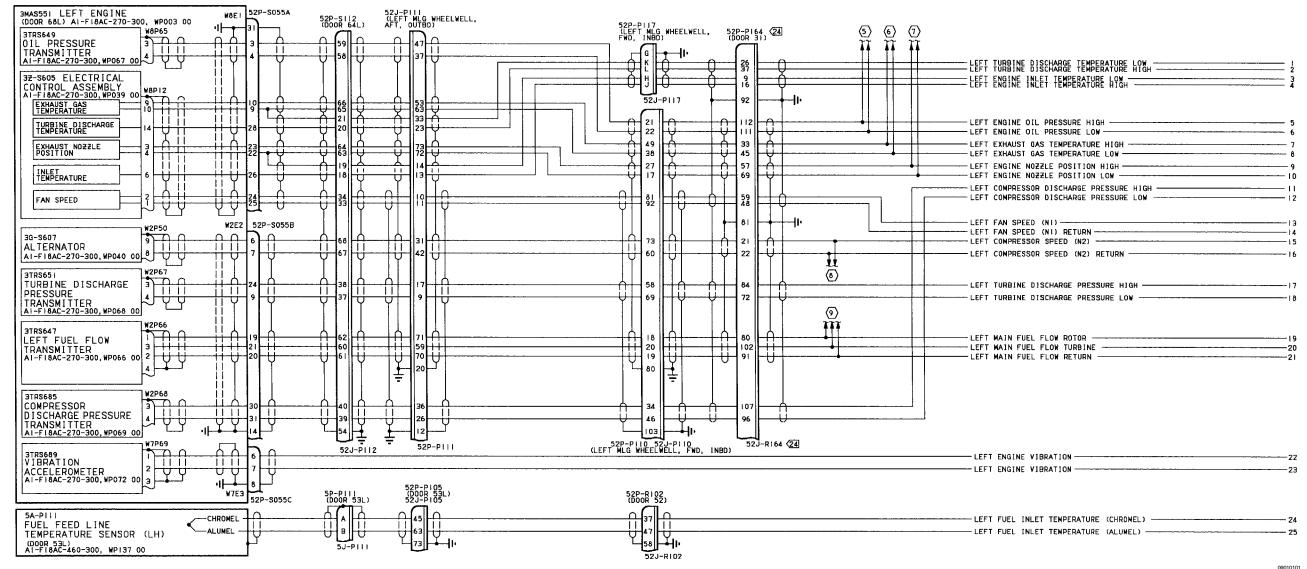


Figure 1.

Figure 1. Left Engine Interface Schematic (Sheet 1)

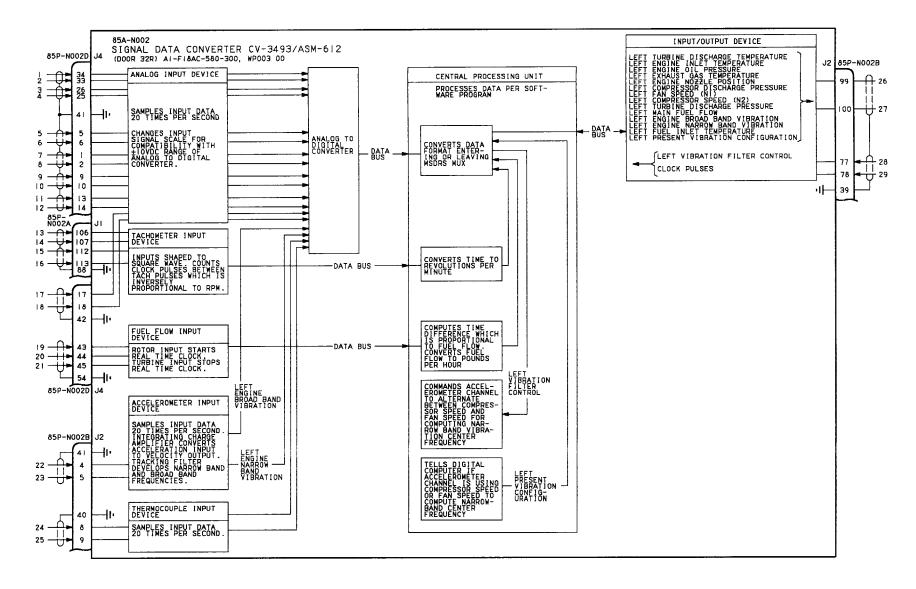


Figure 1. Left Engine Interface Schematic (Sheet 2)

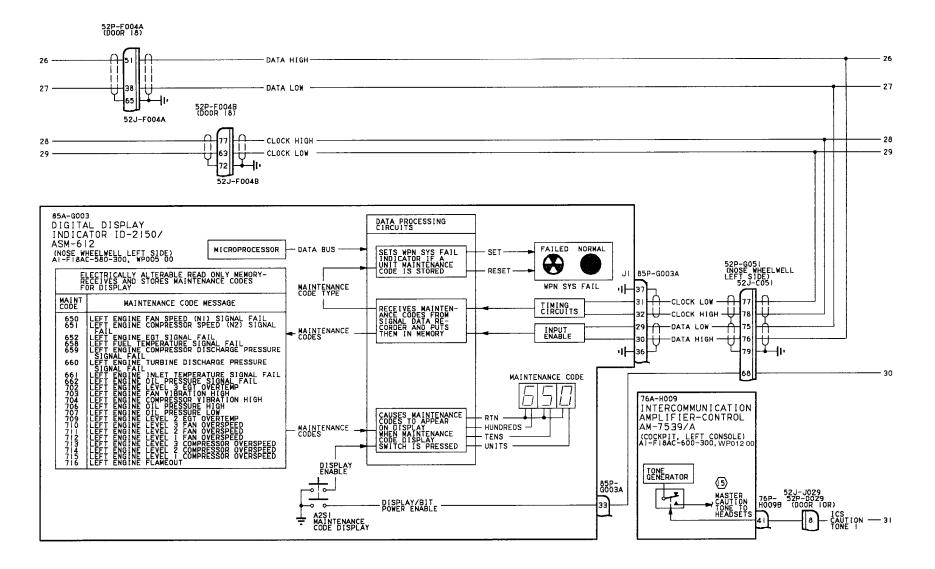


Figure 1. Left Engine Interface Schematic (Sheet 3)

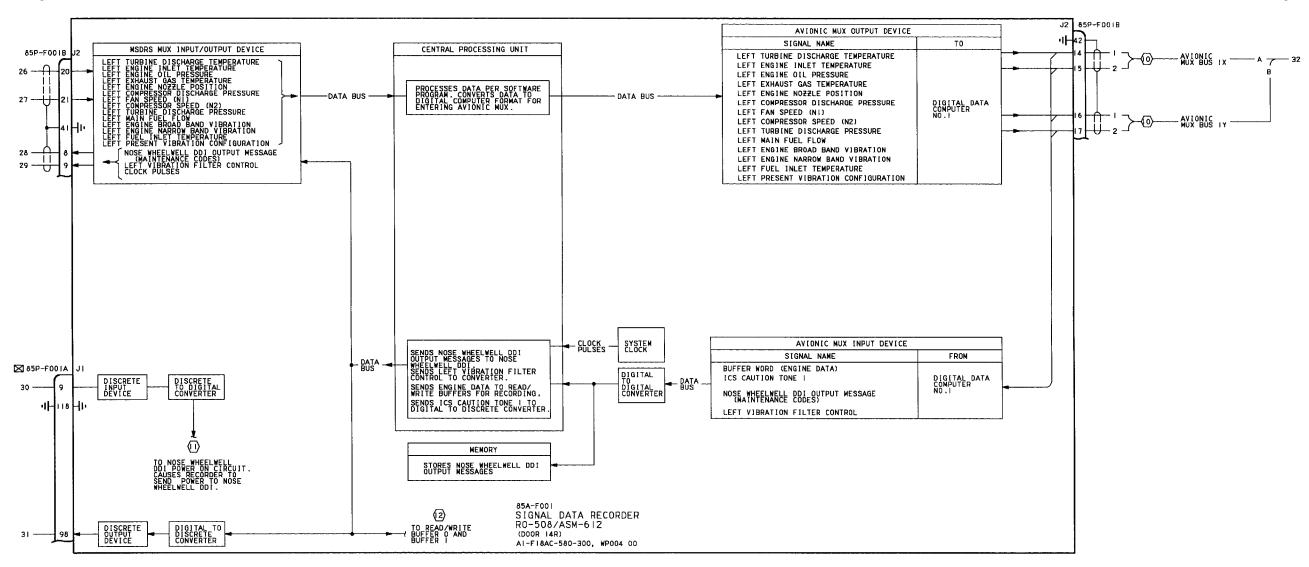


Figure 1. Figure 1. Left Engine Interface Schematic (Sheet 4)

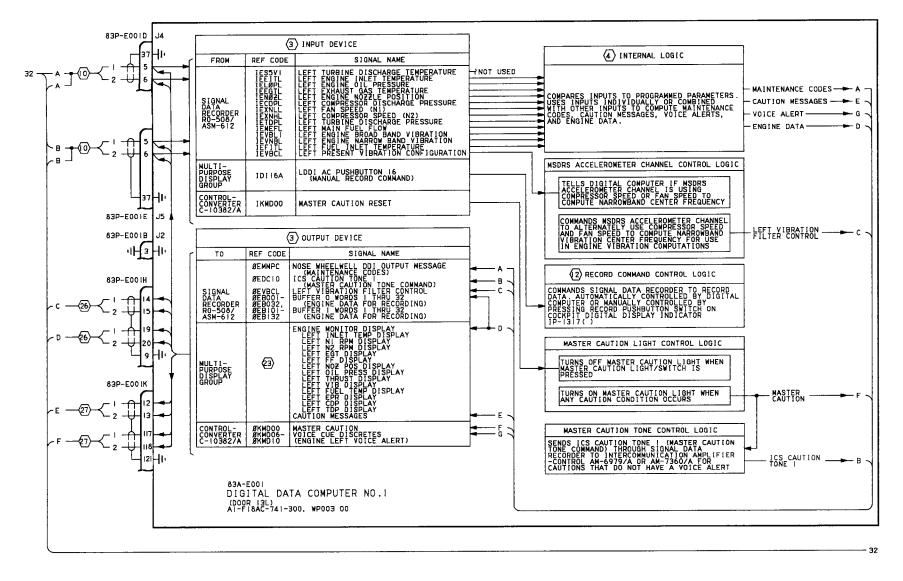


Figure 1. Left Engine Interface Schematic (Sheet 5)

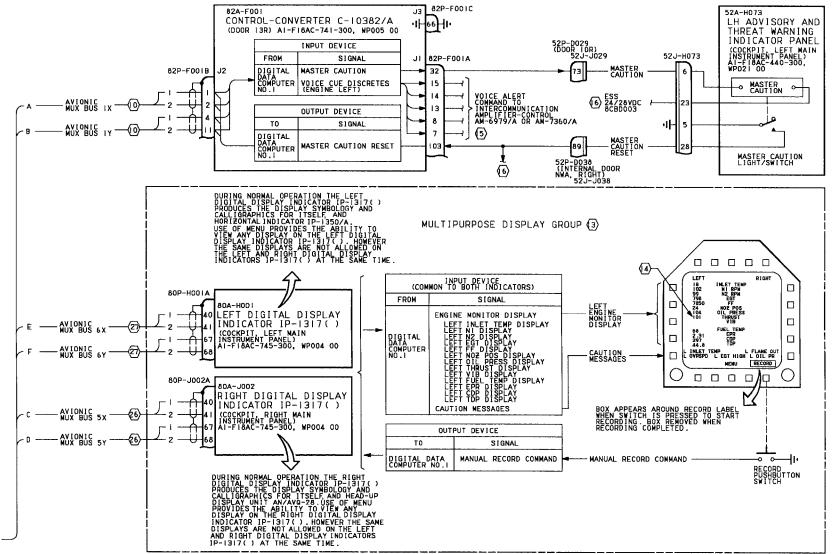


Figure 1. Left Engine Interface Schematic (Sheet 6)

LEGEND

- I. CONTINUITY TEST:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FI8A()-WDM-000.
 - WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY (A)) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY BELAY AND SOCKET FOR CORRECT REINSTALLATION, DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIWETER ON RM SCALE FRIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY [3]). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - → IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE +.
 - ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE !.



- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE. REFER TO AI-FIBA()-OLD-000. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100
- (4) ENGINE SYSTEM CAUTION AND MAINTENANCE CODES SCHEMATICS, AI-F18AC-270-500, WPO10 00.
- (5) OIL PRESSURE INDICATING SYSTEM SCHEMATIC, AI-F18AC-270-500, WPOO7 00.
- (6) EGT INDICATING SYSTEM SCHENATIC, AI-F18AC-270-500, WP007 00
- (7) NOZZLE POSITION INDICATING SYSTEM SCHEMATIC, A1-F18AC-270-500, WPOO7 OO.
- (8) RPM INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WPOO7 DO
- (9) FUEL FLOW INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00.
- (0) AVIONIC MUX CHANNEL I SCHEMATIC, AI-FIBAC-741-500, WP004 00
- (I) POWER SCHEMATIC, WP005 00.
- (2) RECORD FUNCTION SCHEMATIC, WP014 00.
- (4) DISPLAYED ONLY DURING THRUST TEST BEFORE TAKEOFF
- (5) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-600-500, WPOI3 00.
- (6) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00
- (7) DELETED
- (8) DELETED.
- 19) DELETED.
- 20) F/A-18B
- 21) DELETED .
- 22 DELETED
- (3) DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. FROM LESHOOT USING AI-FIBA() -010-000 IMPUT REF CODES, IF MALFUNCTION EXISTS ON THE INDICATOR. TROUBLESHOOT USING SHOOT BY DOING DISPLAY TEST, AI-FIBAC-745-200, MP004 00 (F/A-18A) OR MP005 00 (F/A-18B)
- 24) 162445 AND UP
- (5) DELETED.
- (6) AVIONIC MUX CHANNEL 5 SCHENATIC, AI-FIBAC-741-500, WPOIB 00.
- \$7) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00

Figure 1.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - RIGHT ENGINE INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292, AND F/A-18B

This WP supersedes WP010 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No
Right Engine Interface Schematic, Figure 1	 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 49	-	Addition of Sealed Lead Acid Battery (ECP MDA-F/A-18-00074)	1 Sep 86	ECP coverage only

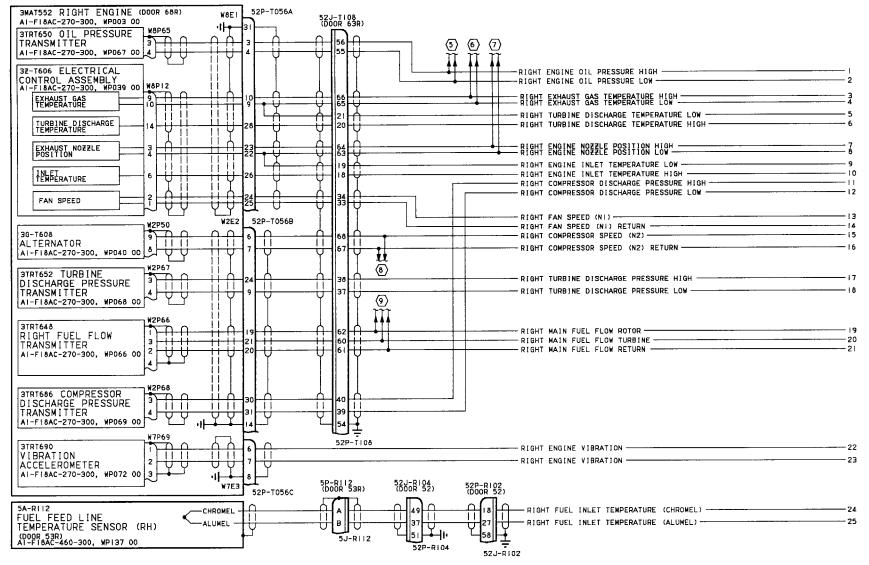


Figure 1. Right Engine Interface Schematic (Sheet 1)

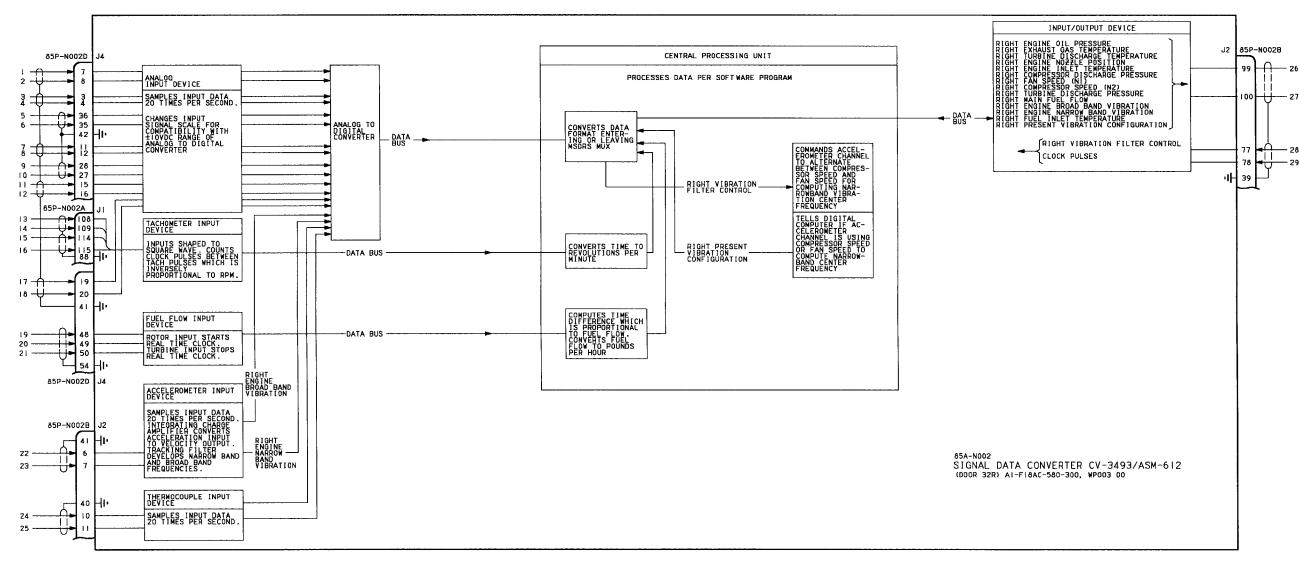


Figure 1.

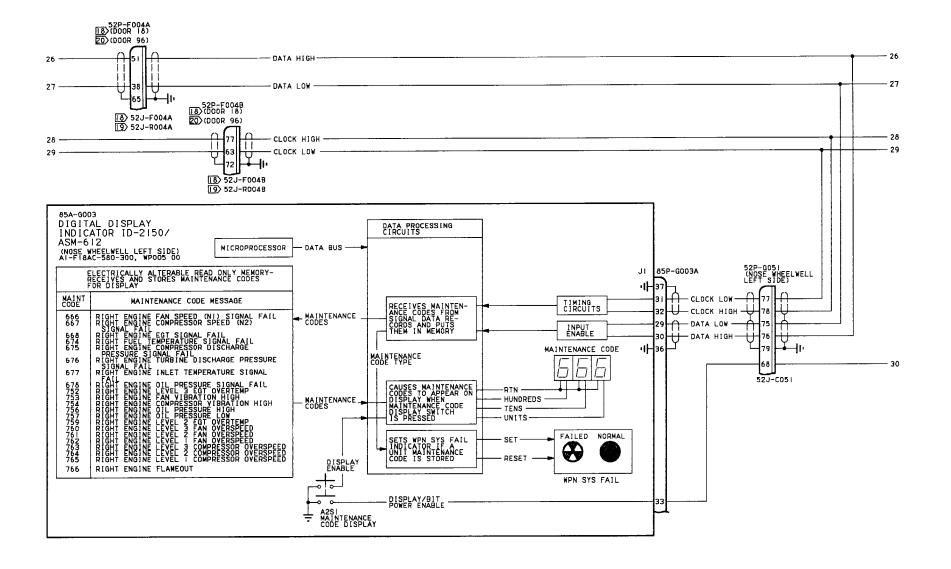


Figure 1. Right Engine Interface Schematic (Sheet 3)

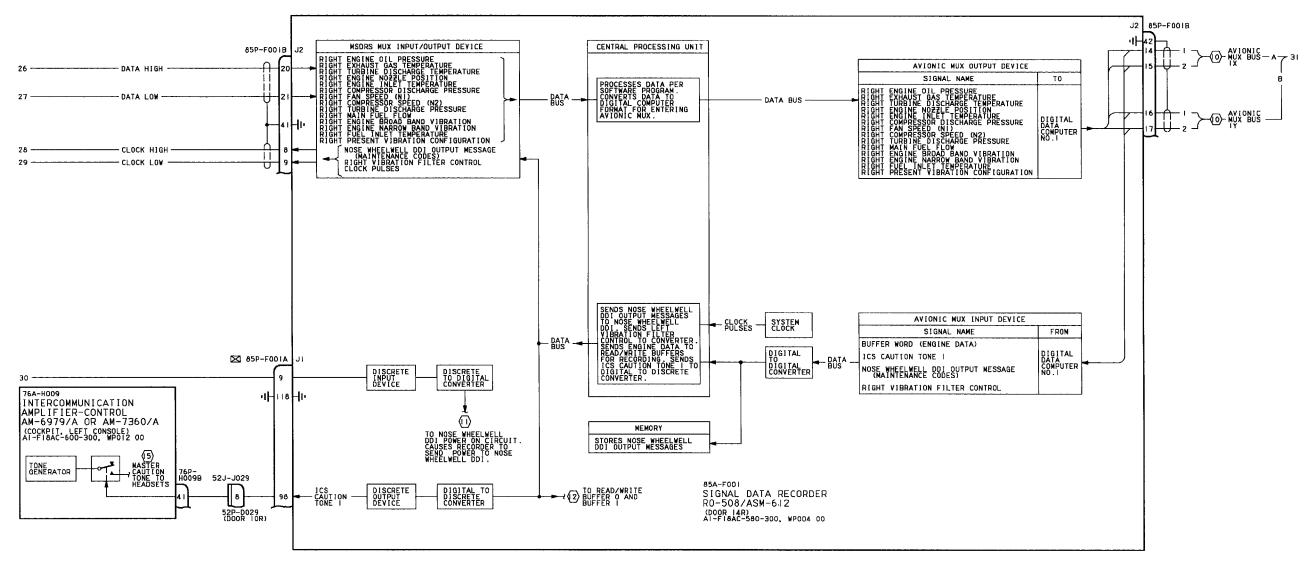


Figure 1.

Figure 1. Right Engine Interface Schematic (Sheet 4)

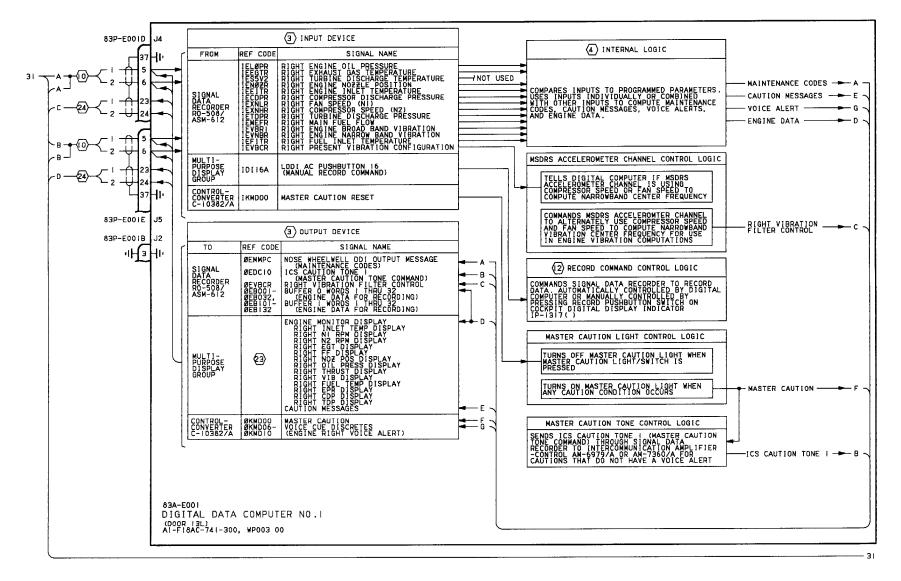
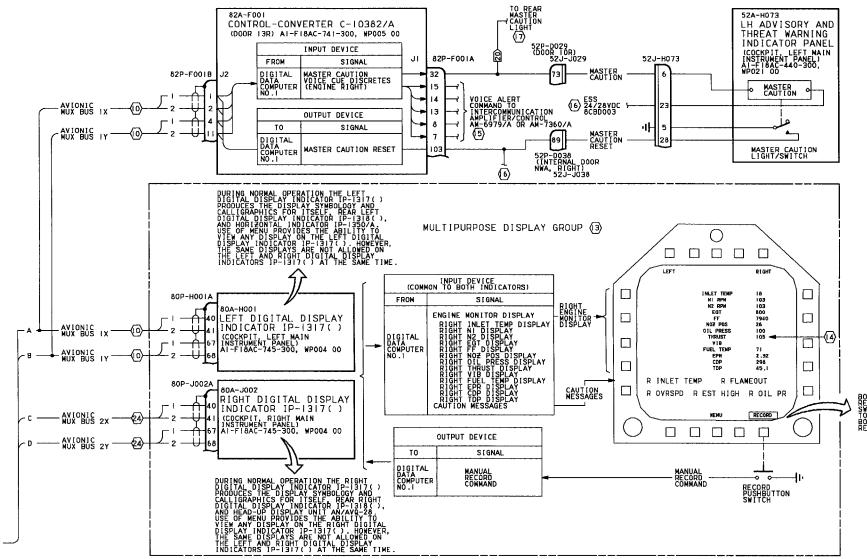


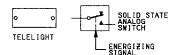
Figure 1. Right Engine Interface Schematic (Sheet 5)



LEGEND

- I. CONTINUITY TESTS:
- ONTINUITY TESTS:

 ALL AIPERAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AIF 16AC, I MOWN COLOR OF THE SHOWN IN AIF 16AC, I MOWN COLOR OF THE SHOWN IN AIF 16AC, I MOWN COLOR OF THE SHOW COLOR
- WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY MAKE SURE MULTIMETER LEADS/JUMPER HIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
- IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT.
- IDENTIFIES 24VDC BATTERY VOLTAGE EX-ISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE |.



- (3) FOR LOGIC DIAGRAMS RELATING TO REF FOR MEMORY INSPECT ACCESS FOR ACTION RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100.
- ENGINE SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATICS, AI-F18AC-270-500, WP010 00.

BOX APPEARS AROUND RECORD LABEL WHEN SWITCH IS PRESSED TO START RECORDING. BOX REMOVED WHEN RECORDING COMPLETED.

- (5) OIL PRESSURE INDICATING SYSTEM SCHEM-ATIC, AI-FIBAC-270-500, WP007 00.
- 6 EGT INDICATING SYSTEM SCHEMATIC, AI-F18AC-270-500, WP007 00.
- 7 NOZZLE POSITION INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00.
- RPM INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00.
- 9) FUEL FLOW INDICATING SYSTEM SCHEMATIC, AI-F:8AC-270-500, WP007 00.
- AVIONIC MUX CHANNEL : SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- (I) POWER SCHEMATIC, WP005 00
- RECORD FUNCTION SCHEMATIC, WPO14 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY INDICATOR IP-1350/A, AND ON F/A-18B. THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1318(), REAR REMERT DIGITAL DISPLAY INDICATOR IP-1318(), REFERENTER DIGITAL DISPLAY INDICATOR INDIC
- DISPLAYED ONLY DURING THRUST TEST BEFORE TAKEOFF.
- INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500 WP013 00.
- COCKPIT CAUTION LIGHTS SCHEMATIC, AI-F18AC-440-500, WP006 00.
- REAR COCKPIT CAUTION LIGHTS SCHEMATIC, AI-F18AC-440-500. WP007 00.
- (8) F/A-18A.
- F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 20> F/A-18B
- 21> DELETED.
- 22 DELETED
- AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FIBAC-741-500, WP005 00.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - RIGHT ENGINE INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Right Engine Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 49	-	Addition of Sealed Lead Acid Battery (ECP MDA-F/A-18-00074)	1 Sep 86	ECP coverage only
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

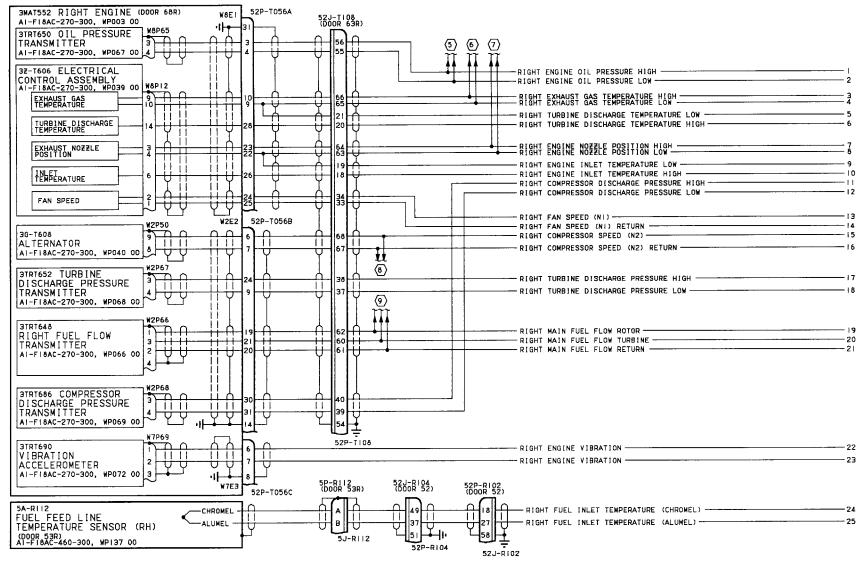


Figure 1. Right Engine Interface Schematic (Sheet 1)

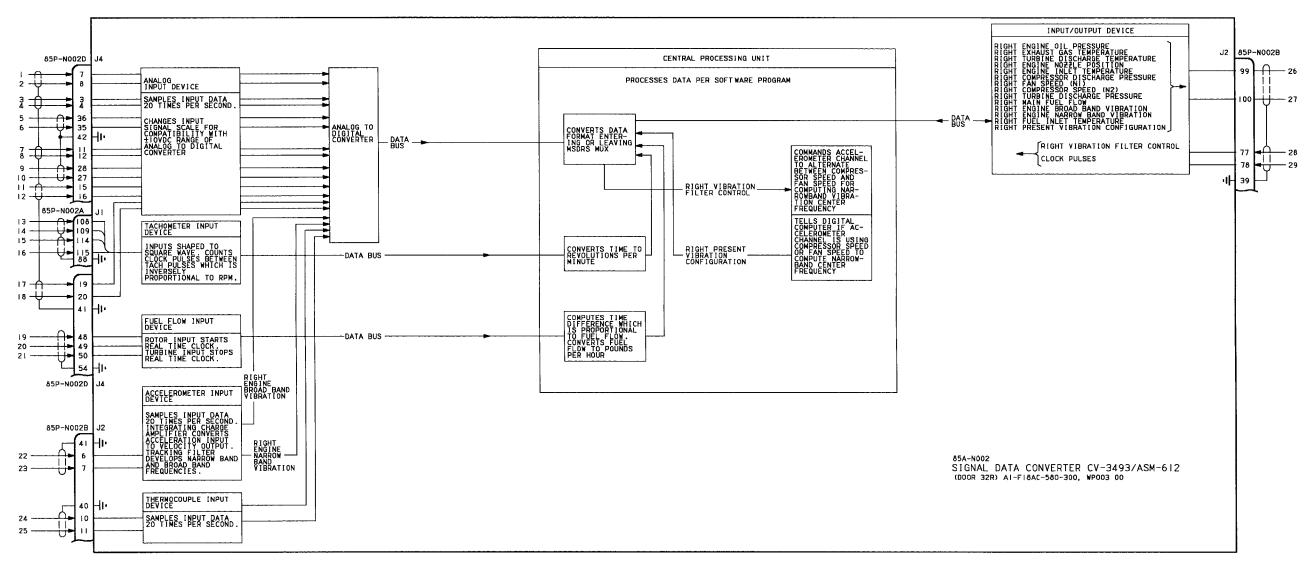


Figure 1. Right Engine Interface Schematic (Sheet 2)

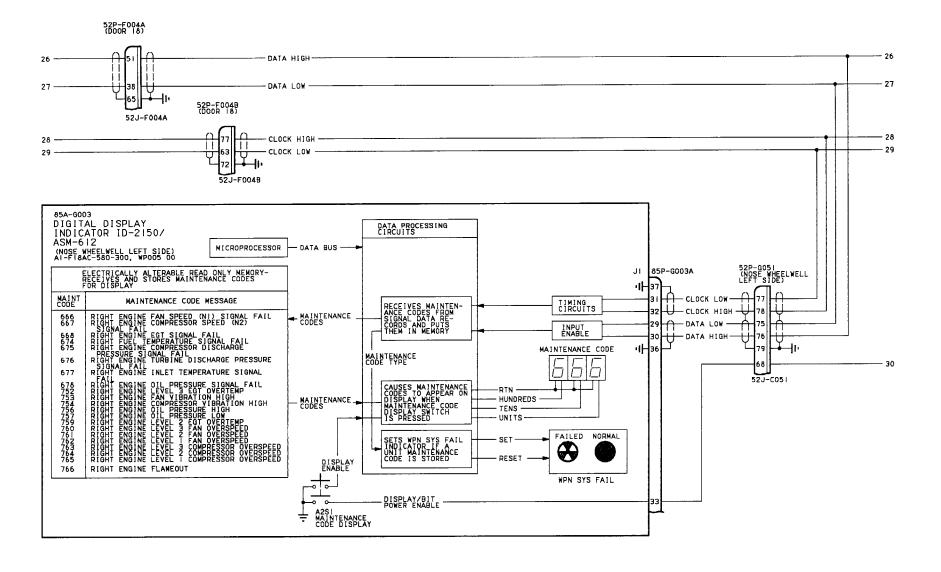


Figure 1. Right Engine Interface Schematic (Sheet 3)

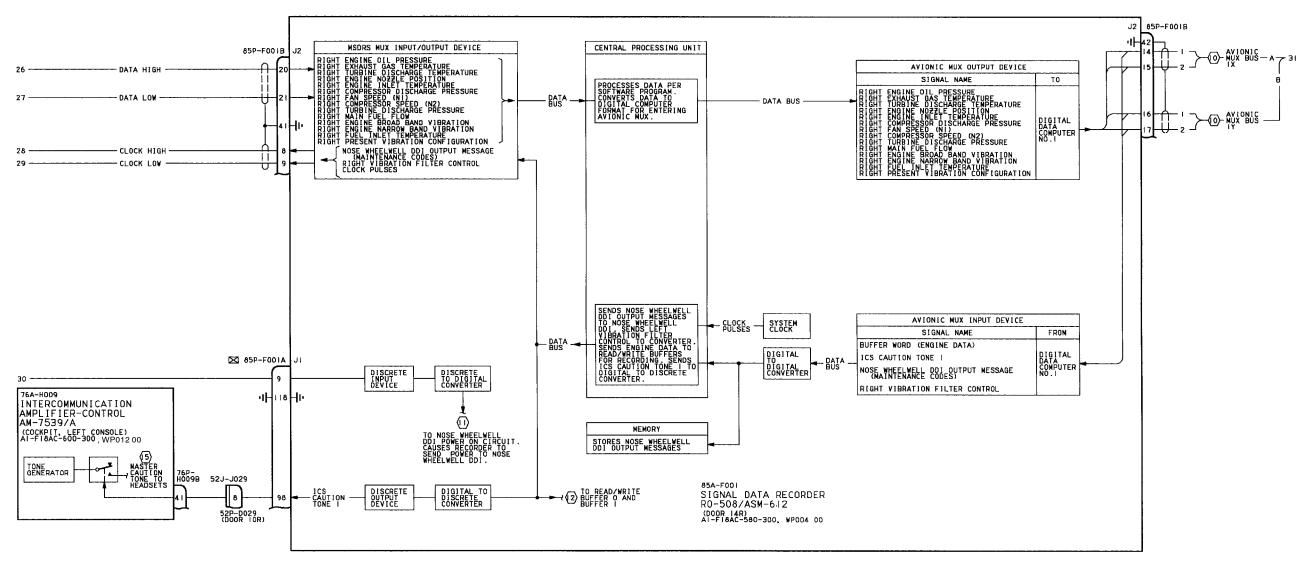


Figure 1. Right Engine Interface Schematic (Sheet 4)

Figure 1.

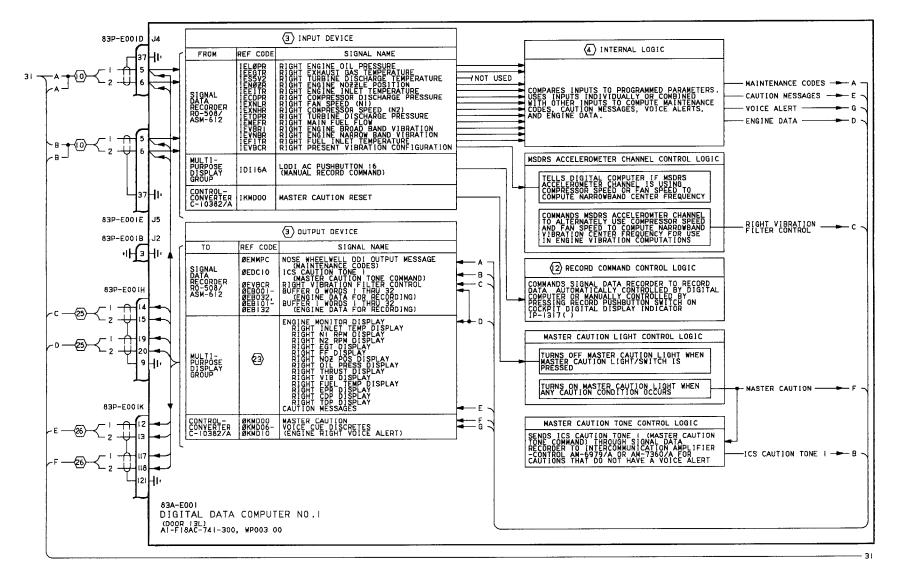
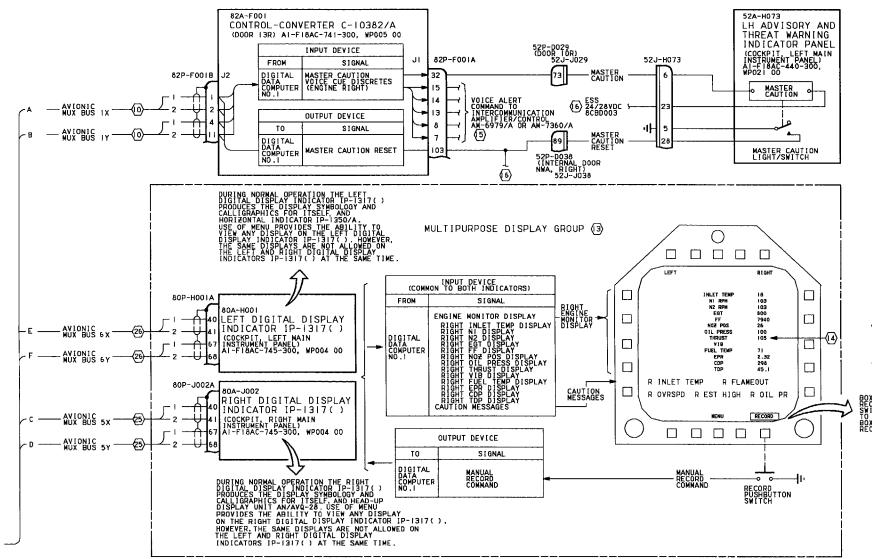
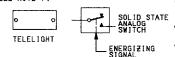


Figure 1. Right Engine Interface Schematic (Sheet 5)



- I. CONTINUITY TESTS:
- ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-OOO.
- MHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY (#)) IS REMOVED FOR TROUBLESHOOTING IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SMITCHING RELAY WITH ANY OTHER USED RELAY IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- DO NOT TEST LOW LEYEL DEVICES (SMITCH-ES, RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON PXI SCALE, PIN TO PIN TESTS THAT DO NOT CO THROUGH SMITCHES/TELAY CONTACTS MAY USE THE RXI SCALE.
- WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED SOME ALL SOME WILL TIMETER LEADS/JUMPER HIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
- IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT.
- IDENTIFIES 24VDC BATTERY VOLTAGE EX-ISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE | .



- ENGINE SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATICS, AI-F18AC-270-500, WP010 00.

BOX APPEARS AROUND RECORD LABEL WHEN SWITCH IS PRESSED TO START RECORDING. BOX REMOVED WHEN RECORDING COMPLETED.

- 5) OIL PRESSURE INDICATING SYSTEM SCHEM-ATIC. AI-FIBAC-270-500, WP007 00.
- (6) EGT INDICATING SYSTEM SCHEMATIC,
- 7 NOZZLE POSITION INDICATING SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP007 00.
- *(8) RPM INDICATING SYSTEM SCHEMATIC, AI-F18AC-270-500, WP007 00.
- AVIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- POWER SCHEMATIC, WP005 00
- RECORD FUNCTION SCHEMATIC, WPO14 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LETT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY UNIT ANXO-28. HORIZONTAL INDICATOR IP-1350/A. FOR MULTIPURPOSE DISPLAY GROUP, REFER TO A1-F18AC-745-500.
- DISPLAYED ONLY DURING THRUST TEST
- NTERCOMMUNICATION AND AUDIO SYSTEM UNCTIONAL SCHEMATIC, AI-FIBAC-600-500 POOLS 00
- COCKPIT CAUTION LIGHTS SCHEMATIC, AI-F18AC-440-500. WP006 00.
- (7) DELETED.
- (8) DELETED.
- 19 DELETED
- 20 DELETED
- DELETED .
- 22 DELETED
- DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MELVINCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS. ON MORE THAN ONE INDICATOR. TROUBLESHOOT USING AI-FLBA. () -0.D-000 INPUT REF CODES. IF MALFUNCTION EXISTS ONLY ON ONE INDICATOR. TROUBLESHOOT BY DOING DISPLAY TEST. AI-FLBAC-745-200, MPO04 00 (F/A-18A), OR WP005 00 (F/A-18B).
- (24) DELETED.
- AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-F18AC-741-500, WP018 00.
- (6) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FI8AC-741-500, WP019 00.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FUEL SYSTEM INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292, AND F/A-18B

This WP supersedes WP011 00, dated 15 August 1992.

Reference Material

None

Alphabetical Index

Subject	Page No
Fuel System Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 IAFC 056	27 Mar 85	Fuel System Components Replacement and System Inspection (ECP MDA-F/A-18-00158R1 and ECP MDA-F/A-18-100160)	15 Oct 83	
F/A-18 AFC 48	28 Feb 90	Automatic AC Bus Isolation, Incorporation of (ECP-MDA-F/A-18-00121R1)	1 Sep 86	
F/A-18 AFC	31 Dec 90	Motive Flow Fuel Boost Pump Pressure Switch Installation of (ECP MDA-F/A-18-00158R2)	15 Oct 87	

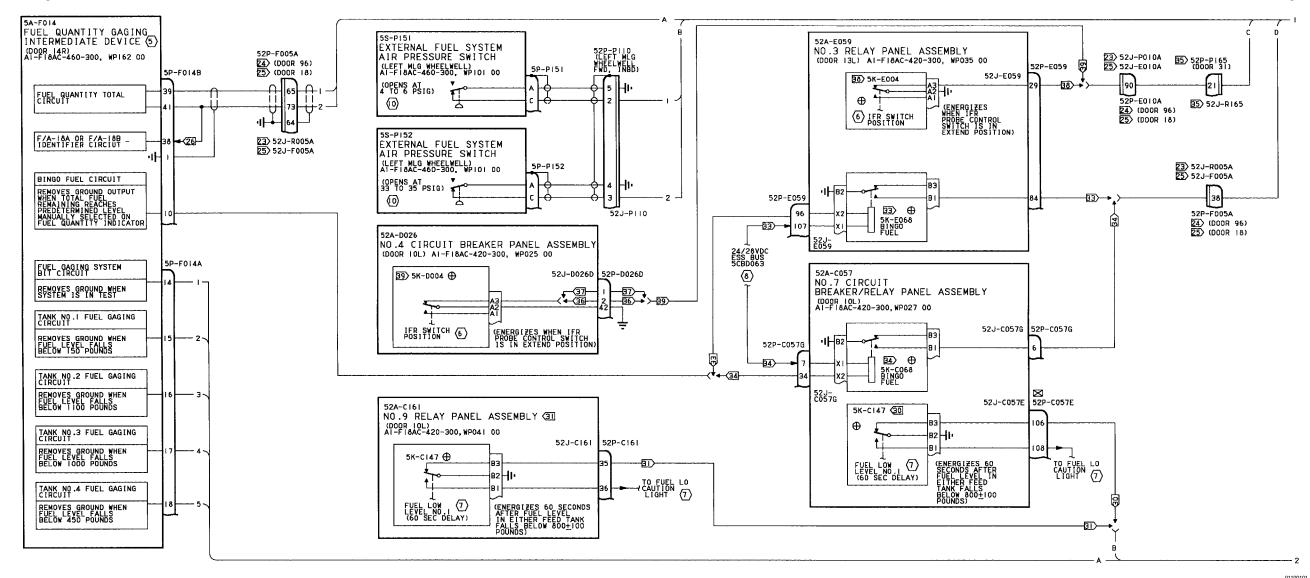


Figure 1. Figure 1. Fuel System Interface Schematic (Sheet 1)

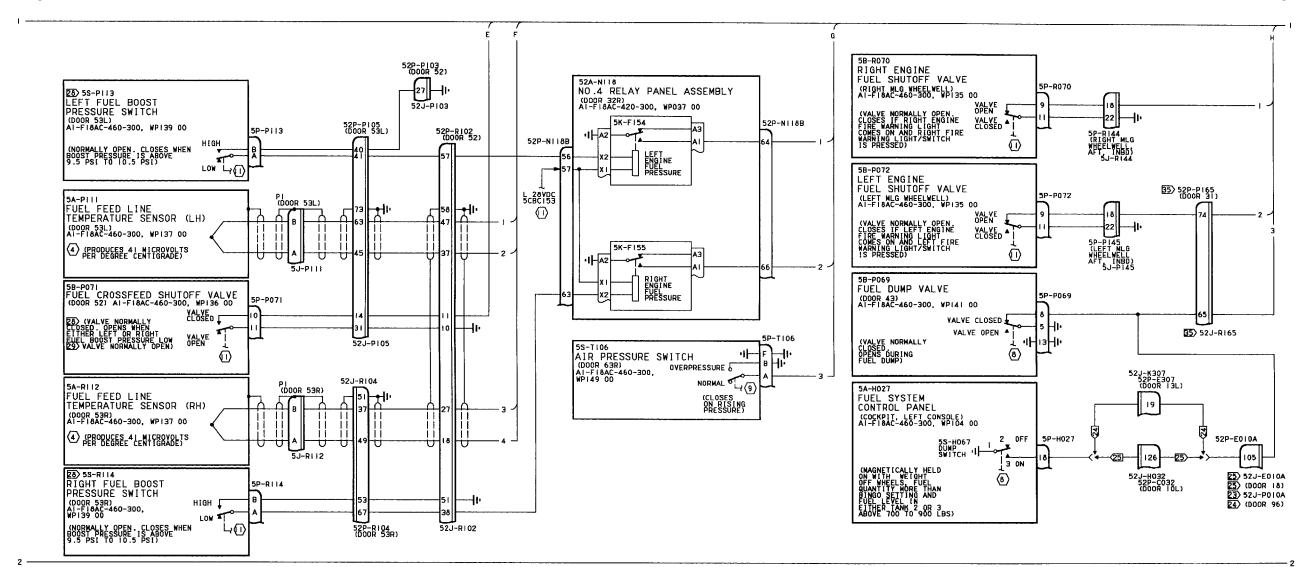
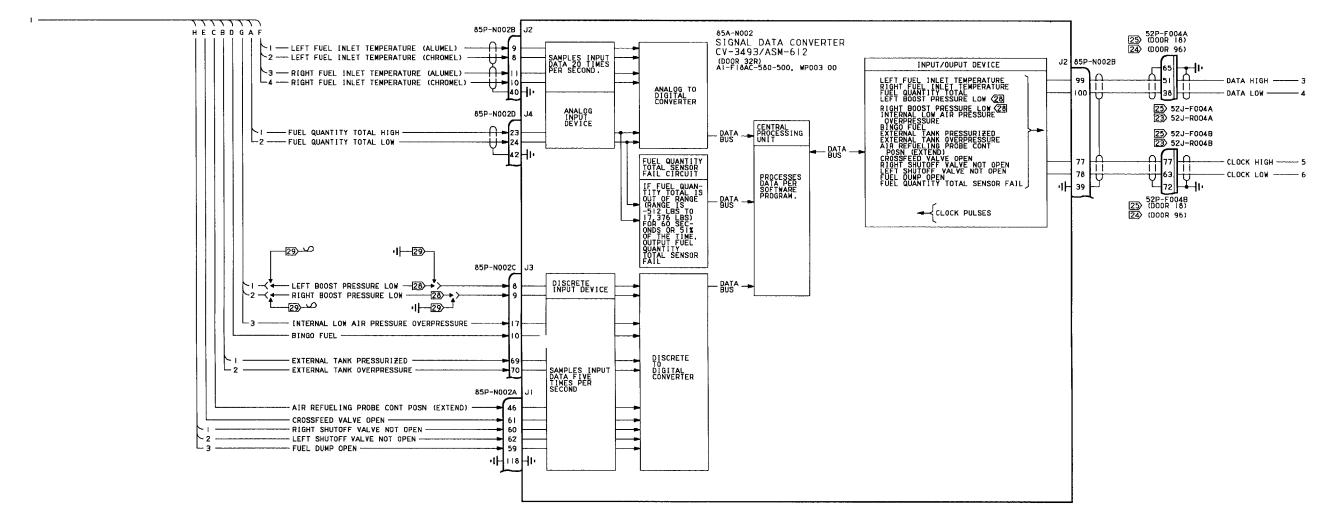


Figure 1.

Figure 1. Fuel System Interface Schematic (Sheet 2)



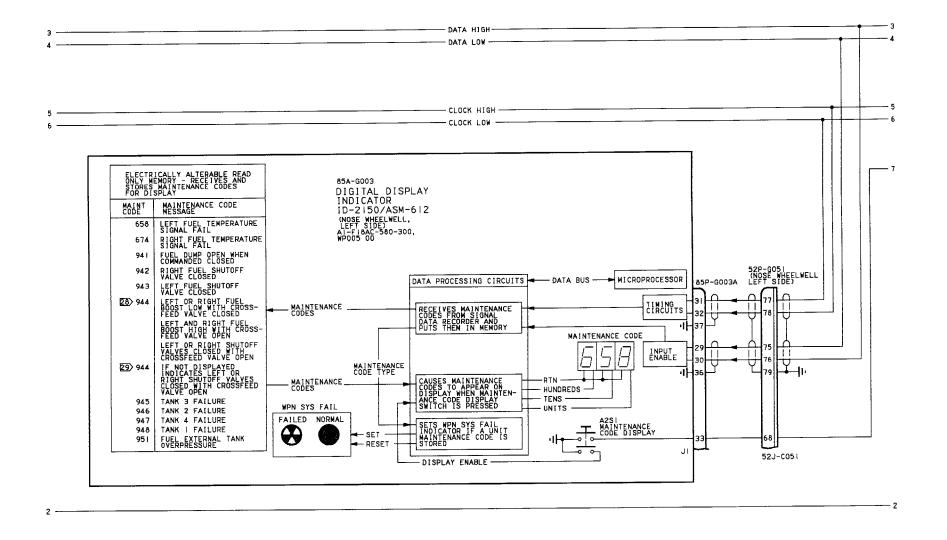


Figure 1. Fuel System Interface Schematic (Sheet 4)

52J-J029

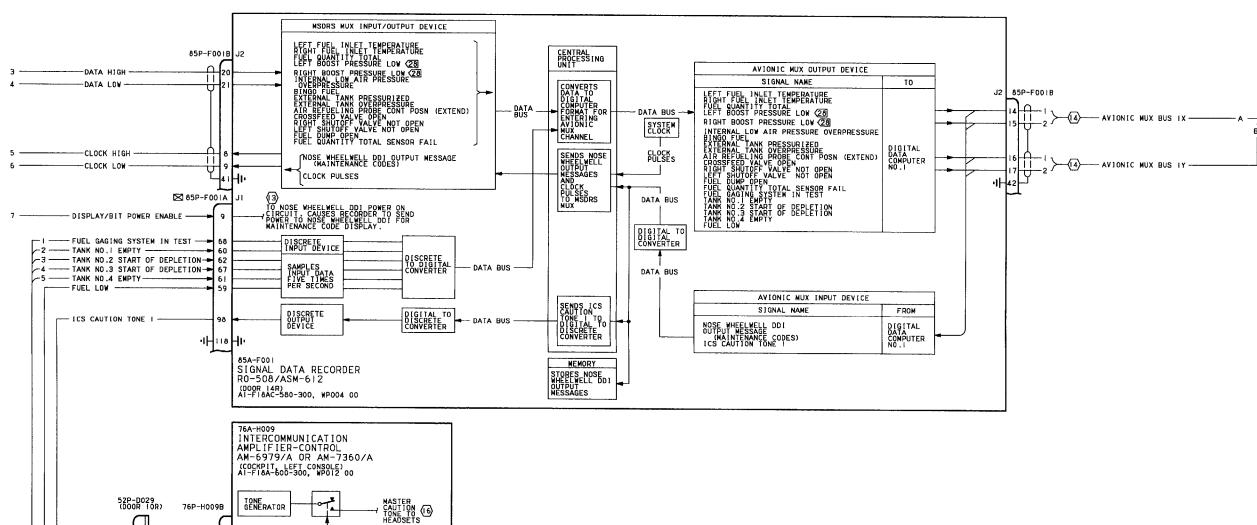


Figure 1. Figure 1. Fuel System Interface Schematic (Sheet 5)

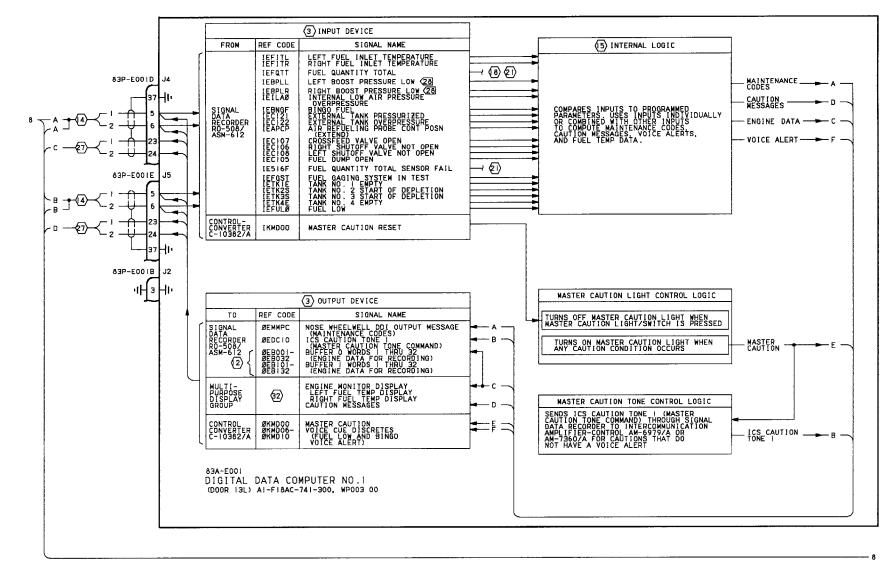


Figure 1. Fuel System Interface Schematic (Sheet 6)

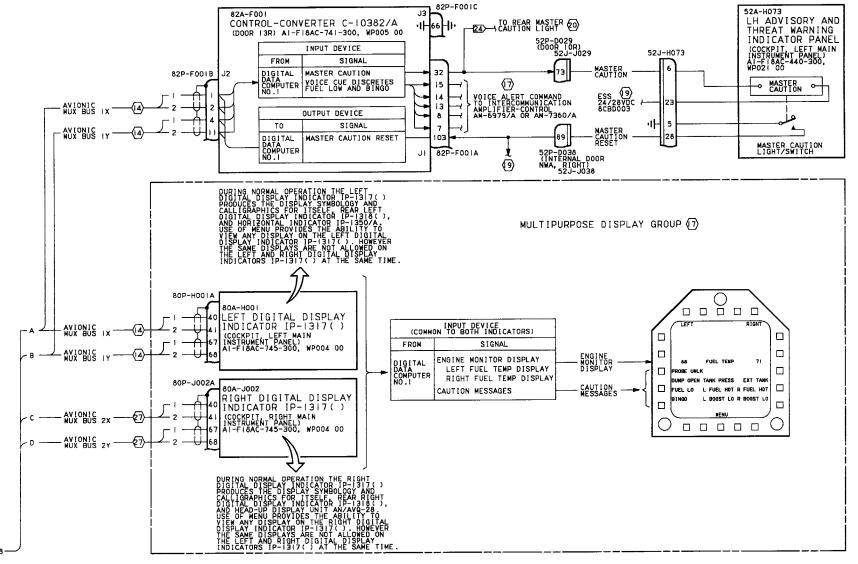


Figure 1. Fuel System Interface Schematic (Sheet 7)

I. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND
- B. WHEN A LOW LEYEL CURRENT SWITCHING RELAY (IDENTIFIED BY THE AS THE ASSOCIATION OF THE ASSOCIATION OF THE ASSOCIATION OF THE ASSOCIATION OF THE USED LOW LEYEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY IT FRIAY IS DEFECTIVE. REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SNITCHES/RELAY CONTACTS)
 FOR CONTINUITY WITH MULTIMETER ON RXI SCALE PIN TO PIN
 TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS
 MAY USE THE RXI SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - 1) SHORTS TO GROUND. 2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS. 3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
- E. WHEN ELECTRICAL POWER IS OFF. 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY) MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. NONSTANDARD SYMBOLS

- DIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT,
- ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE 1.



- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO ALL-FIBAC FIM-100.
- 4 HOT FUEL RECIRCULATION SYSTEM SCHEMATIC,
- (5) INTERNAL FUEL TRANSFER SYSTEM SCHEMATIC, A1-F18AC-460-500,
- (6) INFLIGHT REFUELING SYSTEM SCHEMATIC, AI-FIBAC-460-500, WP005 00.
- 7 FUEL QUANTITY LOW LEVEL WARNING SYSTEM SCHEMATIC, A1-F18AC-460-500, WP013 00.
- 8 FUEL DUMP SYSTEM SCHEMATIC, AI-F18AC-460-500,
- (9) FUEL PRESSURIZATION AND VENT SYSTEM SCHEMATIC, AI-F18AC-460-500, WP011 00.
- AI-F18AC-460-500, WP011 00.
- (0) EXTERNAL FUEL SYSTEM SCHEMATIC, AI-FI8AC-460-500,
- ENGINE FUEL SUPPLY SYSTEM SCHEMATIC, AI-FI8AC-460-500, WP008 00.
- (2) RECORD FUNCTION SCHEMATIC, WP014 00.

- (3) POWER SCHEMATIC, WP005 00.
- (4) AVIONIC MUX CHANNEL | SCHEMATIC, AI-F18AC-741-500, WP004 00
- (5) AVOINIC MUX CHANNEL I SCHEMATIC, AI-F18AC-741-500, WP005 00.
- (6) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL
 DISPLAY INDICATOR IP-13177), RIGHT DIGITAL DISPLAY INDICATOR
 IP-1317(), HEAD-UP DISPLAY UNIT AN/AVG-28, HORIZONTAL INDICATOR
 IP-1350/A, AND ON F/A-188 THE REAR LEFT DIGITAL DISPLAY INDICATOR
 IP-1318(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318(), FOR MULTIPURPOSE
 DISPLAY GROUP, REFER TO A1-F186-745-500
- (8) FATIGUE STRAIN DATA SCHEMATIC, WP013 00.
- (9) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500,
- REAR COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500,
- DATA LINK SYSTEM VECTOR MODE 2-WAY OPERATION FUNCTIONAL SCHEMATIC. AI-F18AC-630-5107(C). WP012 04.
- 22 DELETED .
- F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 24> F/A-18B.
- 25> F/A-18A.
- 26 F/A-18A 161520 AND UP.
- 27) AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FIBAC-741-500, WP005 00
- 28) 163119 AND UP: ALSO 161353 THRU 161924 BEFORE F18 1AFC 056, OR 161353 THRU 163118 AFTER F18 AFC 70.
- 29) 161353 THRU 161924 AFTER F18 1AFC 056, OR 161353 THRU 163118 BEFORE F18 AFC 70.
- 30 161353 THRU 161987 BEFORE F18 AFC 48.
- []) 162394 AND UP, ALSO 161353 THRU 161987 AFTER F18 AFC 48.
- (22) DISPLAY REF CODES ARE NOT SHOWN IF DISPLAY MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ON WORKE THAN ONE INDICATOR, TROUBLESHOOT USING AI-FIBA()-OLD-OOD INPUT REF CODES. IF MALFUNCTION EXISTS ONLY IN ONE INDICATOR TROUBLESHOOT BY DOING DISPLAY TEST. AI-FIBAC-745-200, WPO04 OO (F/A-16B) ON WPO05 OO (F/A-16B).
- 33> 161353 THRU 161761.
- 34> 161924 AND UP.
- 35 162445 AND UP.
- 36 F/A-18A 163092 AND UP.
- 37 F/A-18B 163104 AND UP.
- 38> 161353 THRU 162909.
- 39> 163092 AND UP.

Subject

F/A-18 AFC

F/A-18 AFC

253

292

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FUEL SYSTEM INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Fuel System Interface Schematic, Figure 1

USNR F/A-18 A+ Avionics

USMCR F/A-18 A+ Avionics

Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)

Upgrade, Incorporation of,

(ÉCP MDA-F/A-18-0583)

Record of Applicable Technical Directives					
Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks	
F/A-18 IAFC 056	27 Mar 85	Fuel System Components Replacement and System Inspection (ECP MDA-F/A-18-00158R1 and ECP MDA-F/A-18-100160)	15 Oct 83		
F/A-18 AFC 48	28 Feb 90	Automatic AC Bus Isolation, Incorporation of (ECP-MDA-F/A-18-00121R1)	1 Sep 86		
F/A-18 AFC	31 Dec 90	Motive Flow Fuel Boost Pump Pressure Switch Installation of (ECP MDA-F/A-18-00158R2)	15 Oct 87		

Page No.

1 Dec 00

1 Dec 00

Figure 1.

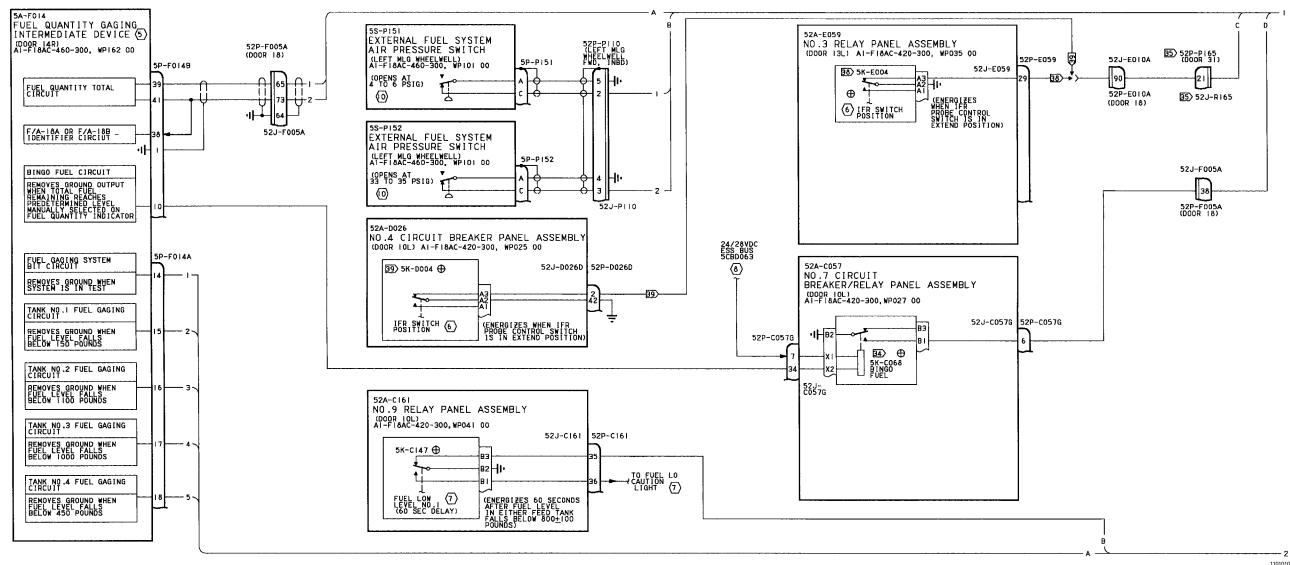


Figure 1. Fuel System Interface Schematic (Sheet 1)

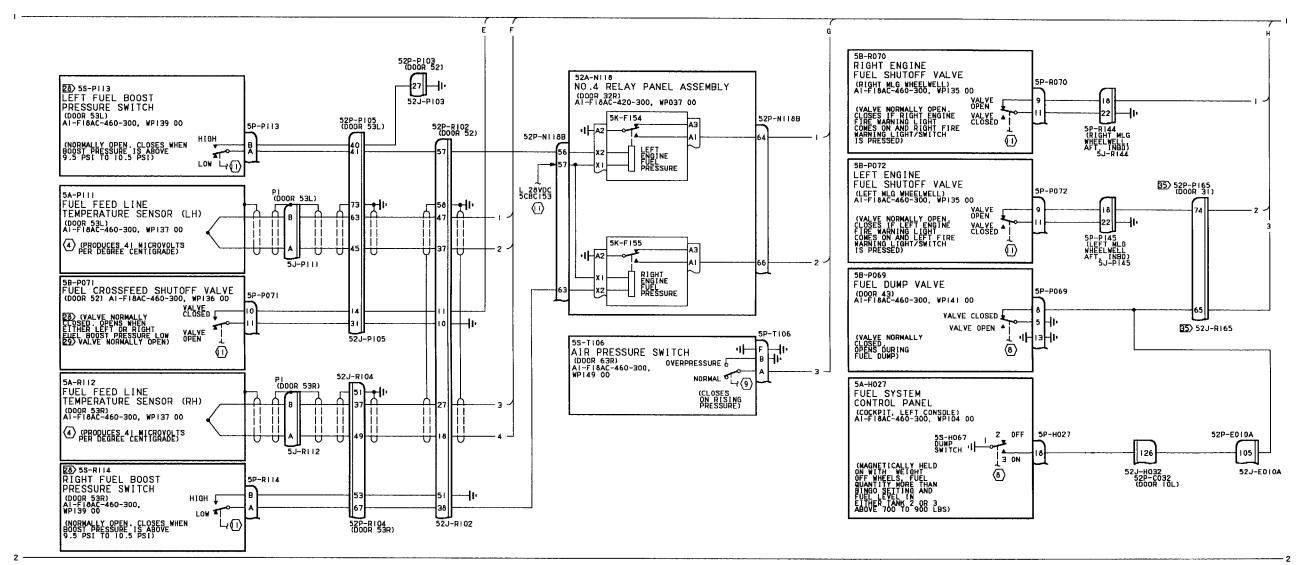
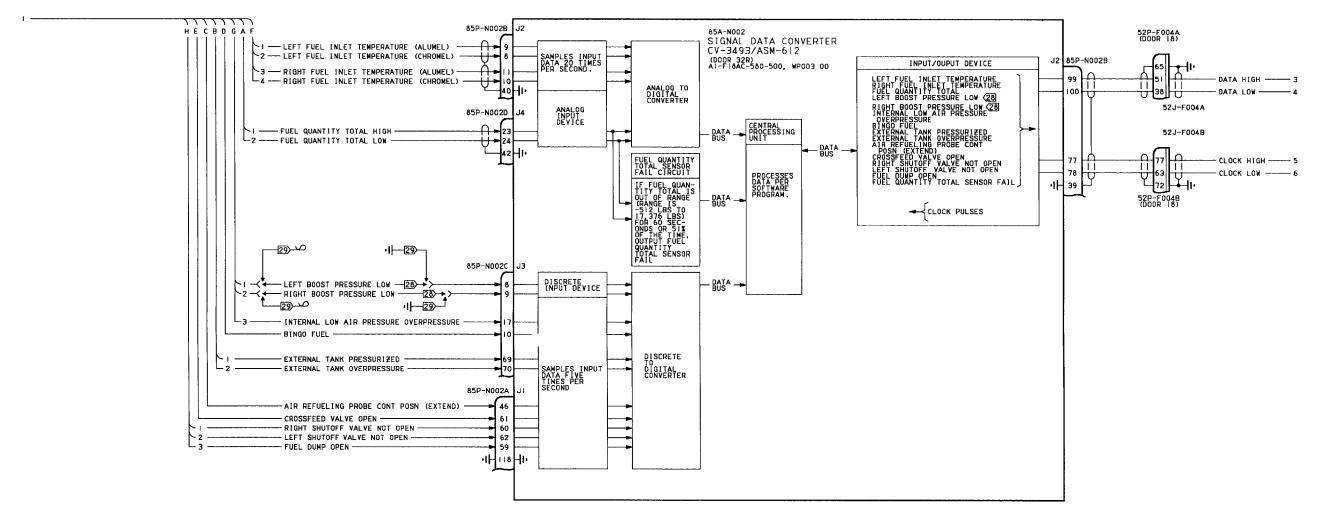


Figure 1.

Figure 1. Fuel System Interface Schematic (Sheet 2)



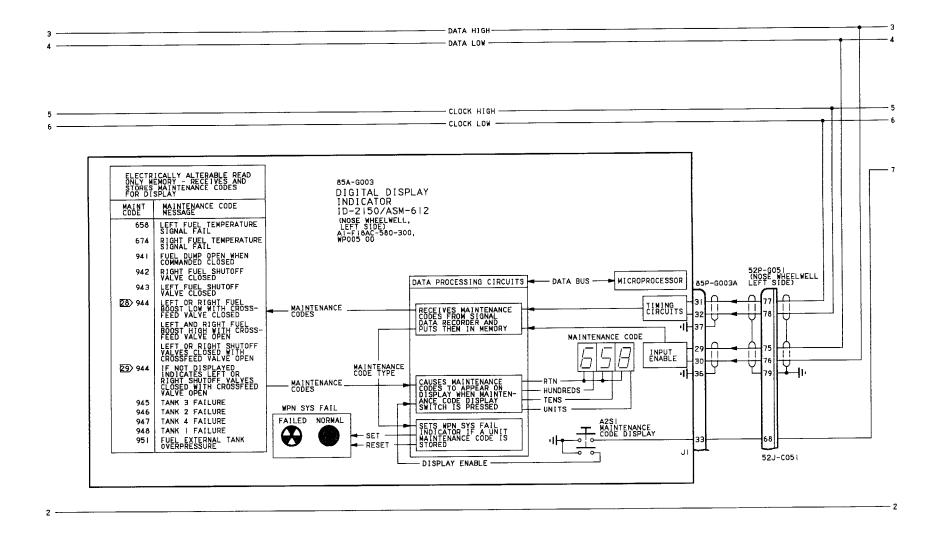


Figure 1. Fuel System Interface Schematic (Sheet 4)

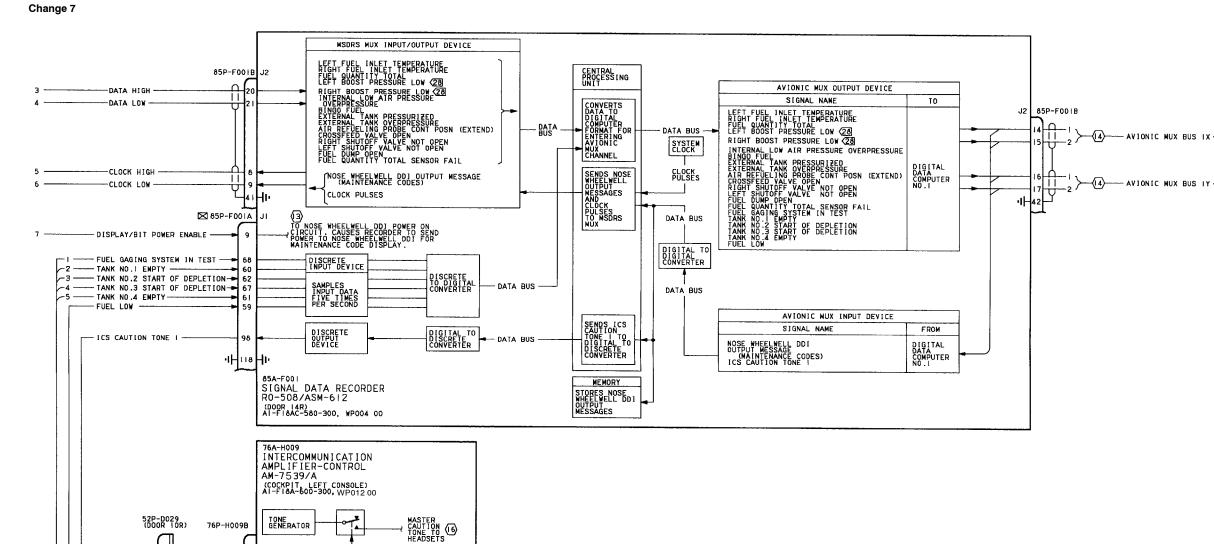


Figure 1.

52J-J029

Figure 1. Fuel System Interface Schematic (Sheet 5)

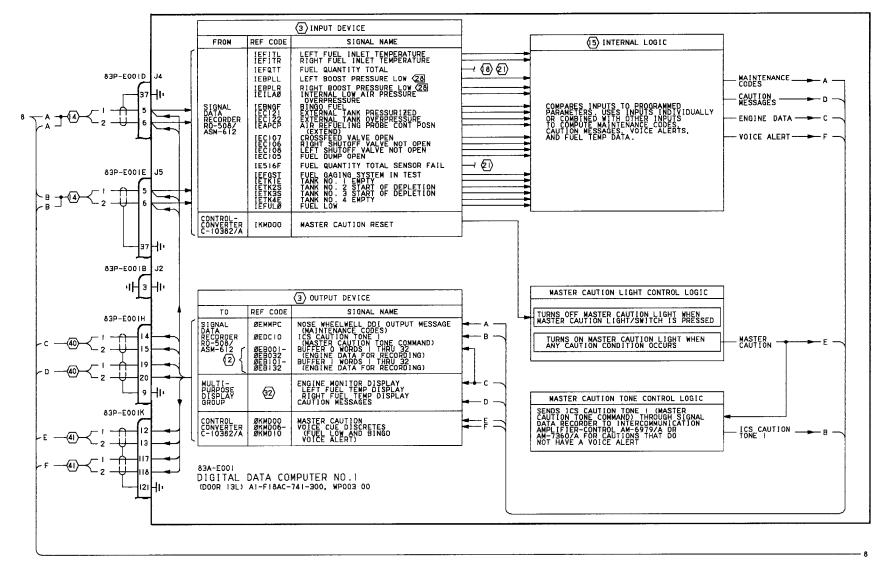


Figure 1. Fuel System Interface Schematic (Sheet 6)

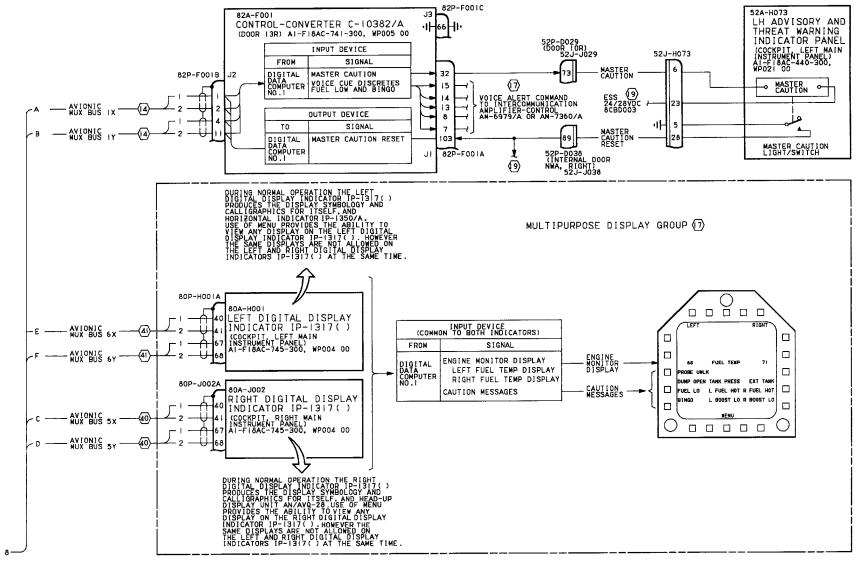


Figure 1. Fuel System Interface Schematic (Sheet 7)

- D. WHEN TESTING CONTINUITY, TEST FOR:
 - SHORTS TO GROUND. SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS. SHORTS BETWEEN SHIELD AND CONDUCTORS. SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF. 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY) MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
- DIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT,
- ☑ IDENTIFIES 24 VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE 1.



- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO ALL-FIBAC FIM-100.
- 4 HOT FUEL RECIRCULATION SYSTEM SCHEMATIC,
- (5) INTERNAL FUEL TRANSFER SYSTEM SCHEMATIC, A1-F18AC-460-500,
- (6) INFLIGHT REFUELING SYSTEM SCHEMATIC, AI-FIBAC-460-500, WP005 00.
- 7 FUEL QUANTITY LOW LEVEL WARNING SYSTEM SCHEMATIC, A1-F18AC-460-500, WP013 00.
- 8 FUEL DUMP SYSTEM SCHEMATIC, AI-FIBAC-460-500,
- 9 FUEL PRESSURIZATION AND VENT SYSTEM SCHEMATIC, AI-F18AC-460-500, WPOII 00.
- (0) EXTERNAL FUEL SYSTEM SCHEMATIC, AI-FI8AC-460-500,
- ENGINE FUEL SUPPLY SYSTEM SCHEMATIC, AI-FIBAC-460-500,
- (2) RECORD FUNCTION SCHEMATIC, WP014 00.

- (3) POWER SCHEMATIC, WP005 00.
- (4) AVIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00
- (5) AVOINIC MUX CHANNEL 2 SCHEMATIC, AI-F18AC-741-500, WP005 00
- (6) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-600-500, WP013 00.
- (T) THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY WITT AN/AVG-28, HORIZONTAL INDICATOR IP-1350/A, FOR MULTIPURPOSE DISPLAY GROUP, REFER TO AI-F18AC-745-500
- (8) FATIGUE STRAIN DATA SCHEMATIC, WP013 00.
- (9) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- DELETED .
- DATA LINK SYSTEM VECTOR MODE 2-WAY OPERATION FUNCTIONAL SCHEMATIC, AI-F18AC-630-510/(C), WP012 04.
- 22 DELETED .
- 23 DELETED .
- 24 DELETED
- 25> DELETED
- 26 DELETED
- (27) DELETED .
- 28) 163119 AND UP: ALSO 161353 THRU 161924 BEFORE F18 1AFC 056, OR 161353 THRU 163118 AFTER F18 AFC 70.
- 29 161353 THRU 161924 AFTER F18 IAFC 056, OR 161353 THRU 163118 BEFORE F18 AFC 70.
- 00 DELETED
- DELETED .
- DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS, TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS, TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ONLY ON ONE INDICATOR TROUBLESHOOT BY DOING DISPLAY TEST. ALFIBAC-145-200, WPO04 00 (F/A-18A).
- 33 DELETED
- 34) DELETED
- 35> 162445 AND UP
- DELETED .
- 37 DELETED.
- 38> 161353 THRU 162909
- 39 163092 AND UP
- (40) AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-FI8AC-741-500, WP018 00.
- (41) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - BUILT-IN TEST

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR AFC 292, AND F/A-18B

This WP supersedes WP012 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No.
Built-In Teat Schematic, Figure 1	2

Record of Applicable Technical Directives

None

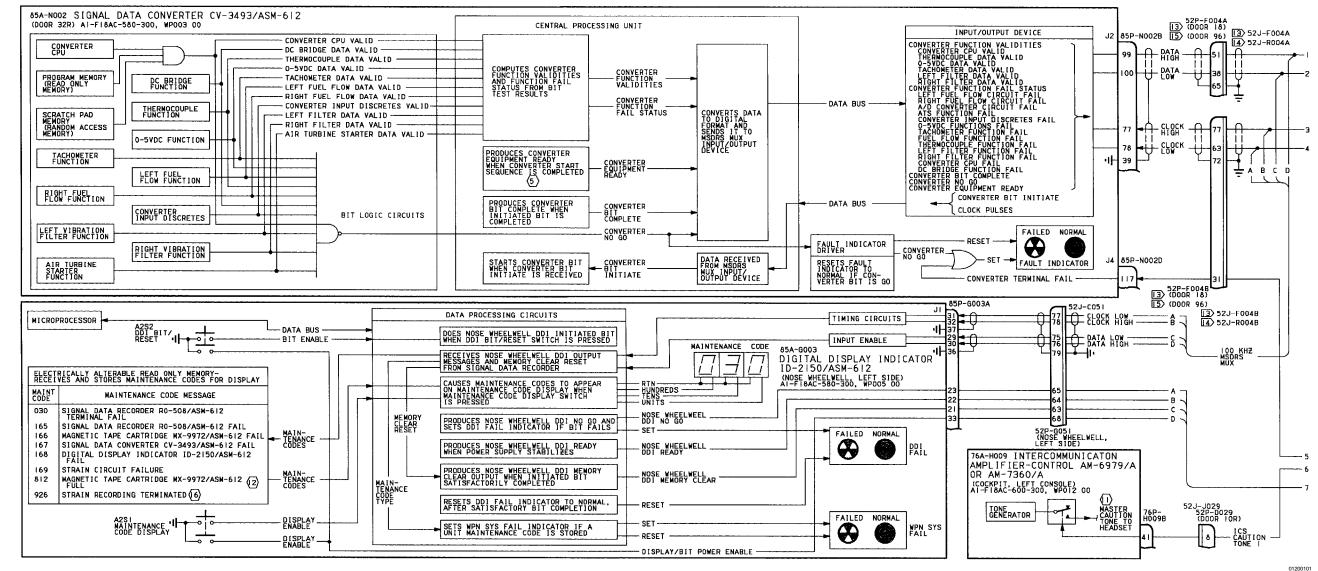


Figure 1. Figure 1. Built-In Test Schematic (Sheet 1)

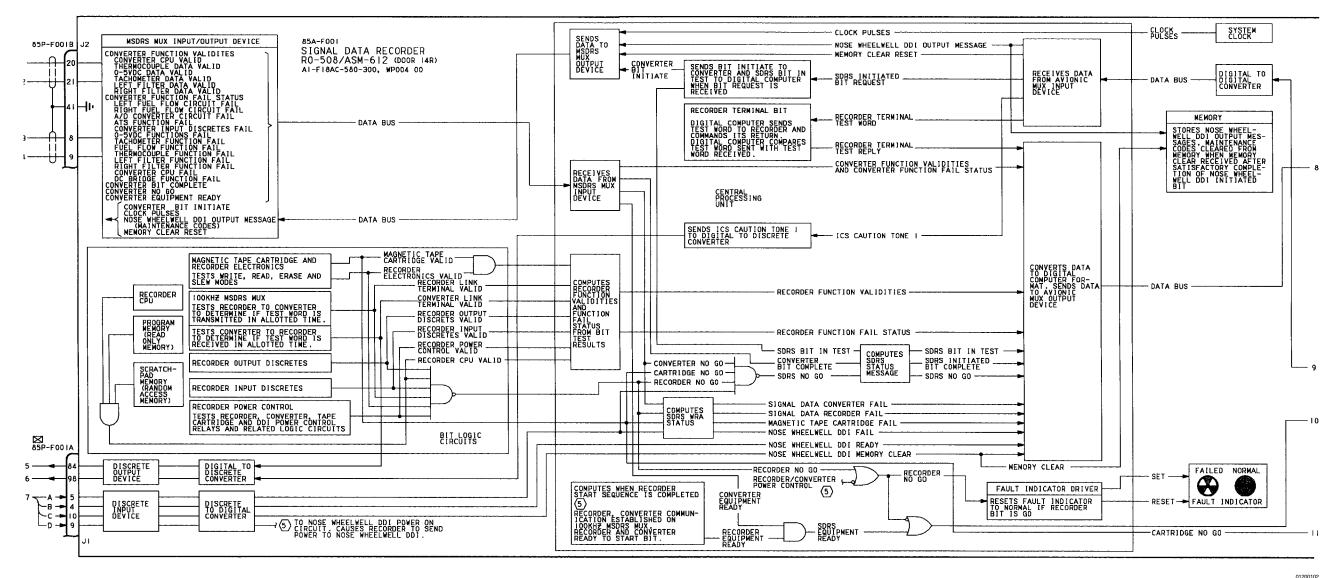


Figure 1.

Figure 1. Built-In Test Schematic (Sheet 2)

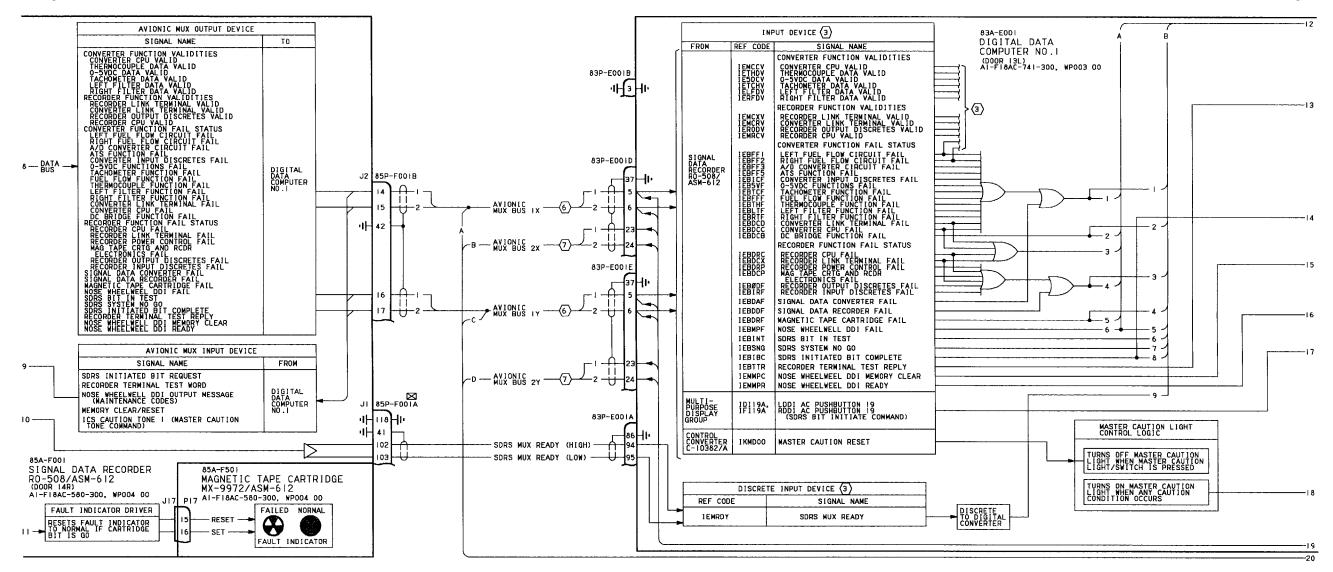


Figure 1. Built-In Test Schematic (Sheet 3)

Figure 1.

01200103

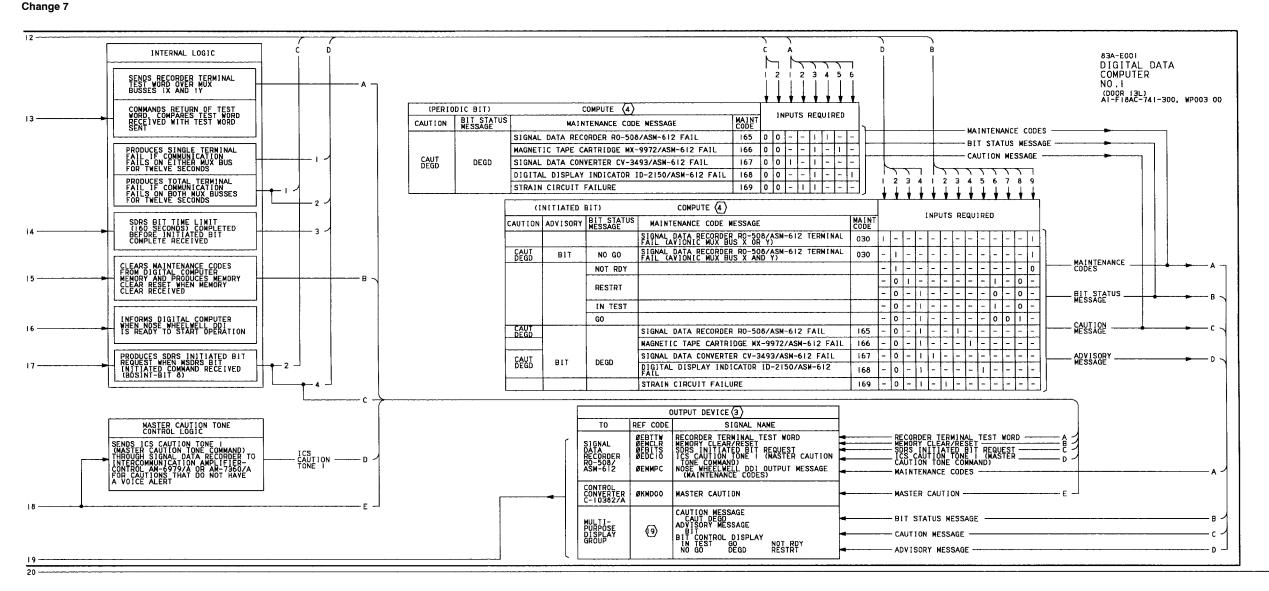


Figure 1. Built-In Test Schematic (Sheet 4)

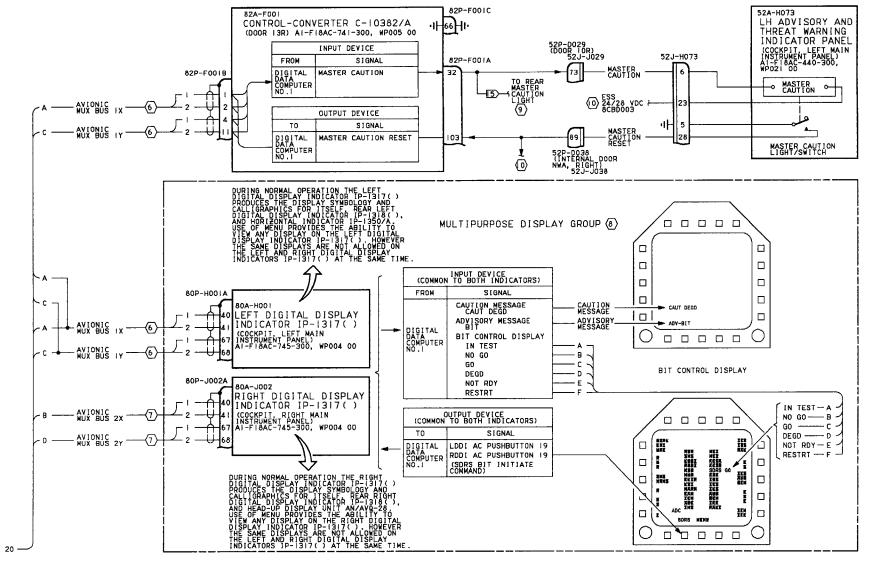
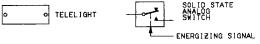


Figure 1.

Figure 1. Built-In Test Schematic (Sheet 5)

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY #) IN REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW REIAY.
 - C. DO NOT TEST LOW LEYEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MITTHETE ON RXI SCALE PRINT TO PINTERSTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (I) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY DE). MAKE SURE NULTIMETER LEADS/JUMPER MIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - DIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 1.



- \boxtimes IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.
- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AI-FIBA()-OLD-000. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100.
- 4 EXPLANATION OF MATRIX
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED.
 - (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
 - (3) DASH (-) INDICATES THE DUTPUT DOES NOT DEPEND ON THIS INPUT

- (5) POWER SCHEMATIC, WP005 00.
- S) AVIONIC MUX CHANNEL | SCHEMATIC, AI-F18AC-741-500, WP004 00.
- (7) AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FIBAC-741-500, WP005 0
- (8) THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY HEAD OF THE LEFT DIGITAL DISPLAY HEAD OF THE LEFT DIGITAL DISPLAY HEAD OF THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1350/A, AND ON F/A-188 THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318 () REAR DISPLAY INDICATOR IP-1318 () REAR DISPLAY INDICATOR IP-1318 () REFER TO ALF-1816-745-500.
- 9) REAR COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WPOO7 00.
- 0) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-F18AC-440-500, WP006 00.
- INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- (2) RECORD FUNCTION SCHEMATIC, WP014 00.
- (3) F/A-18A.
- [4] F/A-188 PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- [5] F/A-18B.
- (6) FATIGUE STRAIN DATA SCHEMATIC, WP013 00.
- DELETED.
- B DELETED.
- DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS, TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ON MORE THAN ONE INDICATOR, TROUBLESHOOT USING AI-FIBA()-OLD-OOO INPUT REF CODES. IF MALFUNCTION EXISTS ONLY ON ONE INDICATOR TROUBLESHOOT BY DOING DISPLAY TEST, AI-FIBAC-745-200, WP004 DO (F/A-18A) OR WP005 DO (F/A-18B).

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - BUILT-IN TEST

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No
Built-In Teat Schematic Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	1
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

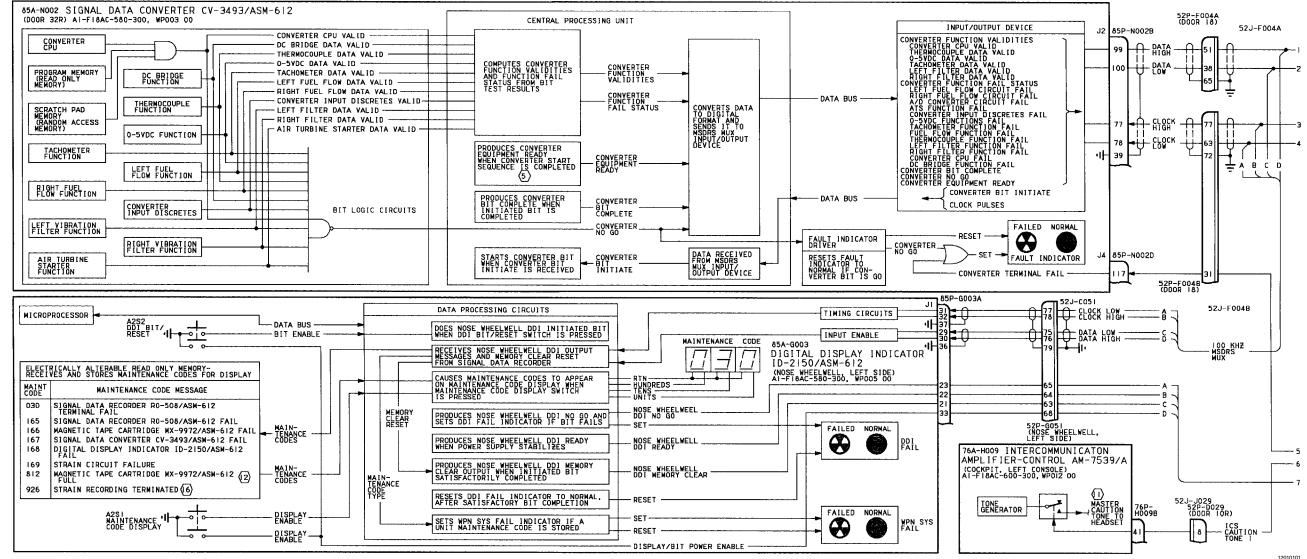


Figure 1. Figure 1. Built-In Test Schematic (Sheet 1)

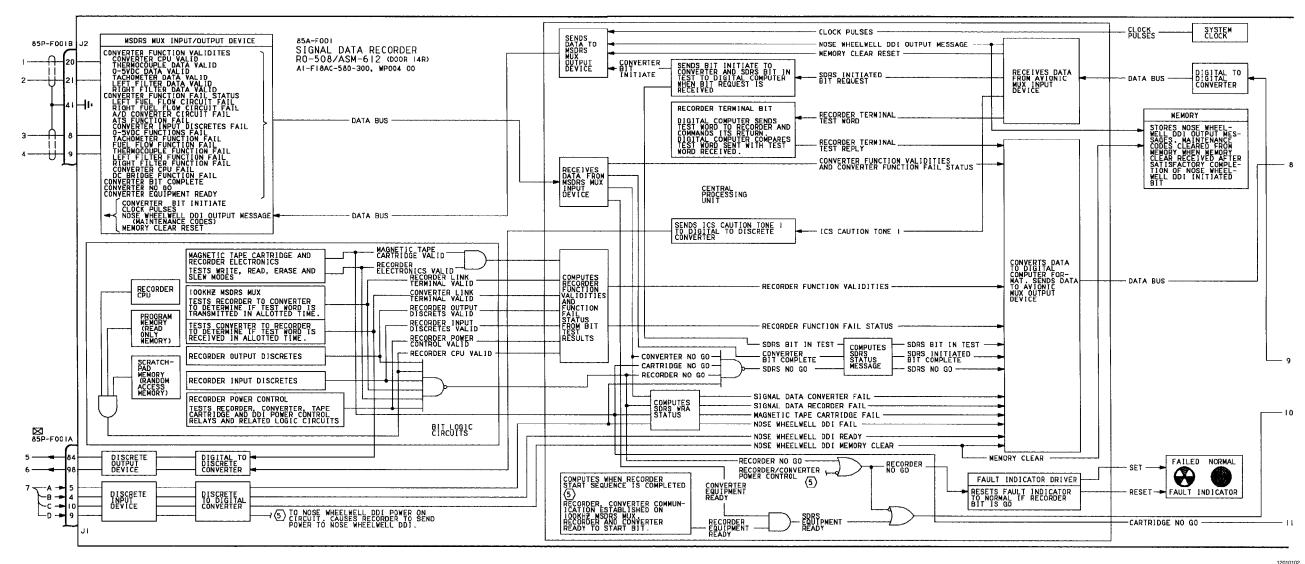


Figure 1.

Figure 1. Built-In Test Schematic (Sheet 2)

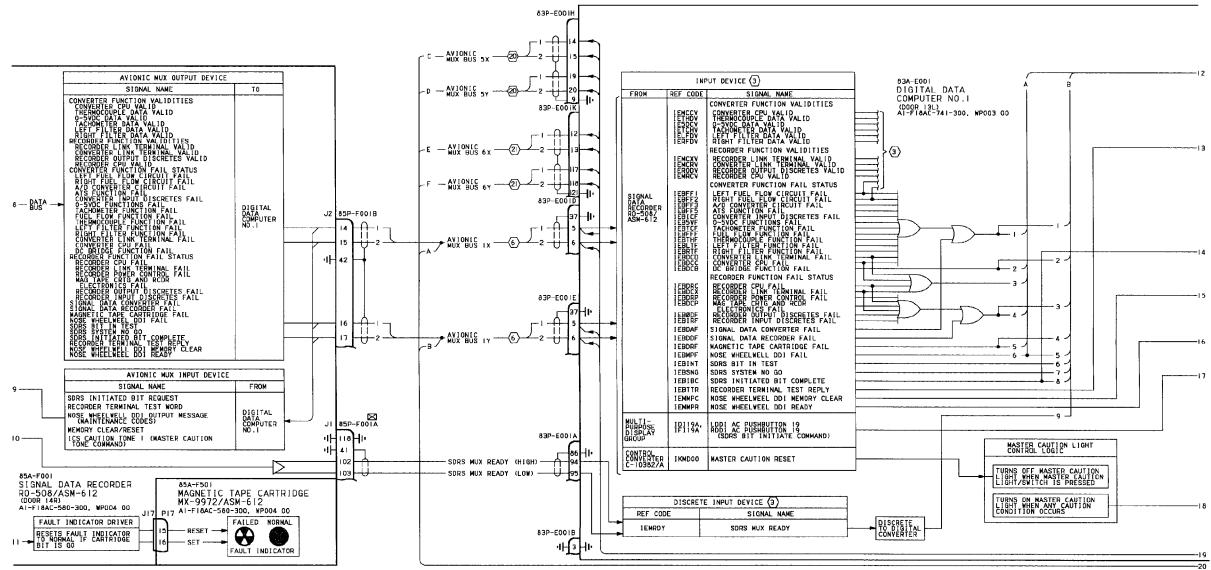
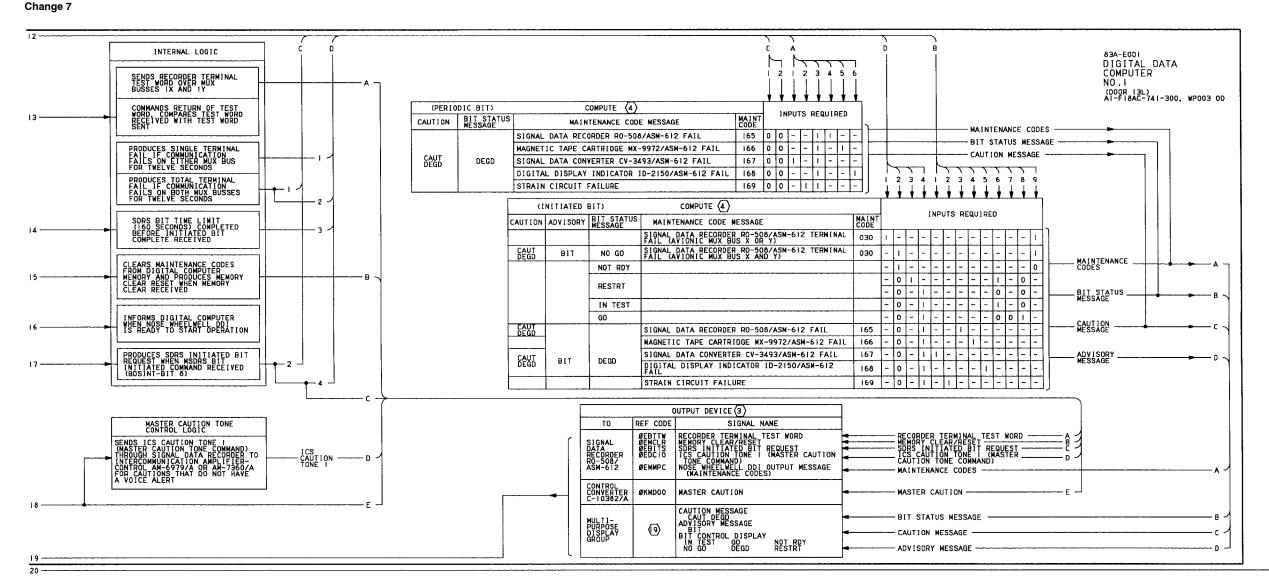


Figure 1. Figure 1. Figure 1. Built-In Test Schematic (Sheet 3)

12010103



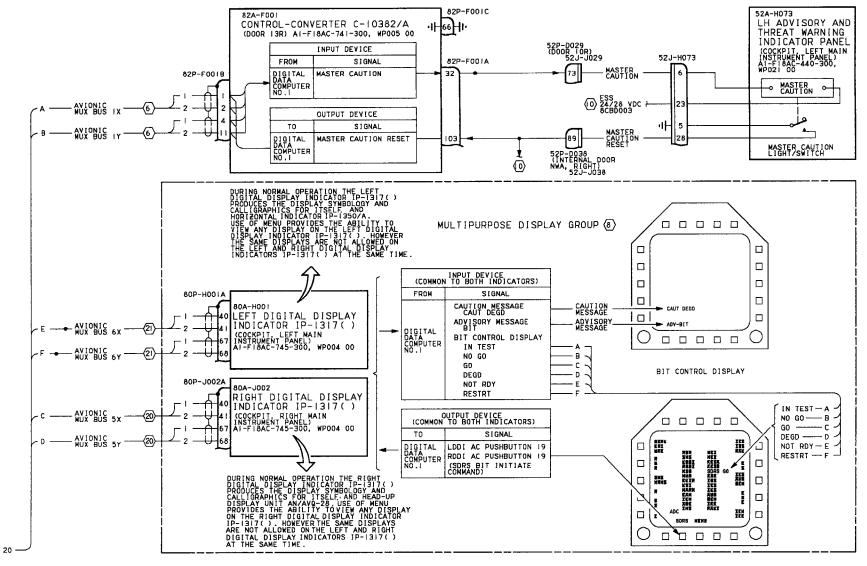
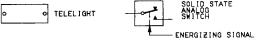


Figure 1. Built-In Test Schematic (Sheet 5)

Figure 1.

LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FISA()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING FRLAY (IDENTIFIED BY (#)) IN REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY, AND SOCKET FOR CORRECT FOR STATE OF THE LAY AND SOCKET FOR CORRECT FOR STATE OF THE LAY IN THE LAY SWITCHING RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEYEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MITTER ON RXI STACTS PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (I) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY) MAKE SUBE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - ⊕ IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE |.



- ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE |.
- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AI-FIBA()-OLD-000. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100.
- EXPLANATION OF MATRIX
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED.
 - (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
 - (3) DASH (-) INDICATES THE DUTPUT DOES NOT DEPEND ON THIS INPUT

- POWER SCHEMATIC, WP005 00.
- (E) AVIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY UNIT ANYAVQ-28, HORIZONTAL INDICATOR IP-1350/A. FOR MULTIPURPOSE DISPLAY GROUP, REFER TO AI-F18AC-745-500.
- COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FI8AC-440-500, WP006 00.
- INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC AI-F18AC-600-500, WP013 00.
- (2) RECORD FUNCTION SCHEMATIC, WP014 00.
- (3) DELETED.
- 14) DELETED.
- DELETED .
- (6) FATIGUE STRAIN DATA SCHEMATIC, WP013 00
- DELETED .
- DELETED.
- DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS TRANSFER DISPLAY TO MOTHER NOT AND ICATOR IF MALFUNCTION EXISTS OF THAN ONE INDICATOR, TROUBLESHOOT USING AI-FLAK, OLD COONINDIT REF CODES IF MALFUNCTION EXISTS ONLY ON DISPLAY DOING DISPLAY TEST, AI-FLAKOC-745-200, WPOO4 DO.
- AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-FIBAC-741-500, WP018 00
- AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00

Change 4 - 15 October 1987

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FATIGUE STRAIN DATA

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes WP013 00, dated 1 May 1986.

Title	WP Number
Fatigue Strain Data Schematic - WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 87X AND UP	013 01
Fatigue Strain Data Schematic - WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 210	013 02
Fatigue Strain Data Schematic - WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 85A+	013 03

Change 4 - 15 October 1987

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FATIGUE STRAIN DATA

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 87X AND UP

Reference Material

None

Alphabetical Index

Subject	Page No.
Fatigue Strain Data Schematic, Figure 1	2

Record of Applicable Technical Directives

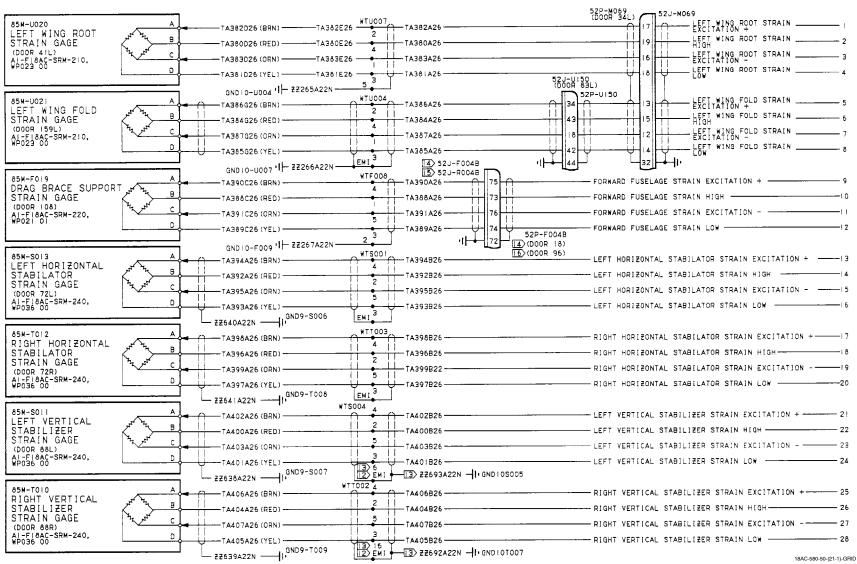


Figure 1. Fatique Strain Data Schematic (Sheet 1)

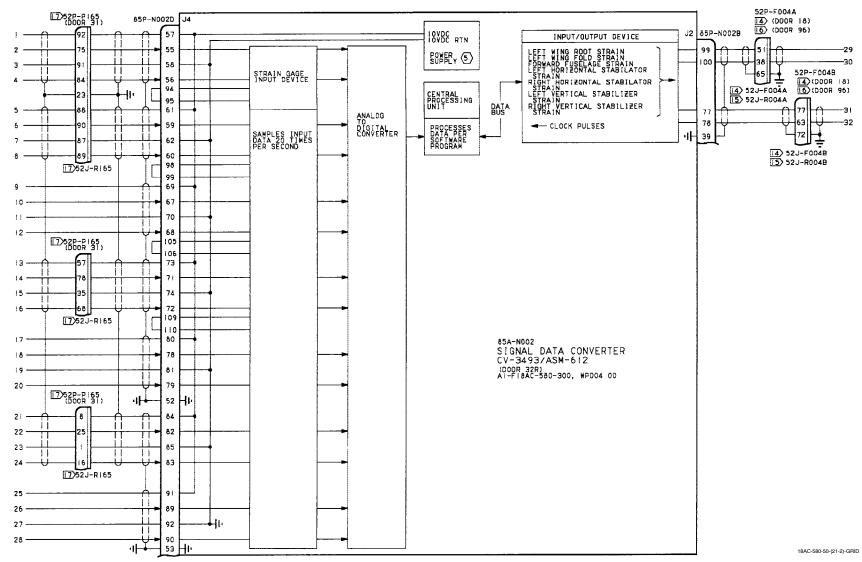
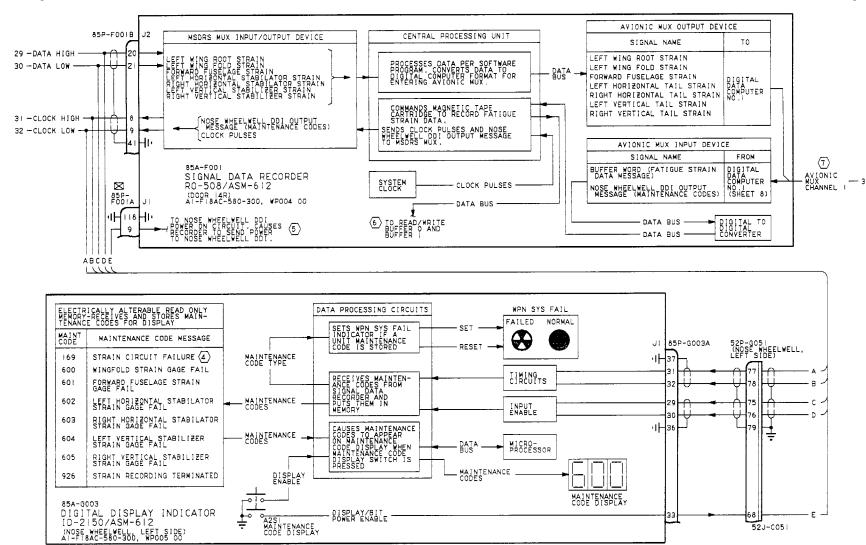
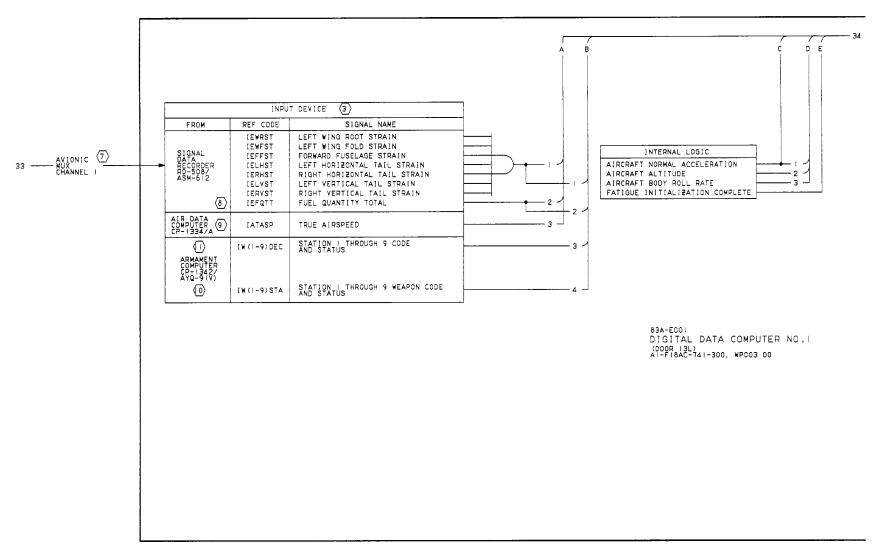


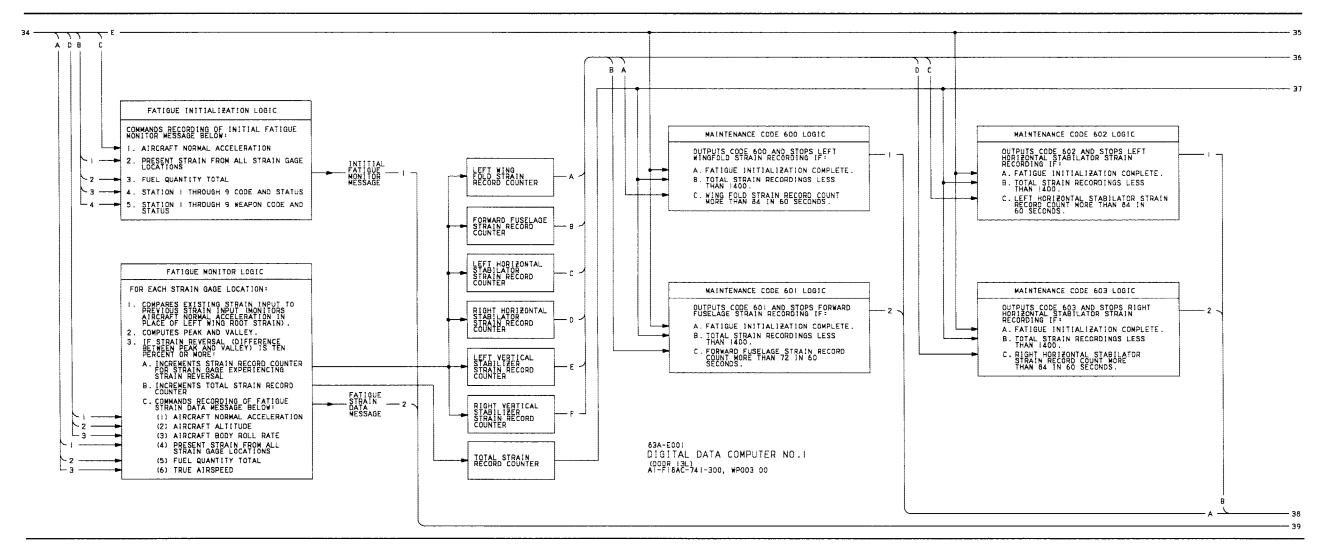
Figure 1.

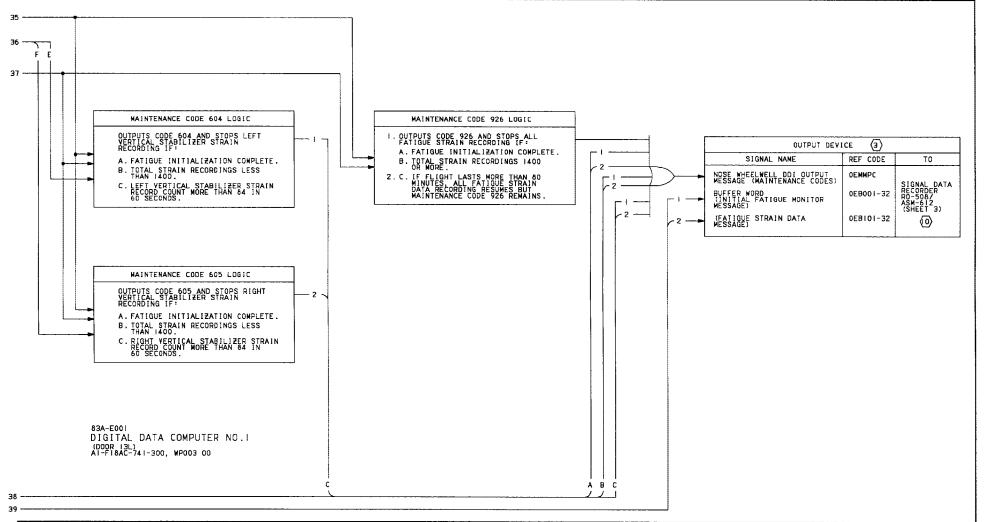
Figure 1. Fatigue Strain Data Schematic (Sheet 2)



18AC-580-50-(21-3)-GRID







LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FIBA()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY A) IS REMOVED FOR TROUBLESHOOTING JOENTIFY RELAY AND SOCKET FOR CORRECT REINSTALATION DO NOT REPLACE OWNEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RXI SCALE. PIN TO PIN TEST THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING FOR CONTINUITY, TEST FOR:
 - (I) . SHORTS TO GROUND
 - (2). SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS
 - (3). SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) . SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY 🔀). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
 - → INDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT SEE NOTE:
- ☐ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS
 OF THE CONNECTOR. SEE NOTE !...
 FOR LOGIC DIAGRAMS BELATING TO BEE CODE. BEEFED TO
- FOR LOGIC DIAGRAMS RELATING TO REF CODE REFER TO THE PROPERTY OF THE MEMORY INSPECTOR ACCESS RELATING TO REFER TO ALL FIRAC FIN-100.
- 4 BUILT-IN TEST SCHEMATIC, WP012 00.
- (5) POWER SCHEMATIC, WP005 00.
- (6) RECORD FUNCTION SCHEMATIC, WPO14 00.
- 7 AVIONIC MUX CHANNEL I SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- (8) FUEL SYSTEM SCHEMATIC, WPO++ 00.
- AIR DATA COMPUTER SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-560-500, MP004 00.
- (0) AIM-9 SIDEWINDER AVIONIC INTERFACE SCHEMATIC, AI-F18AC-740-510, WP036 00.
- (1) BOMB AVIONIC INTERFACE SCHEMATIC, AI-FIBAC-740-510, WP048 00.
- (2) 161925 THRU 162414.
- [3] 162415 AND UP.
- 14 F/A-18A.
- (5) F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 16) F/A-18B.
- 17 162445 AND UP.

18AC-580-50-(21-6)-GRID Figure 1. Change 4 - 15 October 1987

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FATIGUE STRAIN DATA

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 210

Reference Material

None

Alphabetical Index

Subject	Page No
Fatigue Strain Data Schematic, Figure 1	2

Record of Applicable Technical Directives

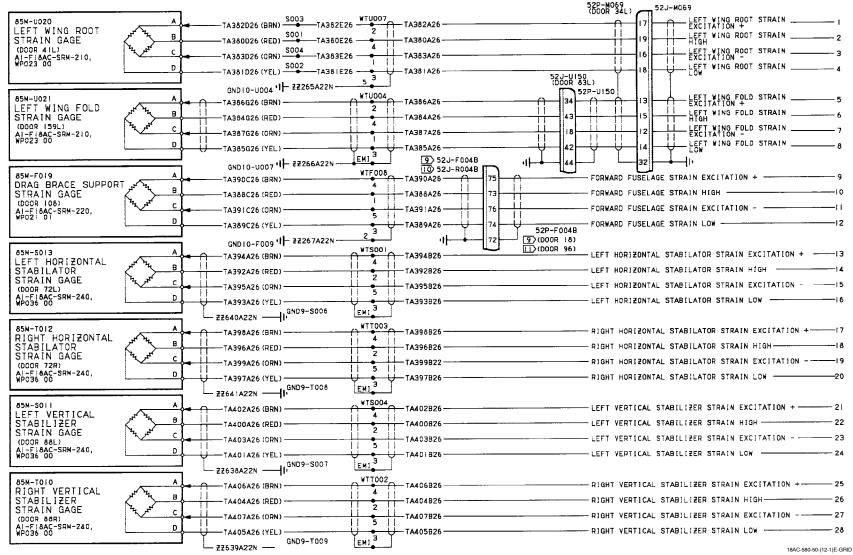


Figure 1.

Figure 1. Fatigue Strain Data Schematic (Sheet 1)

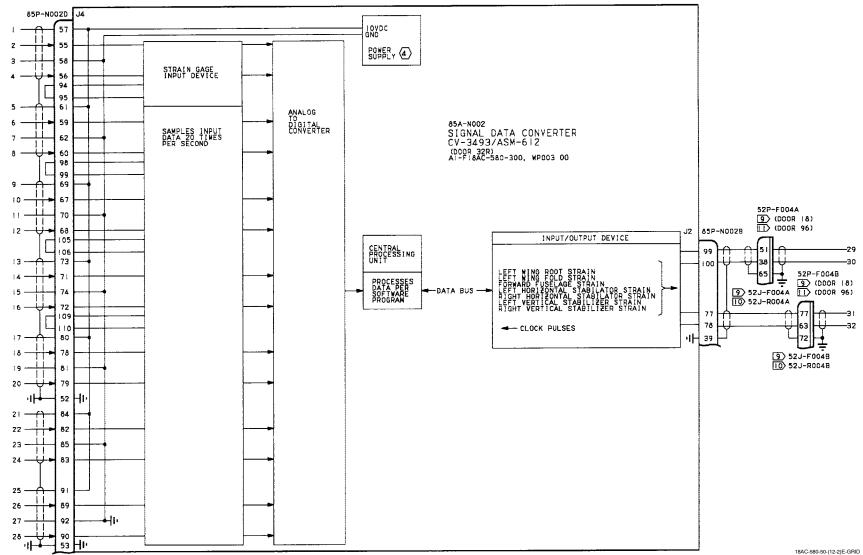
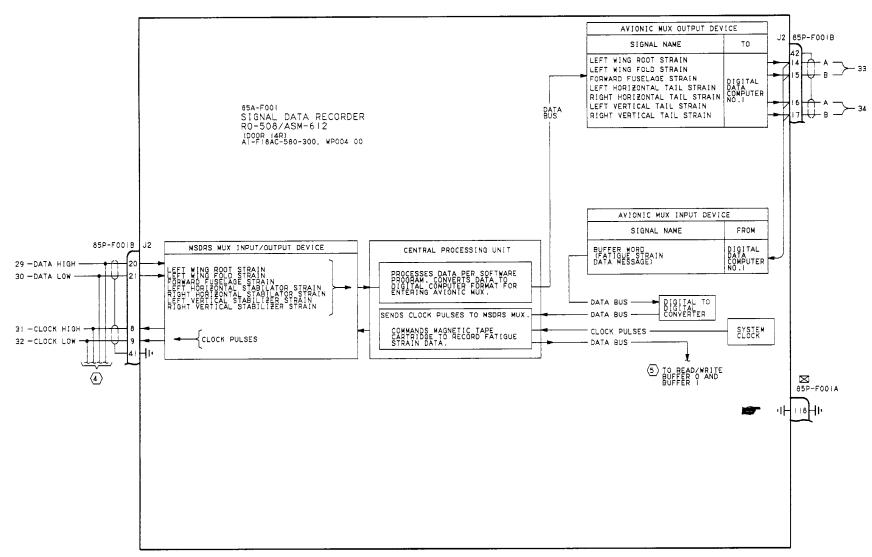


Figure 1.

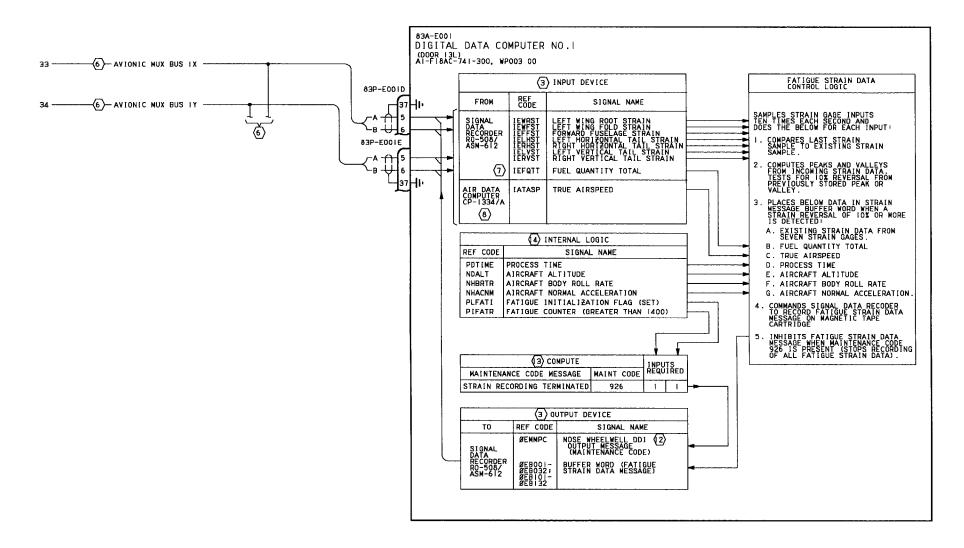
Figure 1. Fatigue Strain Data Schematic (Sheet 2)

Figure 1.



18AC-580-50-(12-3)F-GRID Figure 1.

A1-F18AC-580-500 Change 4



LEGEND

- I. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FIBA()-WDM-000.
- B. WHEN A LOW LEYEL CURRENT SWITCHING RELAY (IDENTIFIED AND SOCKET FOR CORRECT REINSTALLATION, DO NOT REPLACE LOW LEYEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY IF RELAY IS BEFETING RELAY WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS)
 FOR CONTINUITY WITH MULTIMETER ON RXI SCALE. PIN TO
 PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY
 CONTACTS MAY USE THE RXI SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (I). SHORTS TO GROUND.
- (2). SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3). SHORTS BETWEEN SHIELD AND CONDUCTORS.
- (4) . SHIELD CONTINUITY .
- . WHEN ELECTRICAL POWER IS OFF. 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY) AMAKE SURE MULTETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
- IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT.
 SEE NOTE 1.
- ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.
- (3) FOR LOGIC DIAGRAMS FELATING TO REE CODE REFER TO MEMORY INSPECTOR ACCESS RELATING TO REFE CODE, REFER TO ALFIBACE FIM-100.
- (4) POWER SCHEMATIC, WP005 00.
- (5) RECORD FUNCTION SCHEMATIC, WP014 00.
- 6 AVIONIC MUX CHANNEL I SCHEMATIC, AI-FIBAC-741-500, MP004 00.
- 7 FUEL SYSTEM INTERFACE SCHEMATIC, WPOIL 00.
- B AIR DATA COMPUTER SYSTEM SCHEMATIC, A1-F18AC-560-500, WP004 00.
- 9 F/A-18A.
- F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- (2) BUILT-IN TEST SCHEMATIC, WP012 00.
- (3) EXPLANATION OF MATRIX
- A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.

 B. IMPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUTS.
- C. SIGNAL OUTPUT IS READ HORIZONTALLY, EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
- D. INTERPRET MATRIX TABLE AS INDICATED:
- (I) ONE (I) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
- (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
- (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.
- (4) REF CODES USED FOR THESE COMPUTATIONS ARE MISSION COMPUTER INTERNAL REF CODES IN A1-F18AC-0LD-000, USE THE LOGIC DIAGRAMS FOR THE INPUT/OUTPUT REF CODES.

18AC-580-50-(12-4)F-GRID

Figure 1. Fatigue Strain Data Schematic (Sheet 4)

Change 4 - 15 October 1987

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FATIGUE STRAIN DATA

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 85A+

Reference Material

None

Alphabetical Index

Subject	Page No
Fatigue Strain Data Schematic, Figure 1	 2

Record of Applicable Technical Directives

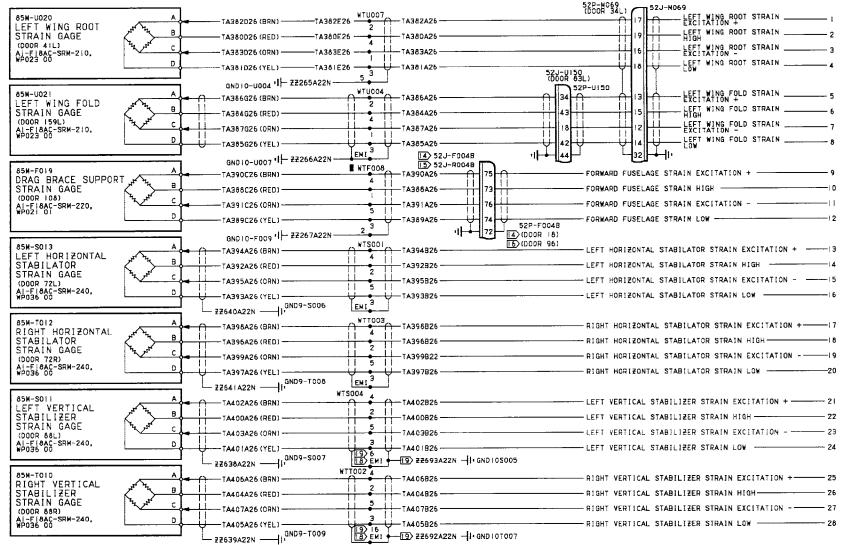


Figure 1. Fatigue Strain Gage Schematic (Sheet 1)

18AC-580-50-(19-1)C-GRID

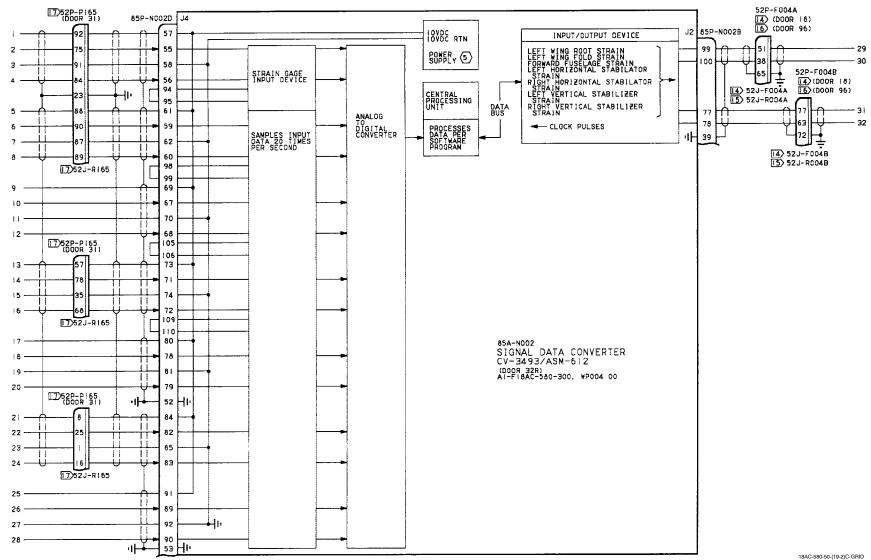
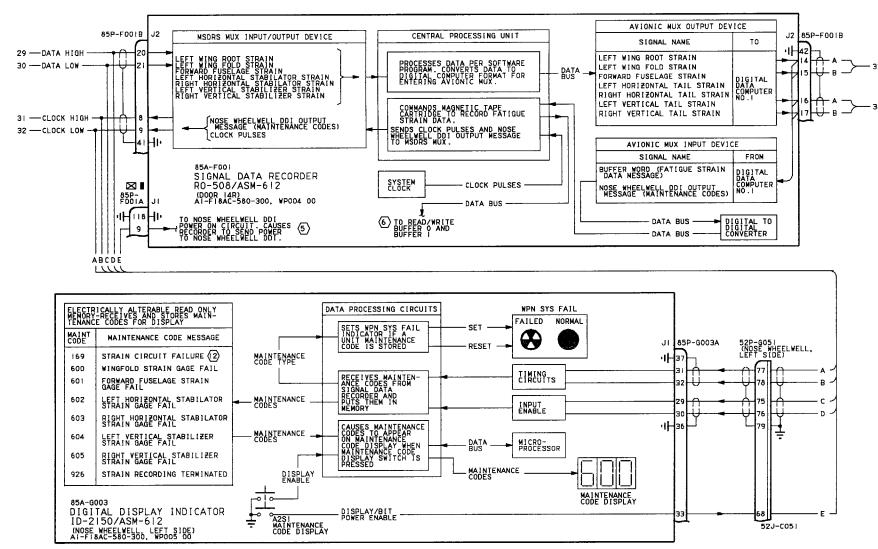


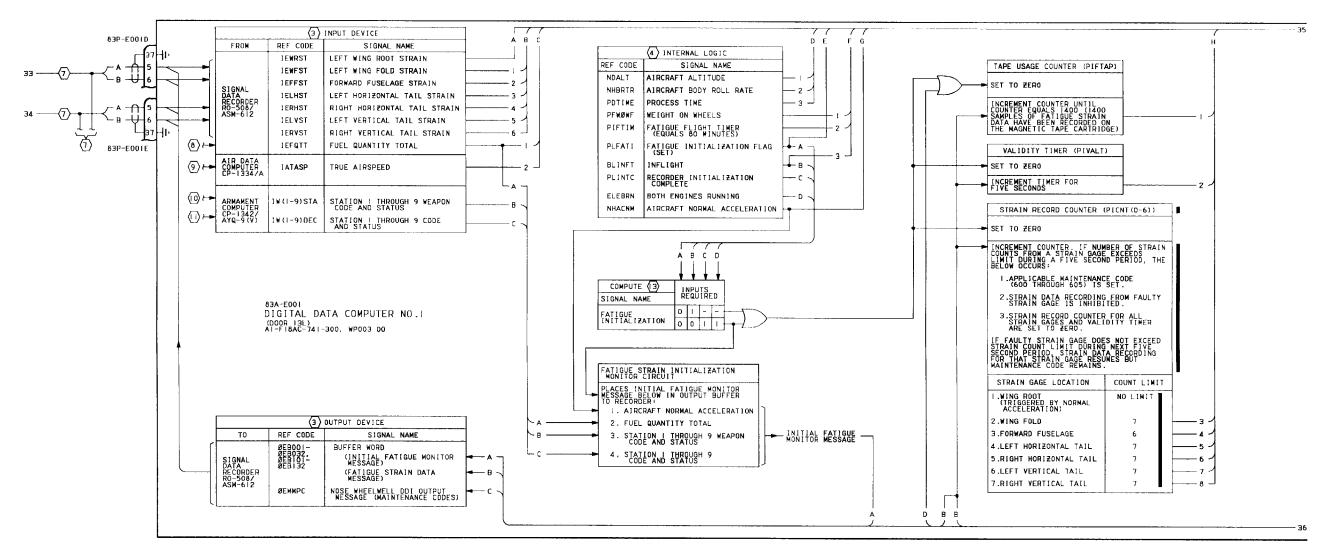
Figure 1.

Figure 1. Fatigue Strain Gage Schematic (Sheet 2)



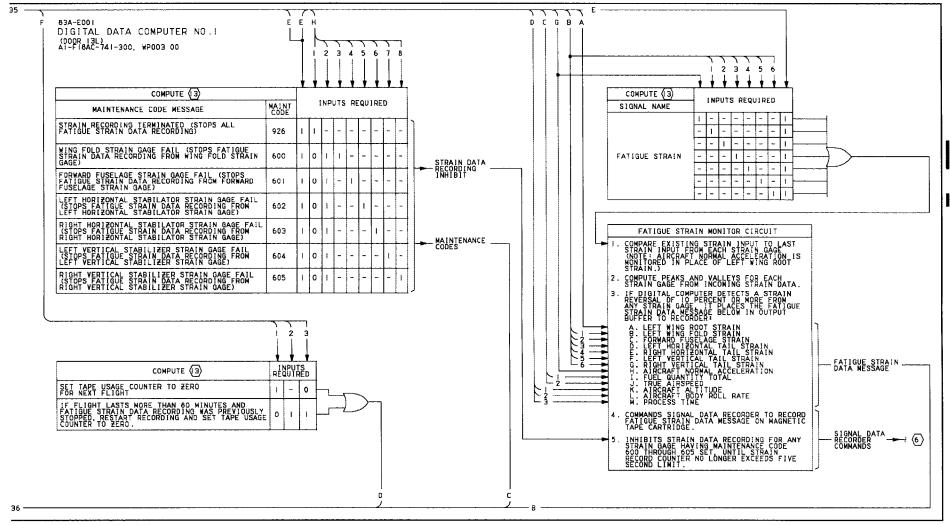
18AC-580-50-(19-3)D-GRID Figure 1.

Figure 1. Fatigue Strain Gage Schematic (Sheet 3)



18AC-580-50-(19-4)C-GRID **Figure 1.**

Figure 1. Fatigue Strain Gage Schematic (Sheet 4)



LEGEND

- I. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- SHUMN IN AITTIGA()-WUM-000.

 WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RXI SCALE, PIN TO PIN TEST THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (I). SHORTS TO GROUND
- (2), SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3) . SHORTS BETWEEN SHIELD AND CONDUCTORS .
- (4), SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☑). MAKE SURE MULTIMETER LEADS/JUNPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - → IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 1.
- ☑ IDENTIFIES 24YDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.
- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AI-FIBAC-OLD-000, FOR MEMORY INSPECT ACCESS RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100.
- (4) REF CODES USED FOR THESE COMPUTATIONS ARE MISSION COMPUTER INTERNAL REF CODES IN AI-FIBAC-OLD-000, USE THE LOGIC DIAGRAMS FOR THE INPUT/OUTPUT REF CODES.
- 5 POWER SCHEMATIC, WP005 00.
- (6) RECORD FUNCTION SCHEMATIC, WP014 00.
- (7) AVIONIC MUX CHANNEL I SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- (8) FUEL SYSTEM INTERFACE SCHEMATIC, WPOIL 00.
- AIR DATA COMPUTER SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-560-500, WP004 00.
- (D) AIN-9 SIDEWINDER AVIONIC INTERFACE SCHEMATIC, AI-FIBAC-740-510, WP036 00.
- (1) BOMB AVIONIC INTERFACE SCHEMATIC, A1-F18AC-740-510, WP048 GO.
- (2) BUILT-IN TEST SCHEMATIC, WPO+2 00.
- (3) EXPLANATION OF MATRIX
- A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
- INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
- SIGNAL OUTPUT IS READ HORIZONTALLY, EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
- D. INTERPRET MATRIX TABLE AS INDICATED:
- (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
- (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
- (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT
- 15 F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- (6) F/A-18B
- 17 162445 AND UP
- (8) 161925 THRU 162414
- (9) 162415 AND UP.

18AC-580-50-(19-5)D-GRID

Change 6 - 15 August 1992

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - RECORD FUNCTION

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

This WP supersedes WP014 00, dated 1 May 1986.

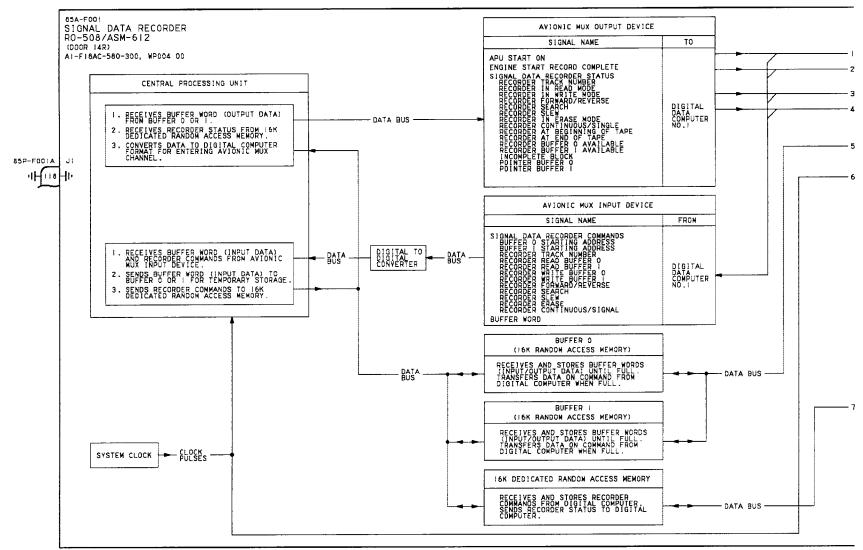
Reference Material

None

Alphabetical Index

Subject	Page No
Record Function Schematic, Figure 1	2

Record of Applicable Technical Directives



18AC-580-50-(18-1)B-GRID Figure 1.

Figure 1. Record Function Schematic (Sheet 1)

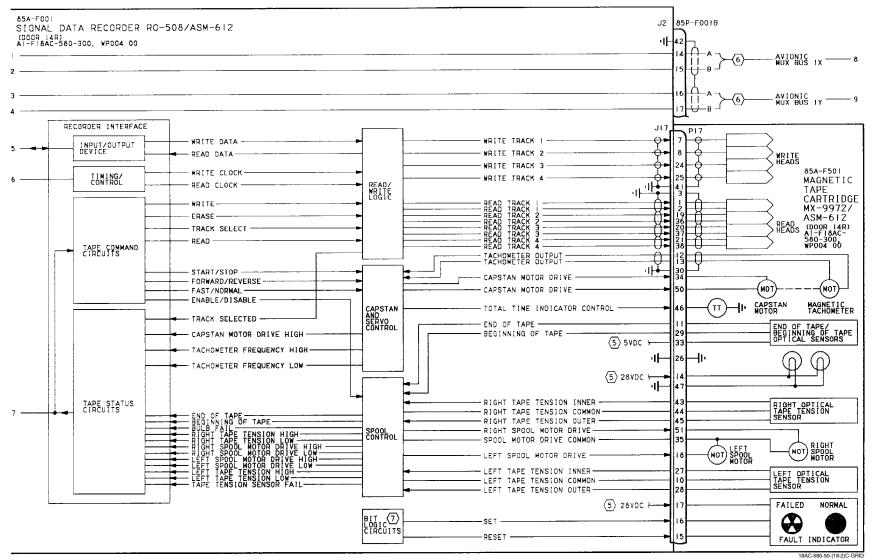
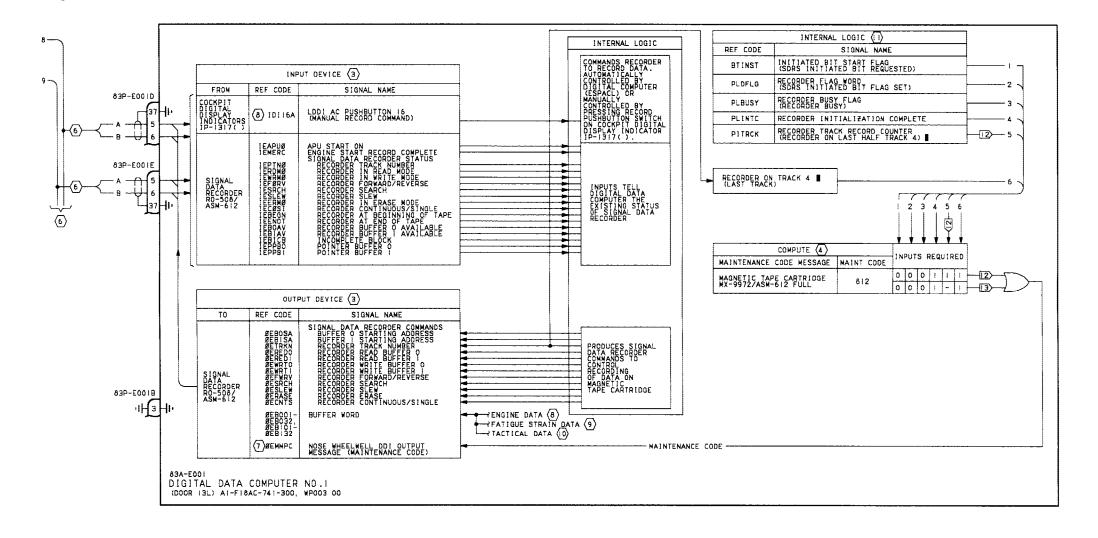


Figure 1. Figure 1. Record Function Schematic (Sheet 2)



418AC-580-50-(18-3)19-GRID **Figure 1.**

Figure 1. Record Function Schematic (Sheet 3)

LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-000.
 - B. WHEN A LOW FEWET CURRENT STITCHING RELAY (IDENTIFIED BY (#)).
 FOR CORRECT RELAYION THE NOT REPLACE LAY AND SOCIETY OF THE THE LAY OF THE LAY.
 - C. DO NOT TEST LOW LEYEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MOTHER ON RXI SCALE PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND. (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELDS AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
 - ⊕ IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 1.
- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AI-FIBAC-OLD-000. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO AI-FIBAC-FIN-100.
- (4) EXPLANATION OF MATRIX
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
 - C. SIGNAL DUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED.

 (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO

 (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE

 (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.
- POWER SCHEMATIC. WP005 00.
- AVIONIC MUX CHANNEL I SCHEMATIC, AI-FI8AC-741-500, WP004 00.
- BUILT-IN TEST SCHENATIC, WP012 00.
- LEFT ENGINE AND RIGHT ENGINE INTERFACE SCHEMATICS, WP009 00 AND WP0+0 00.
- FATIGUE STRAIN DATA SCHEMATIC, WP013 00.
- FOR LIST OF ALL TACTICAL DATA RECORDED, SEE AI-F18AC-580-100, WP004 00.
- REF CODES USED FOR THESE COMPUTATIONS ARE MISSION COMPUTER INTERNAL MEMORIES. TO ALLOCATE THE INPUT OUTPUT REF CODES.
- 12 WITH DIGITAL DATA COMPUTER NO. | CONFIG/IDENT NUMBER 210.
- 13> WITH DIGITAL DATA COMPUTER NO.1 CONFIG/IDENT NUMBER 84A AND UP.

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - ENVIRONMENTAL CONTROL SYSTEMS INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP015 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No
Environmental Control Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

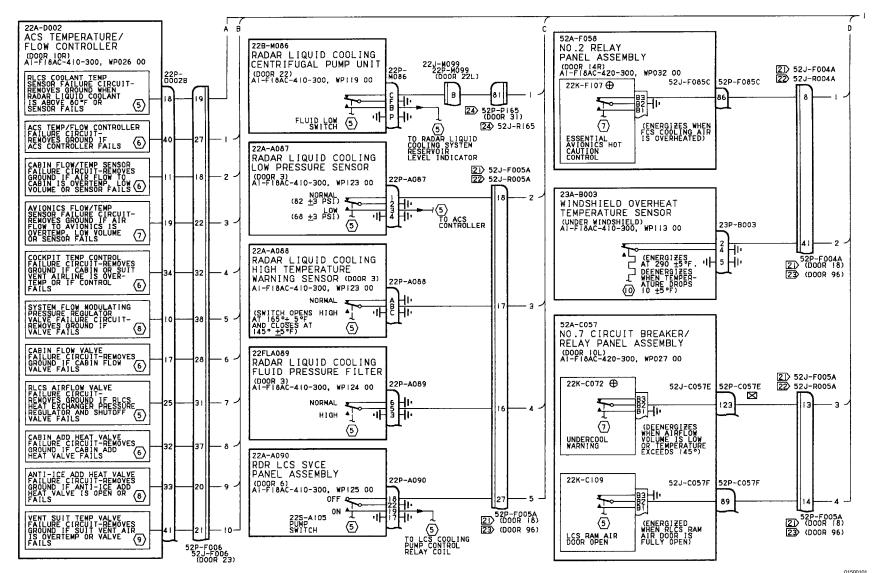


Figure 1.

Figure 1. Environmental Control Systems Interface Schematic (Sheet 1)

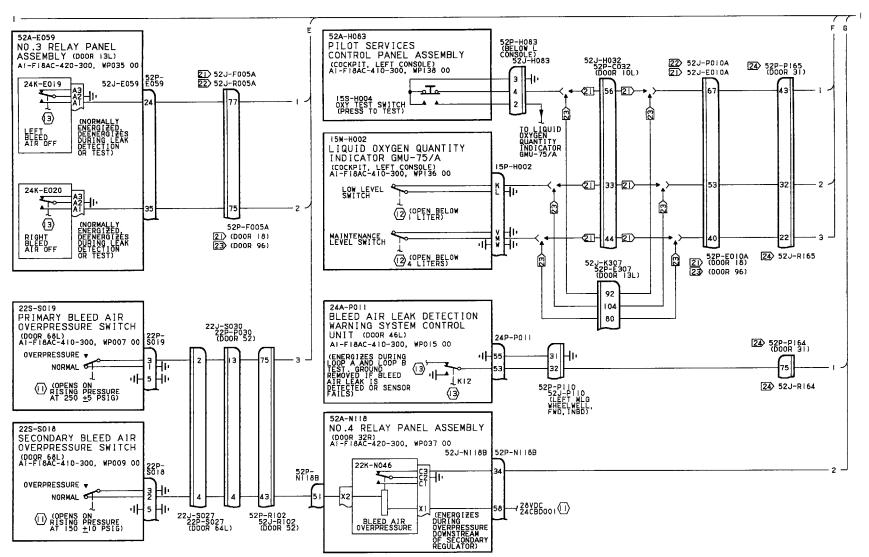
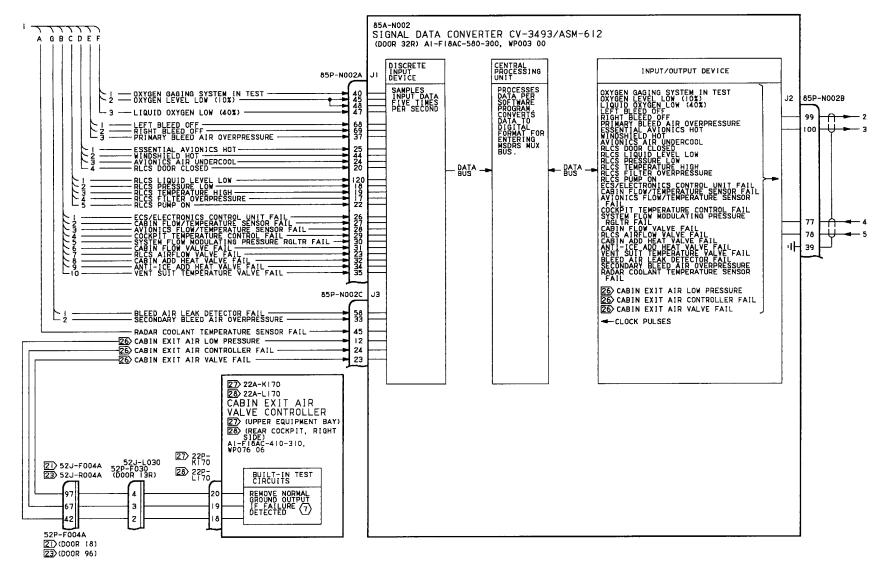
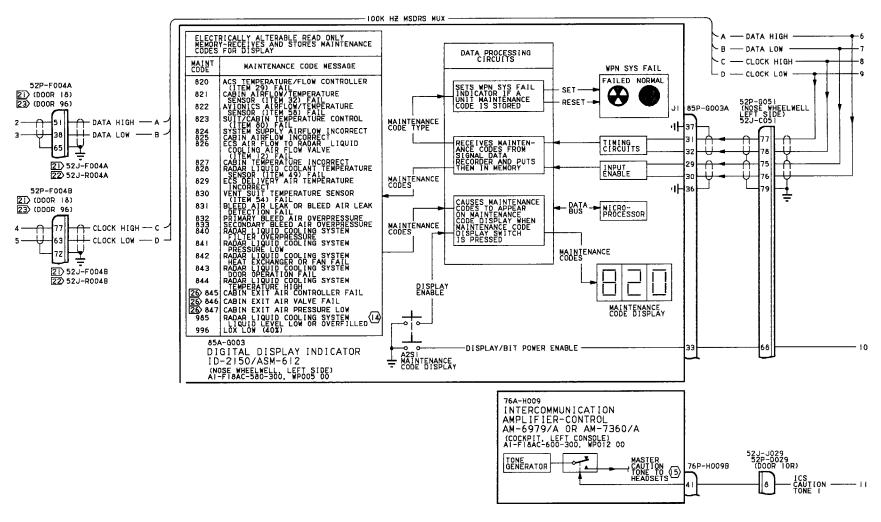


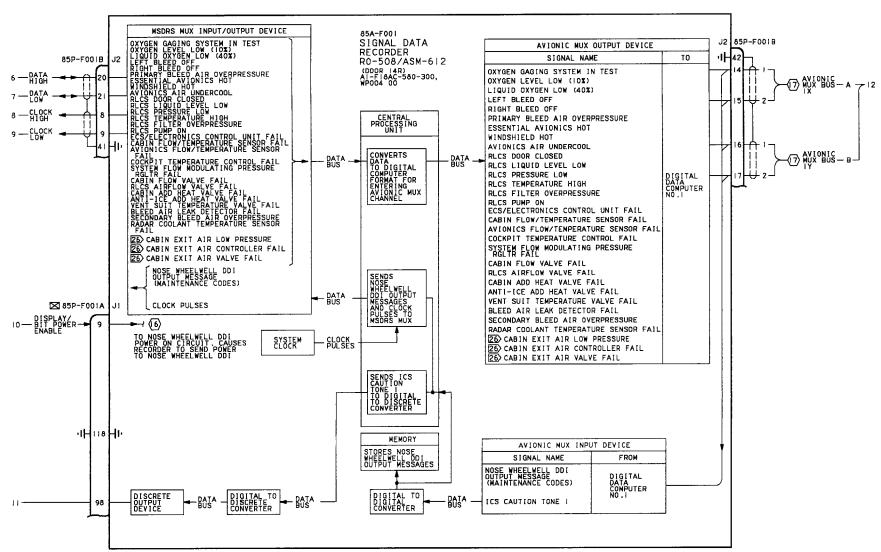
Figure 1. Environmental Control Systems Interface Schematic (Sheet 2)



01500103

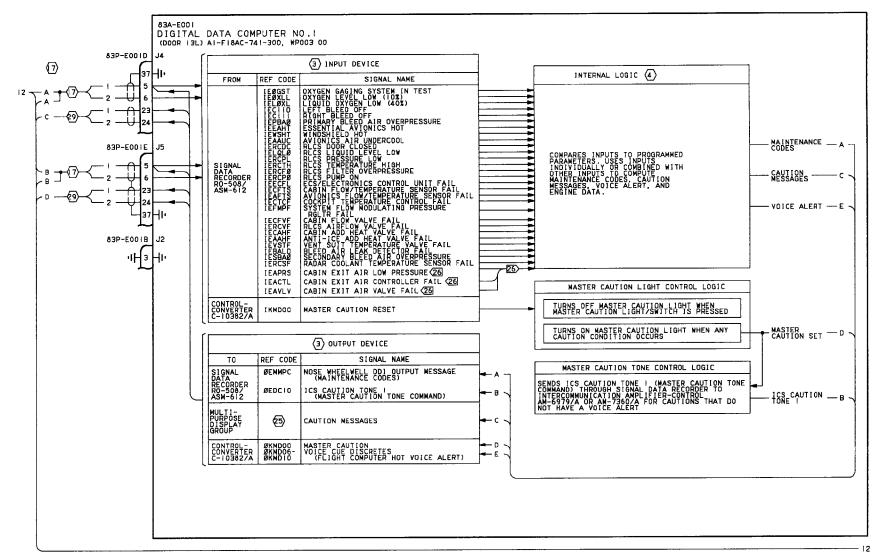
Figure 1. Figure 1. Environmental Control Systems Interface Schematic (Sheet 3)





01500105 Figure 1.

Figure 1. Environmental Control Systems Interface Schematic (Sheet 5)



01500106

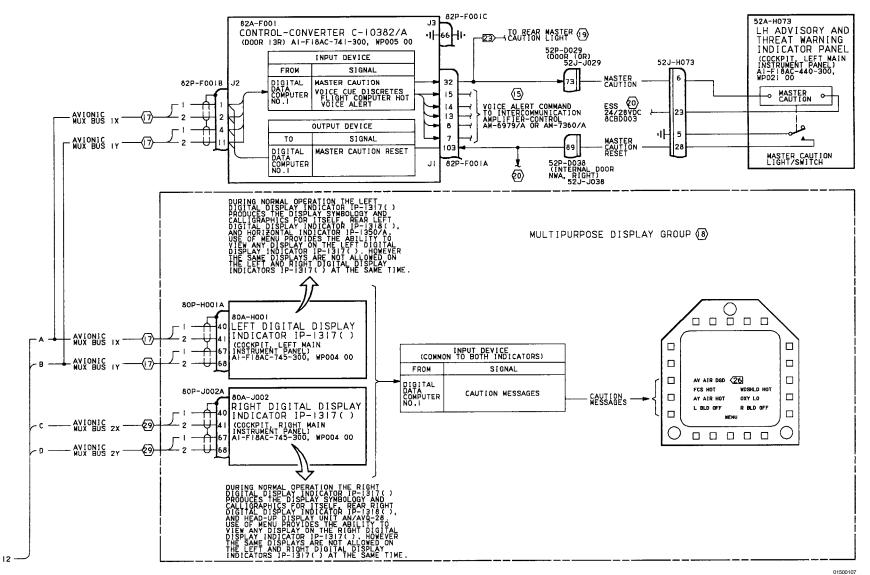


Figure 1.

Figure 1. Environmental Control Systems Interface Schematic (Sheet 7)

LEGEND

NTINUITY TESTS:

AL AIRCRAFT
ALL AIRCRAFT
ALL AIRCRAFT
ALL AIRCRAFT
AIRCRAF

- WHEN TESTING FOR CONTINUITY, TEST FOR:
 (1) SHORTS TO GROUND SURROUNDING PINS ON CONNECTORS.
 (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 (4) SHIELD CONTINUITY.
- 2. NONSTANDARD SYMBOLS:

IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT SEE NOTE | .



- DE THE CONNECTORS. SEE NOTE | .

- 6 CABIN COOLING AND DEFOG SYSTEM SCHEMATIC
- 7 AVIONICS COOLING SYSTEM SCHEMATIC-EXCEPT COCKPIT, AI-FIBAC-410-500, WP009 00.
- B AIR CYCLE AIR CONDITIONING SYSTEM SCHEMATIC. AI-FIBAC-410-500. WP007 00.
- 9 VENT SUIT SYSTEM SCHEMATIC, AI-FIBAC-410-500,
- (0) WINDSHIELD ANTI-ICE AND RAIN REMOVAL SYSTEM SCHEMATIC. AI-FIBAC-410-500. WP013 00.

- INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- (6) POWER SCHEMATIC, WP005 00.

- P REAR COCKPIT CAUTION LIGHTS SCHEMATIC,
- COCKPIT CAUTION LIGHTS SCHEMATIC,
- 21) F/A-18A
- 22 F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 23 F/A-18B
- 24) 162445 AND UP.
- 26) 163092 AND UP
- 27 F/A-18A 163092 AND UP
- 28 F/A-18B 163104 AND UP.
- (29) AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FIBAC-741-500, WP005 00.

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - ENVIRONMENTAL CONTROL SYSTEMS INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Environmental Control Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	1
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

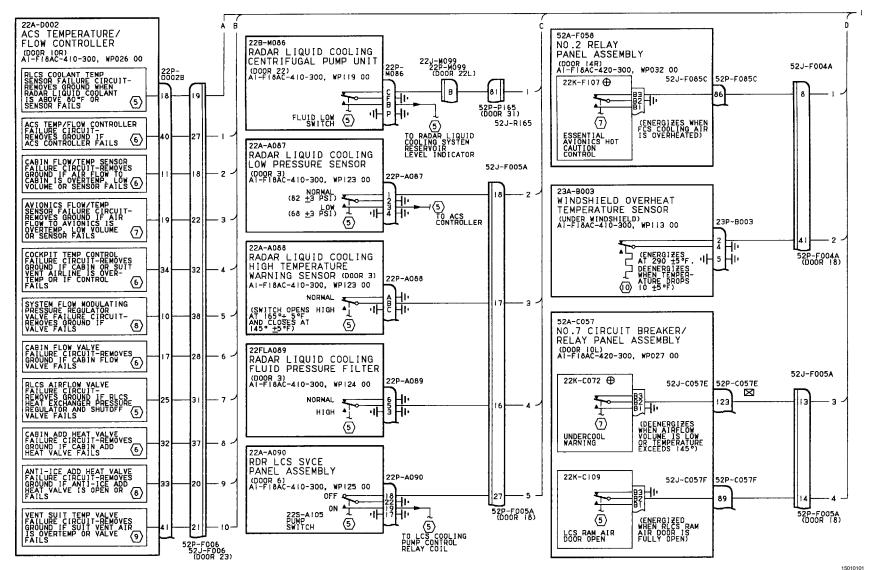


Figure 1. Figure 1.

Figure 1. Environmental Control Systems Interface Schematic (Sheet 1)

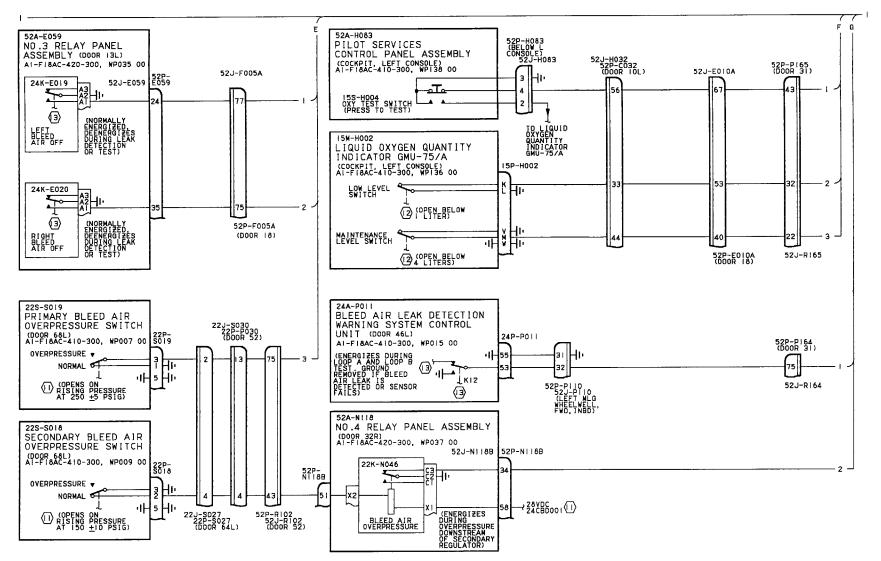
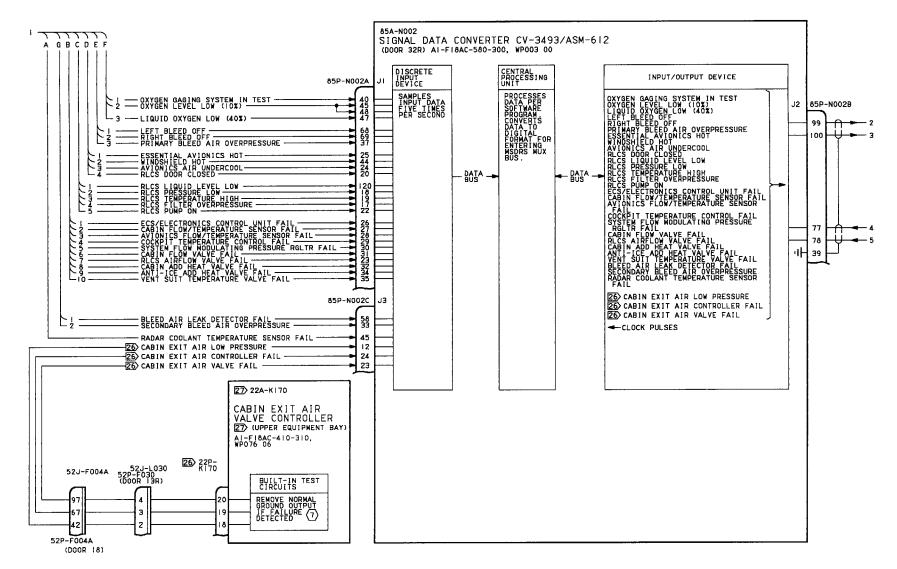
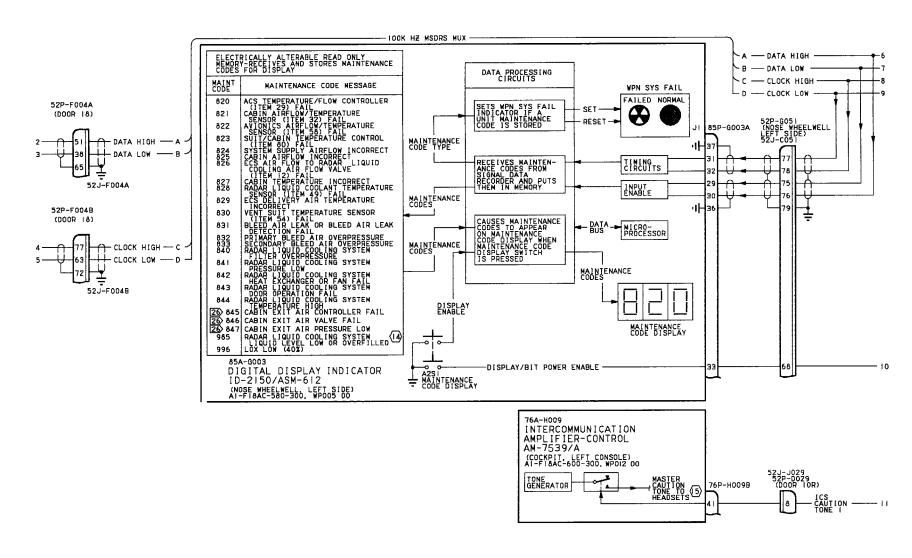


Figure 1. Environmental Control Systems Interface Schematic (Sheet 2)

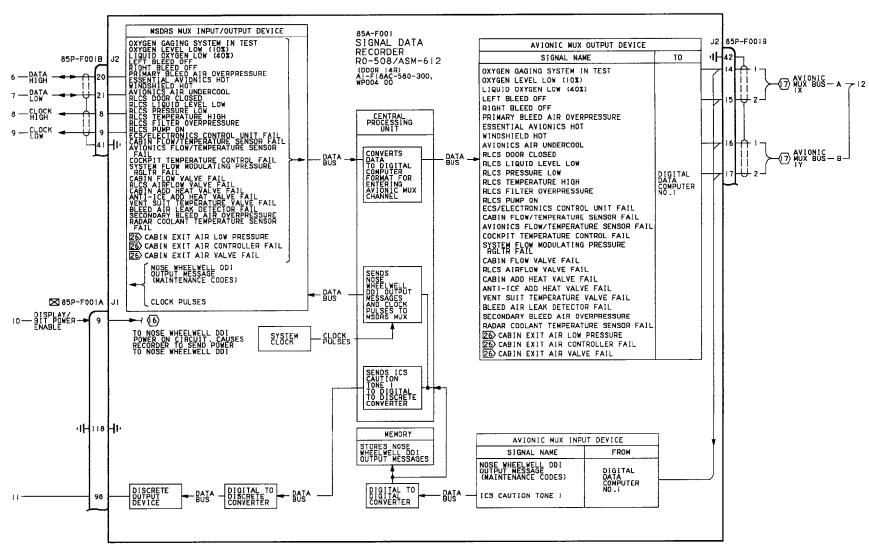


15010103 Figure 1.

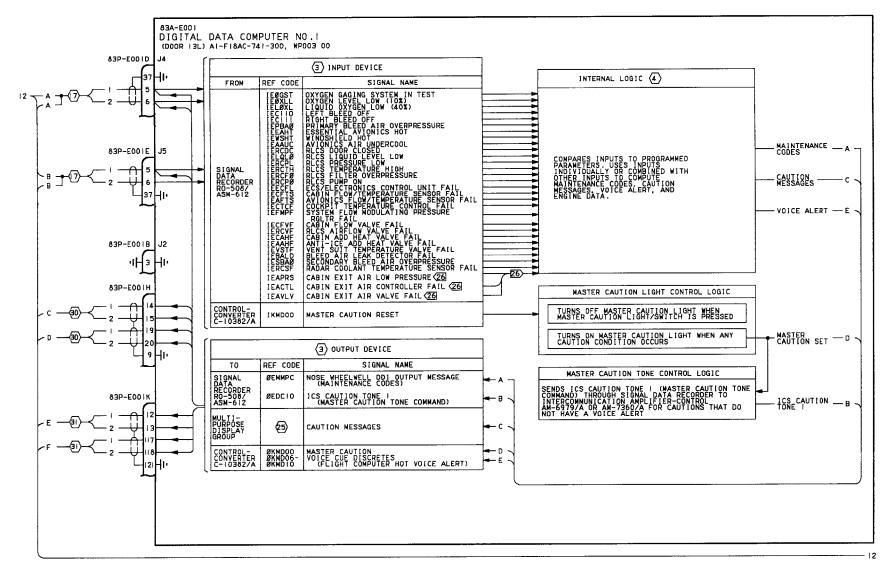
Figure 1. Environmental Control Systems Interface Schematic (Sheet 3)



15010104 Figure 1.



15010105 Figure 1.



15010106

Figure 1.

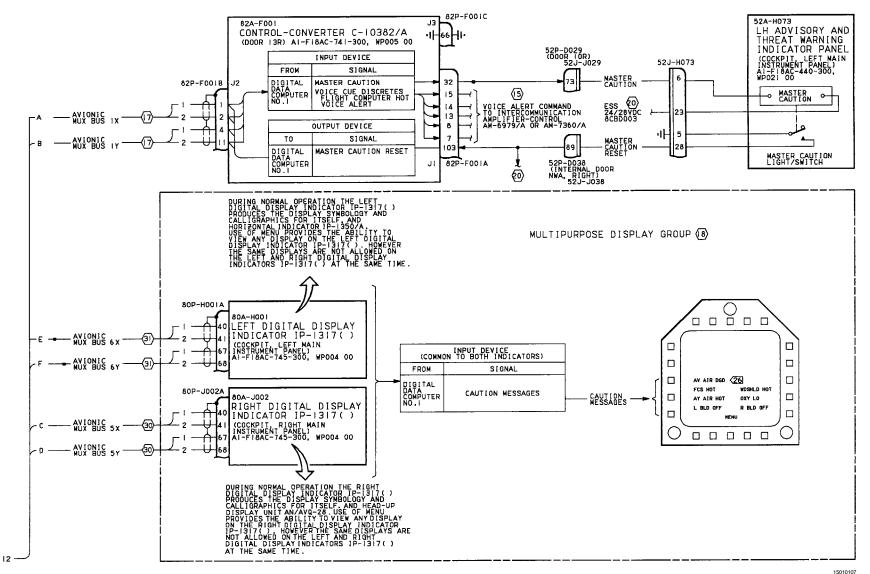


Figure 1.

Figure 1. Environmental Control Systems Interface Schematic (Sheet 7)

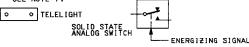
LEGEND

CONTINUITED STATE NUMBERS, SPLICE POINTS, AND
A. ALTHORIST WIRE NUMBERS, SPLICE POINTS, AND
B. WHEN, A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED AND SOCKET FOR CORRECT REINSTALLATION, DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY IF RELAY IS DEFECTIVE, REPLACE WITH ANY OTHER USED RELAY IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS)
FOR CONTINUITY WITH MULTIMETER ON RXI SCALE. PIN TO PIN
TESTS THAT DO NOT BO THROUGH SWITCHES/RELAY CONTACTS
MAY USE THE RXI SCALE

- D. WHEN TESTING FOR CONTINUITY, TEST FOR:
 (1) SHORTS TO GROUND SURROUNDING PINS ON CONNECTORS.
 (2) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF CONNECTORS (IDENTIFIED BY) MAKE SURE MULTI INTERE LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS:

IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT SEE NOTE | .



- ☑ IDENTIFIES 24YDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTORS. SEE NOTE | .
- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE. REFER TO ALFIBAC TO CONTROL OF OR MEMORY INSPECT ACCESS LOCATION RELATING TO REFER TO ALFIBAC FIM-100.
- 4 ENVIRONMENTAL CONTROL SYSTEMS MAINTENANCE CODES AND
- 5 RADAR LIQUID COOLING SYSTEM SCHEMATIC
- 6 CABIN COOLING AND DEFOG SYSTEM SCHEMATIC
- 7 AVIONICS COOLING SYSTEM SCHEMATIC-EXCEPT COCKPIT, AI-FIBAC-410-500, WP009 00.
- (8) AIR CYCLE AIR CONDITIONING SYSTEM SCHEMATIC, AI-FIBAC-410-500, WP007 00.
- 9 VENT SUIT SYSTEM SCHEMATIC, AI-F18AC-410-500.
- (0) WINDSHIELD ANTI-ICE AND RAIN REMOVAL SYSTEM SCHEMATIC. AI-FIBAC-410-500. WP013 00.

- AI-FIBAC-410-500, WP005 00
- (12) OXYGEN SYSTEM SCHEMATIC, A1-F18AC-410-500, WP016 00
- (3) BLEED AIR LEAK DETECTION SYSTEM
 SCHEMATIC, AI-FISAC-410-500, WP006 00.
- (4) MAINTENANCE CODE 985 CAN ALSO BE DETECTED DURING TEST FOR FLUID LOW MAINTENANCE CODES. SEE FLUIDS TEST SCHEMATIC, WPOOF OO.
- (5) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-600-500, WP013 00.
- (6) POWER SCHEMATIC, WP005 00.
- (17) AVIDNIC MUX CHANNEL I SCHEMATIC
- (a) THE MULTIPURPOSE DISPLAY OROUP IS HADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR INTIGATOR INTIGATOR IP-1317() HEAD-UP DISPLAY UNIT ANAMAY 28 HORIZONTAL INDICATOR IP-1350A, FOR MULTIPURPOSE DISPLAY GROUP, REFER TO A1-F18AC-745-500.
- (9) DELETED.
- COCKPIT CAUTION LIGHTS SCHEMATIC,
- 2) DELETED.
- 22 DELETED.
- 23 DELETED
- 24 DELETED.
- (5) DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY
 MALFUNCTION EXISTS TRANSFER DISPLAY TO ANOTHER
 INDICATOR. IF MALFUNCTION EXISTS ON MORE THAN
 ONE INDICATOR. TROUBLESHOOT USING ALFIBA() -OLD-QOO
 INDICATOR. TROUBLESHOOT BY DOING DISPLAY TEST.
 ALFIBAC-745-200. WPOOT BY DOING DISPLAY TEST.
- 26) 163092 AND UP.
- 27 DELETED.
- 28 DELETED.
- 29 DELETED.
- (0) AVIDNIC MUX CHANNEL 5 SCHEMATIC, AI-FIBAC-741-500, WPOIB 00.
- (1) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00.

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - LANDING GEAR AND RELATED SYSTEMS INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP005 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No.
Landing Gear and Related Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 48	-	Alternating Current Bus Isolation (ECP MDA-F/A-18-00121)	1 Sep 86	ECP coverage only

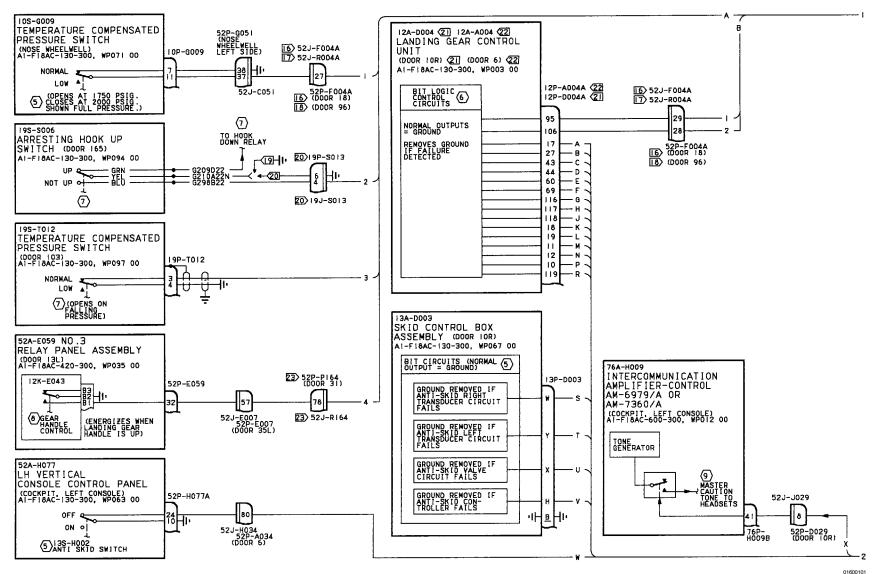


Figure 1.

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 1)

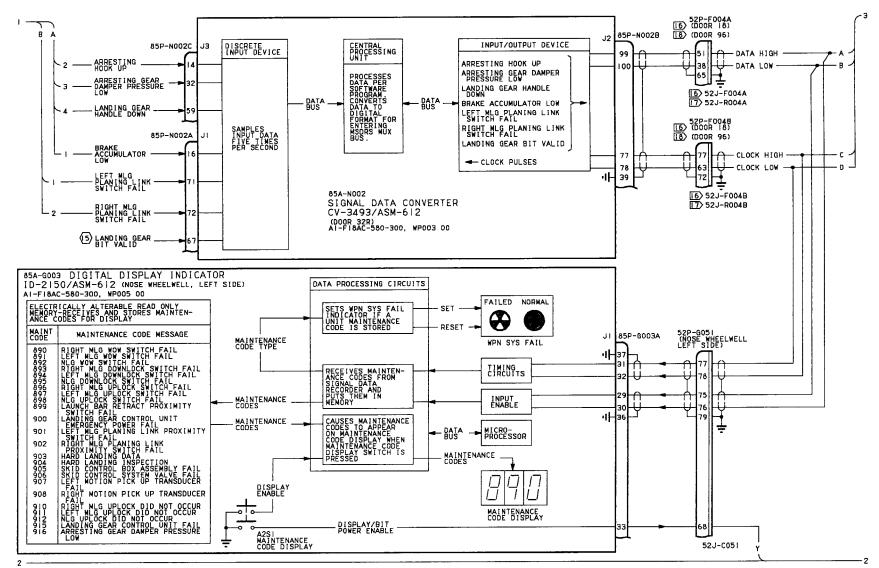


Figure 1.

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 2)

A1-F18AC-580-500

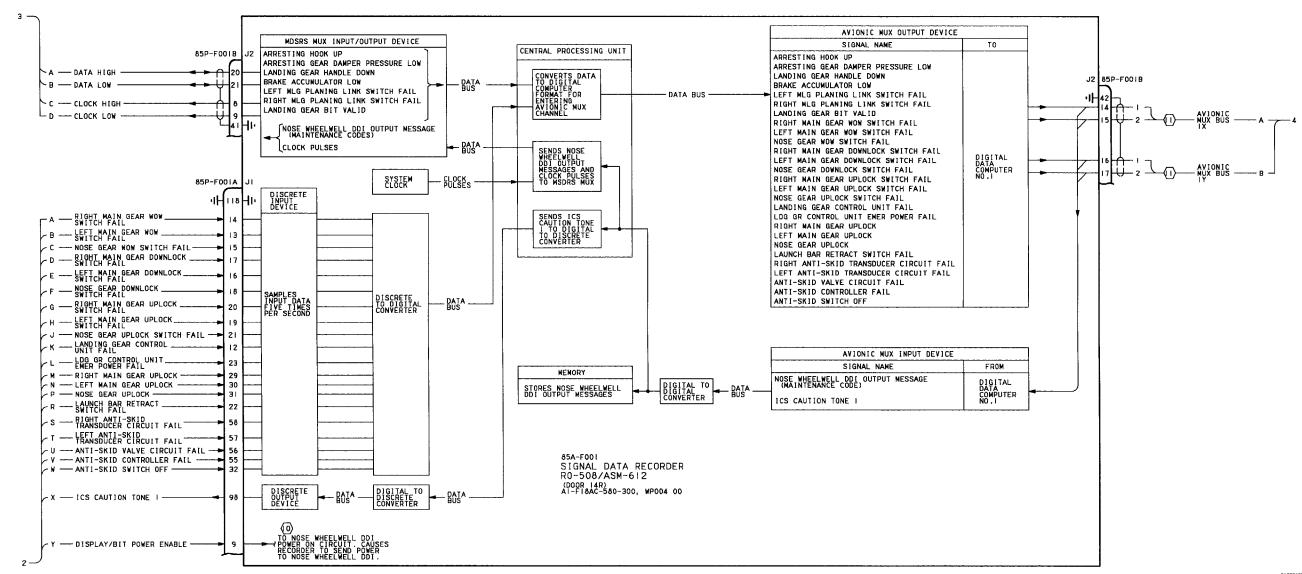
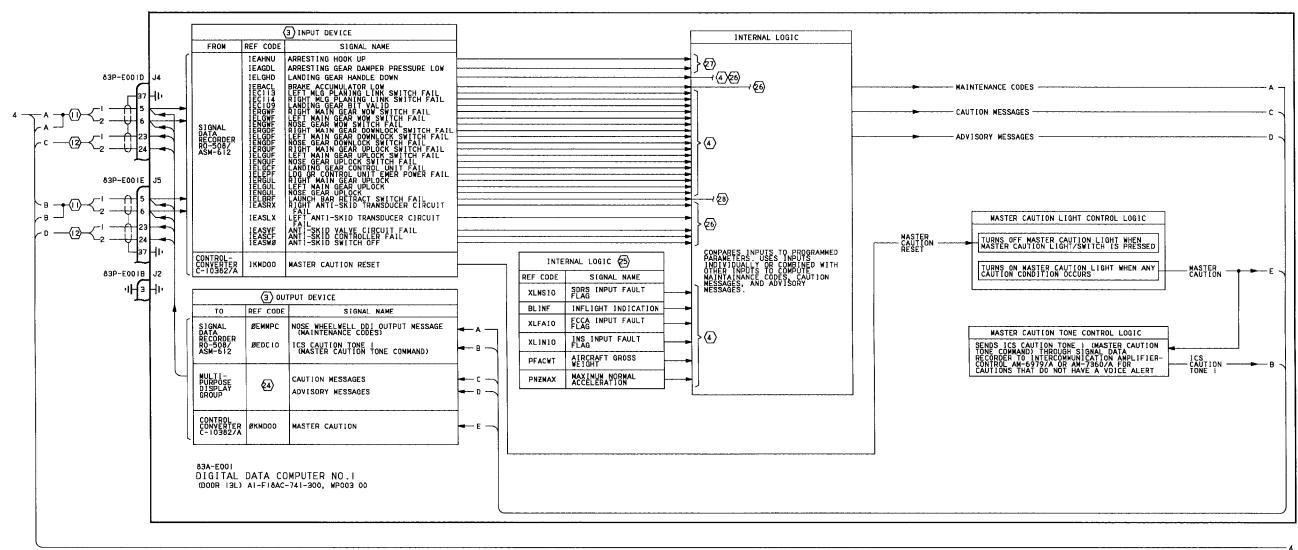


Figure 1.

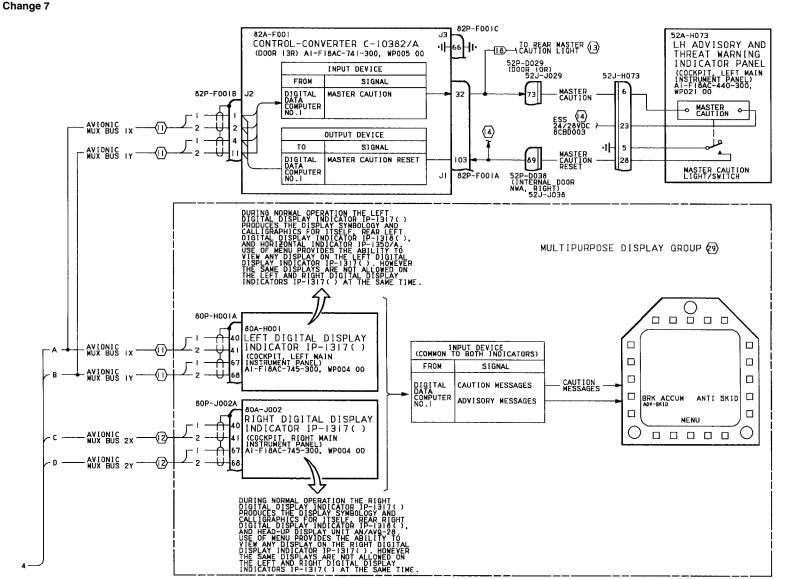
Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 3)

Figure 1.



01600104

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 4)



LEGEND

- I. CONTINUITY TEST:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RXI SCALE, PIN TO PIN TESTS THAT DO NOT GO THOUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
- D. WHEN TESTING FOR CONTINUITY, TEST
 - (1) SHORIS TO GROUND
 (2) SHORIS BETWEEN SÜRROUNDING PINS
 ON CONNECTORS
 (3) SHORIS BETWEEN SHIELD AND
 CONDUCTORS
 (4) SHIELD CONTINUITY.

- 2. NONSTANDARD SYMBOLS
- TINDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE 1.



- (3) FOR LOGIC DIAGRAMS FRELATING TO REF. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REFER TO ALL-FIBAC-F
- LANDING GEAR SYSTEM MAINTENANCE CODES SCHEMATIC, AI-FIBAC-130-500, WP007 00
- (5) WHEELBRAKE AND ANTI-SKID SYSTEM SCHEMATIC, AI-FIBAC-130-500, WP008 OI.
- 6 LANDING GEAR BIT SYSTEM SCHEMATIC,
- ARRESTING GEAR SYSTEM SCHEMATIC,
- B LANDING GEAR CONTROLLED RELAYS SCHEMATIC, AI-FIBAC-130-500, WP006 00.
- 9 INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- (0) POWER SCHEMATIC, WP005 00
- AVIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00.

- 2 AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FISAC-741-500, WP005 00.
- (4) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- (5) INPUT ALWAYS OPEN TO INDICATE LANDING GEAR CONTROL UNIT BIT VALID.
- (6) F/A-18A
- F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- (8) F/A-18B.
- (9) 161353 THRU 161521
- 20 161522 AND UP
- 161353 THRU 161987, BEFORE F/A-18 AFC
- 22) 162394 AND UP, ALSO 161353 THRU 161987 AFTER F/A-18 AFC 48.
- 23) 162445 AND UP
- (5) REE CODES USED FOR THESE COMPUTATIONS OF THE MISSING COMPUTATIONS OF THE CODES IN A 1- CODES OF THE CODES
- (6) WHEEL BRAKE AND ANTI SKID SYSTEM CAUTION AND MAINTENANCE CODES SCHEMATIC, AI-F18AC-130-500, WPOOR OI.
- (8) CATAPULT SYSTEM BIT AND MAINTENANCE CODE SCHEMATIC F/A-18A AND F/A-18B, AI-F18AC-130-50D, WPOII 00.
- AI-FIBAL-13J-350, MFCI. 30.

 (9) THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE EFT DIGITAL DISPLAY INDICATOR IN 17 TO 17 TO 18 TO

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 5)

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - LANDING GEAR AND RELATED SYSTEMS INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Landing Gear and Related Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	1
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

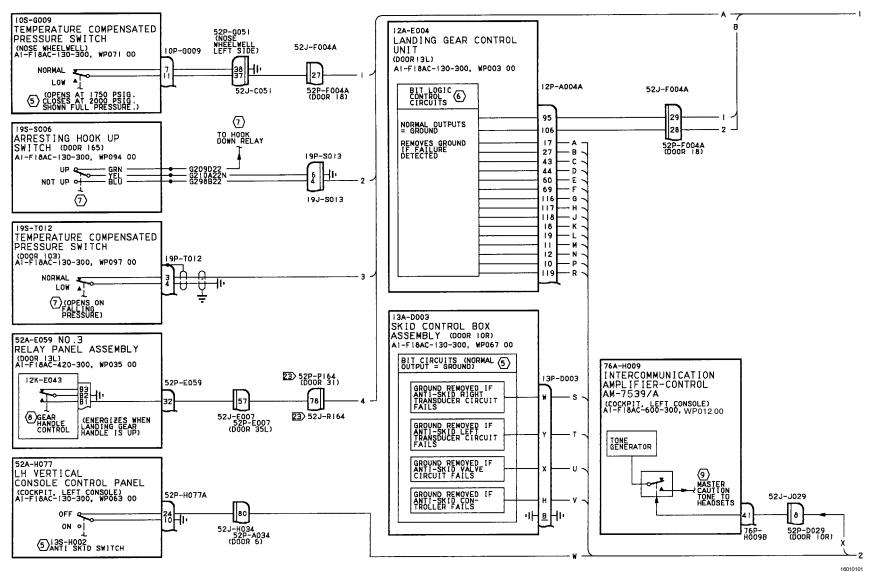


Figure 1.

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 1)

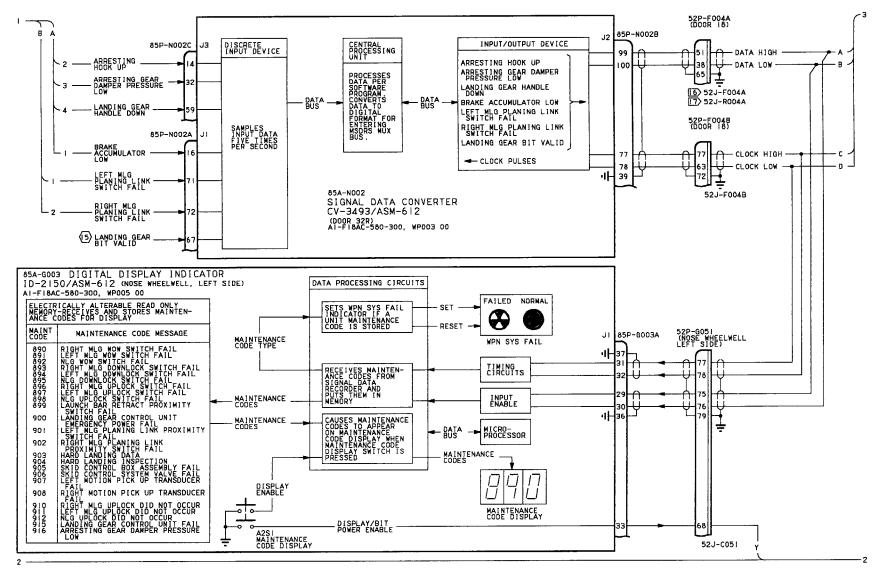


Figure 1. Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 2)

A1-F18AC-580-500

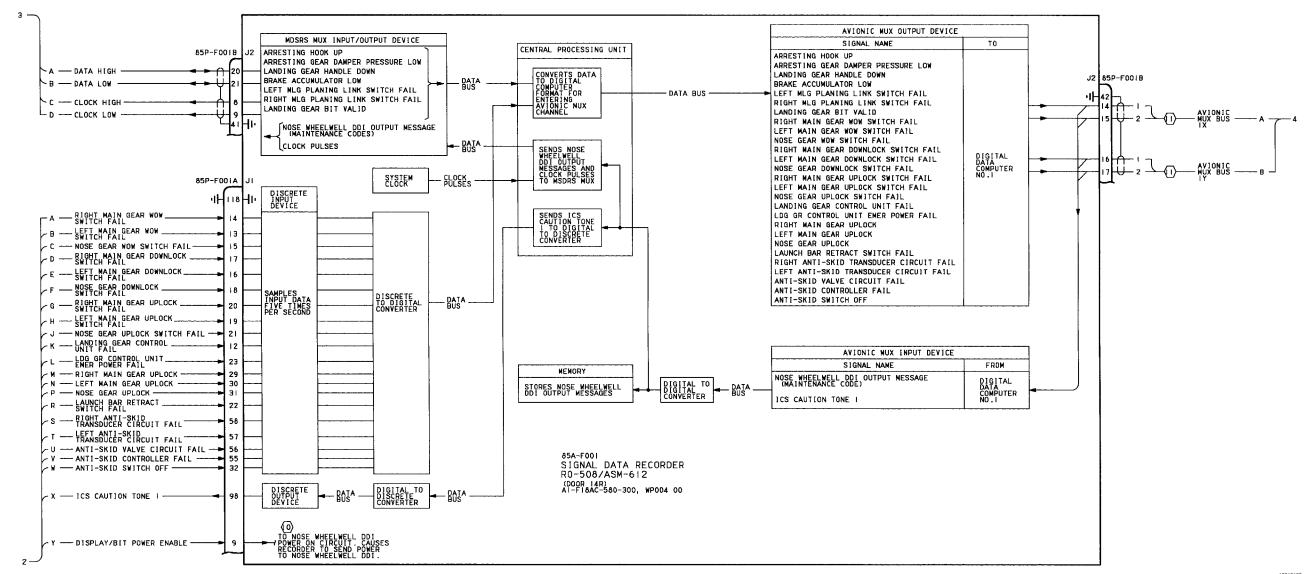


Figure 1.

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 3)

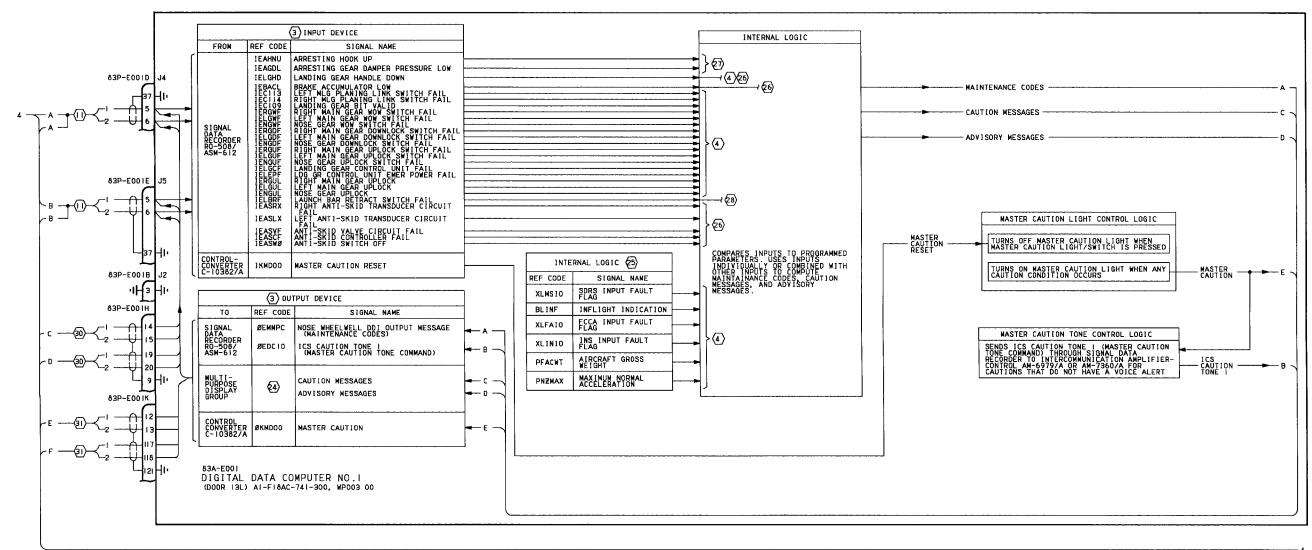
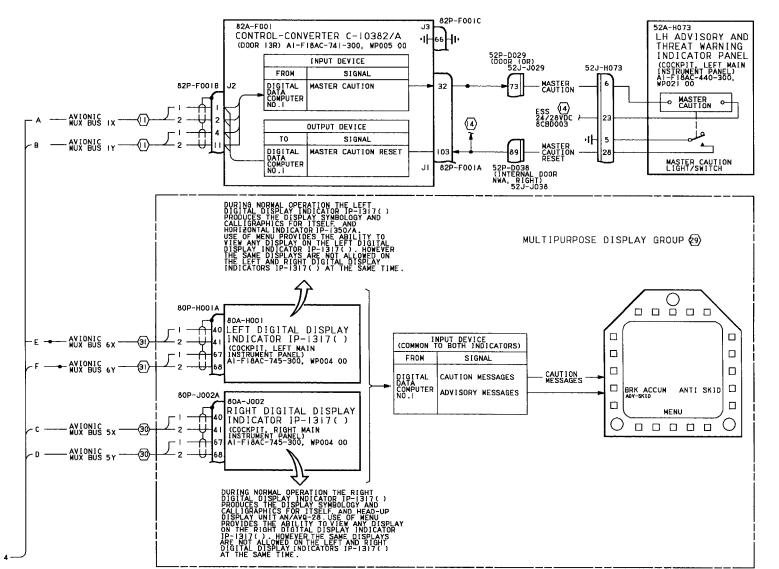


Figure 1.

Figure 1. Landing Gear and Related Systems Interface Schematic (Sheet 4)

Figure 1.



LEGEND

- I. CONTINUITY TEST
- C. DO NOT TEST LOW LEYEL DEVICES
 (SWITCHES, TRELAY CONTACTS) FOR CONTINUITY WITH WULTIMETER ON RXI
 SCALE, PIN TO PIN TESTS THAT DO NOT
 GO THROUGH SWITCHES, TRELAY CONTACTS
 MAY USE THE RXI SCALE.
- D. WHEN TESTING FOR CONTINUITY, TEST
- (1) SHORTS TO GROUND
 (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 (3) SHORTS BETWEEN SHIELD AND CONDUCTORS (4) SHIELD CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - ## INDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE | .



- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE REFER TO AI-FISAC)-OLD-OOD. FOR MEMORY INSPECT ACCESS LOCATION RELAT-ING TO REF CODE, REFER TO AI-FISAC-FIM-100.
- 5 WHEELBRAKE AND ANTI-SKID SYSTEM SCHEMATIC, AI-FIBAC-130-500, WP008 01.
- 6 LANDING GEAR BIT SYSTEM SCHEMATIC,
- ARRESTING GEAR SYSTEM SCHEMATIC,
- (B) LANDING GEAR CONTROLLED RELAYS SCHEMATIC, AI-F18AC-130-500, WP006 00.
- 9 INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-600-500, MP013 00.
- (0) POWER SCHEMATIC, WP005 00.
- AYIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00.

- (2) DELETED.
- (3) DELETED.
- (4) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- (5) INPUT ALWAYS OPEN TO INDICATE LANDING
- (6) DELETED.
- 17 DELETED.
- B DELETED.
- (9) DELETED.
- 20) DELETED.
- DELETED.
- 22 DELETED.
- 23> 162445 AND UP
- DISPLAY REF CODES ARE NOT SHOWN, IF DISPLAY MALFUNCTION EXISTS, TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ONLY ON ONE INDI-CATOR, TROUBLESHOOT BY DOING DISPLAY TEST, AI-FIBAC-745-200, WP004 00.
- (5) REF CODES USED FOR THESE COMPUTATIONS ARE MISSION COMPUTER INTERNAL MNEMONICS
- (6) WHEEL BRAKE AND ANTI SKID SYSTEM CAUTION AND MAINTENANCE CODES SCHEMATIC, AI-F18AC-130-500, WP008 01.
- ARRESTING GEAR SYSTEM MAINTENANCE CODE SCHEMATIC, AL-FISAC-130-500, WP010 00.
- (8) CATAPULT SYSTEM BIT AND MAINTENANCE CODE SCHEMATIC AI-FIBAC-130-50 WPOII OO.
- THE MULTIPURPOSE DISPLAY GROUP IS MADI UP FINE LEFT DIGITAL DISPLAY NOTCA' IND 14 TOR PROBLEM OF THE AD-UP DISPLAY IND 14 TOR PROBLEM OF THE AD-UP DISPLAY IP-138 A GOR WOLT PURPOSE DISPLAY OF GROUP, REFER TO A1-F18AC-745-500.
- (30) AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-FI8AC-741-500, WP018 00
- (a) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-F18AC-741-500, WP019 00.

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC ELECTRICAL AND HYDRAULIC SYSTEMS INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM EFFECTIVITY F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292, AND F/A-18B

This WP supersedes WP017 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No,
Electrical And Hydraulia Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 49	-	Addition of Sealed Lead Acid Battery (ECP MDA-F/A-18-00074)	1 Sep 86	ECP coverage only
F/A-18 AFC 90	-	Incorporation of GFE Battery Relay Control Unit (ECP MDA-F/A-18-00165R1)	1 Oct 88	ECP coverage only

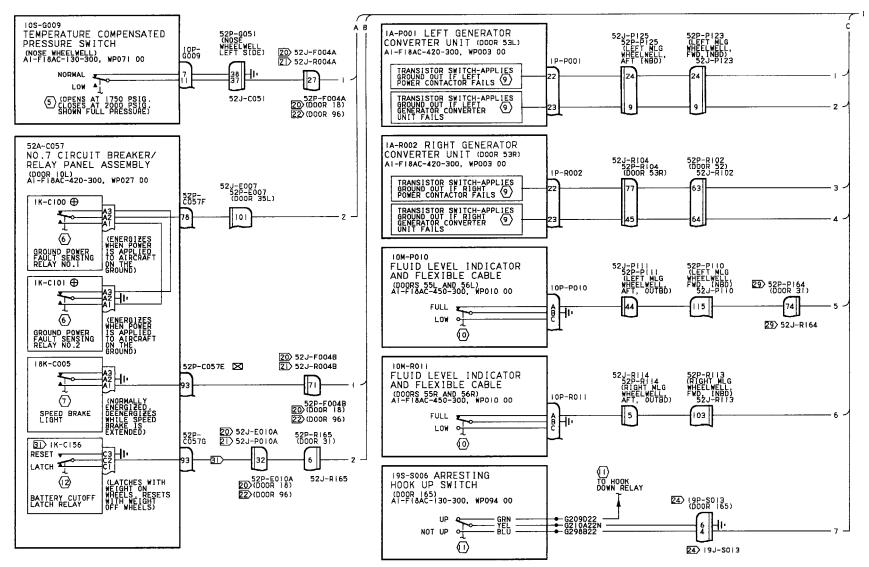


Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 1)

A1-F18AC-580-500

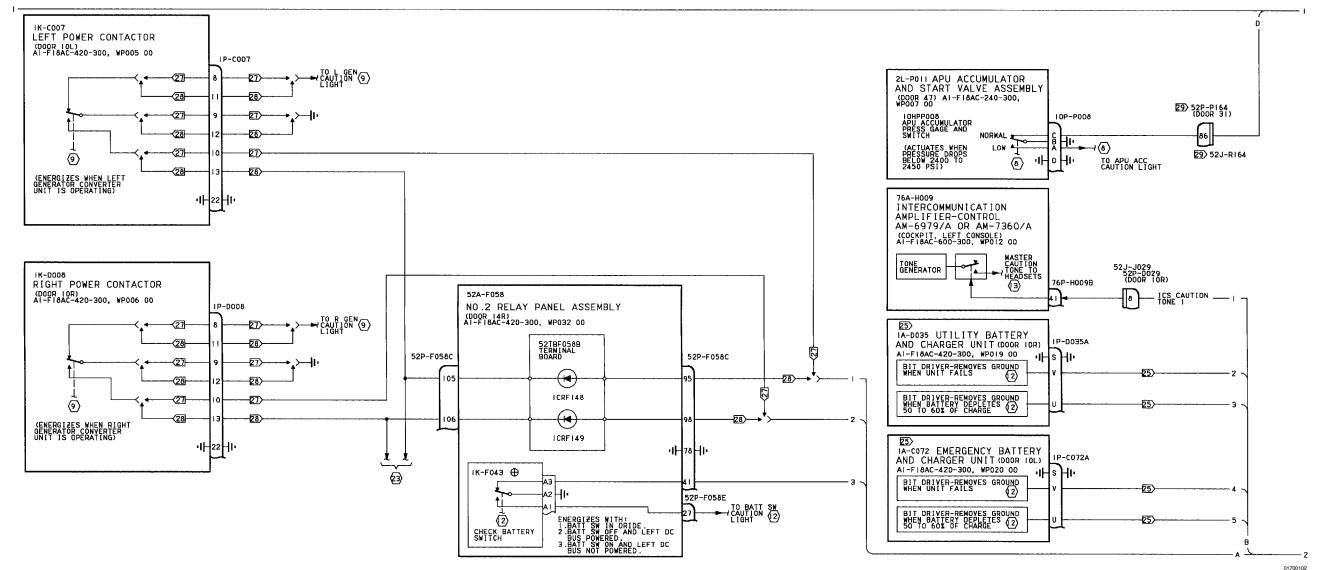
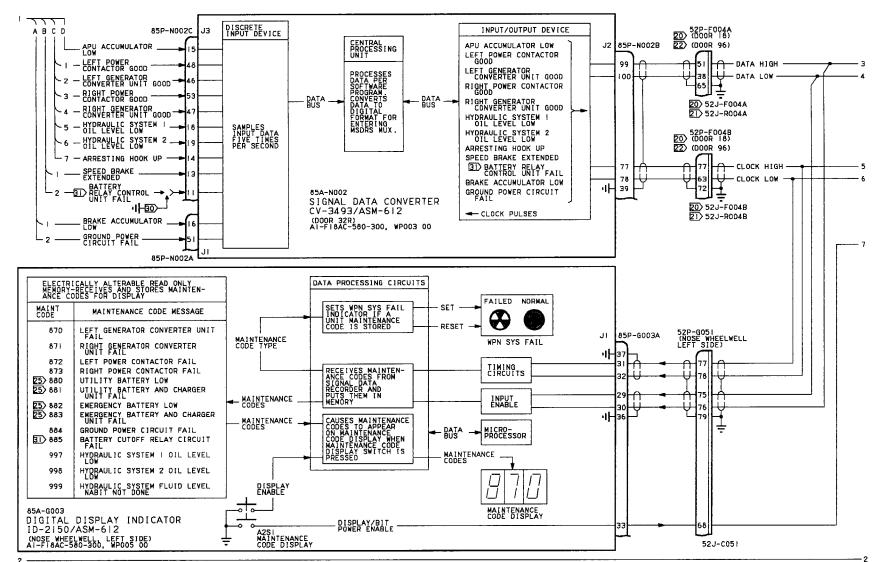


Figure 1.

Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 2)



A1-F18AC-580-500

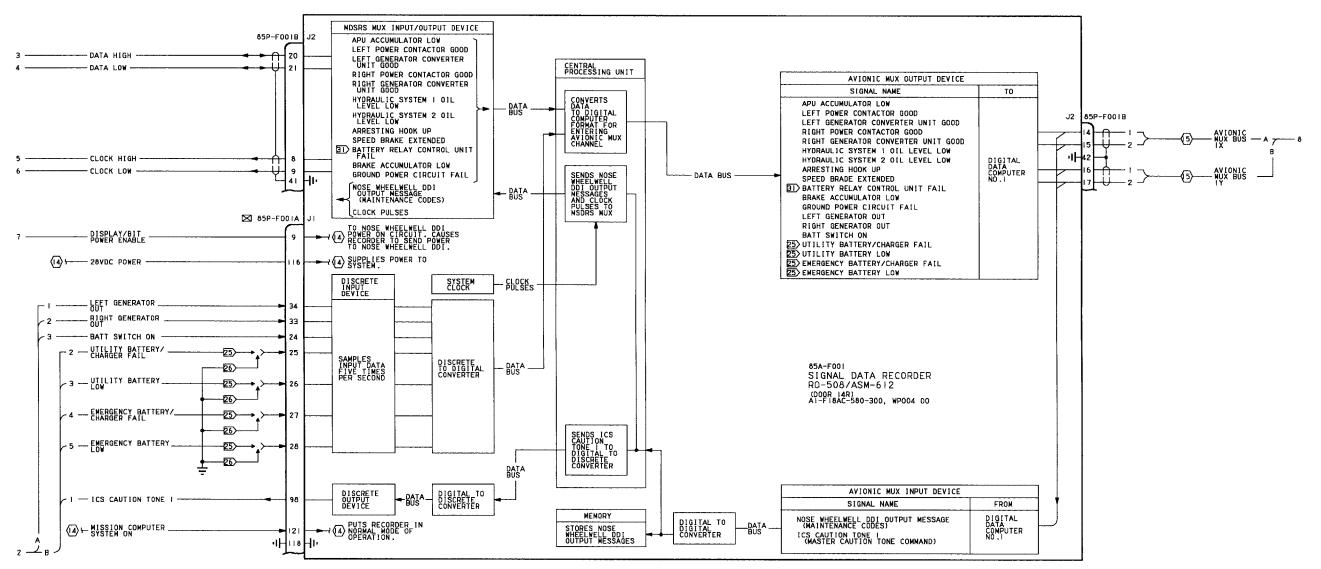


Figure 1.

Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 4)

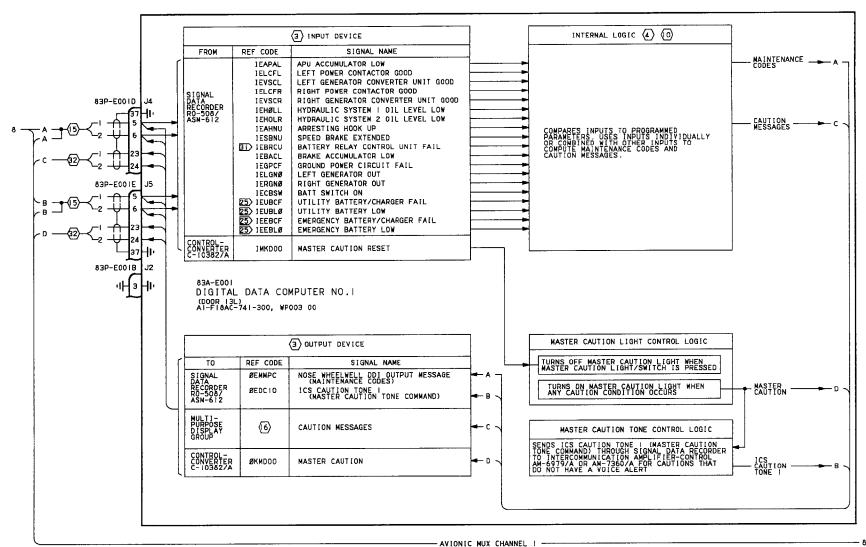


Figure 1.

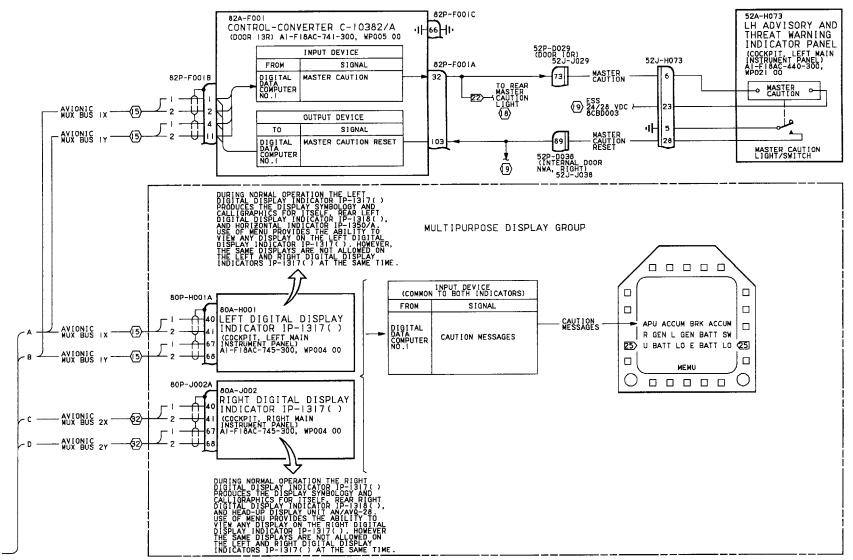
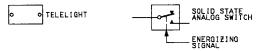


Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 6)

01700106

LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AL-FISA() -WDM-000.
 - B. WHEN A LOW LEYEL CURRENT SMITCHING RELAY (IDENTIFIED BY MO) IS REMOVED FOR TROUBLESHOOTING, IDENTIFI RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEYEL CURRECT SMITCHING RELAY MITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW DEFAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RXI SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING FOR CONTINUITY. TEST FOR:
 - (1) SHORTS TO GROUND (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS. (3) SHORTS BETWEEN SHIELD AND CONDUCTORS. (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24YDC BATTERY YOLTAGE EXISTS ON SOME PINS ON CONNECTORS (JOENTIFIED BY 20). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - # IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRECT.
 - \boxtimes IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.



- FOR LOGIC DIAGRAMS RELATING TO REF CODE . REFER TO AI-FIBAC .)-OLD-OOO. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE. REFER TO AI-FIBAC-FIBM-100.
- 4 ELECTRICAL SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATICS, AI-F18AC-420-500.
- 5 WHEELBRAKE AND ANTI-SKID SYSTEM SCHEMATIC, AI-FIBAC-130-500, WP008 00.
- 6 GROUND POWER SWITCHING SCHEMATIC, AI-FIBAC-420-500, WP005 00.
- 7 SPEED BRAKE SYSTEM SCHEMATIC, AI-FI8AC-570-500, WP026 00.
- (8) APU START SYSTEM SCHEMATIC, AI-FIBAC-240-500, WP003 00.
- 9 AC POWER SYSTEM SCHEMATIC, AI-FIBAC-420-500, WP004 00.
- (0) HYDRAULIC SYSTEM SCHEMATIC, A!-F!8AC-450-500, WP003 00.
- ARRESTING GEAR SYSTEM SCHEMATIC, AI-FIBAC-130-500,
- (2) DC POWER SYSTEM SCHEMATIC, AI-FIBAC-420-500, WP004 00
- (3) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-FIBAC-741-500, WP013 00.

- (3) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- (4) POWER SCHEMATIC, WP005 00.
- (5) AVIONIC MUX CHANNEL I SCHEMATIC, AI-F18AC-741-500, WP004 00.
- (6) DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS IRRANSEE DISPLAY TO THOUGHT INDICATOR. TROUBLE-SHOOT USING AL-E 8A ()-000 INDUT REF CODES MALFUNCTION EXISTS ON HOUSE OF THE CODES MALFUNCTION EXISTS ONLY ON ONE TO THE CODES OF THE CODE OF T
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317()) RIGHT DIGITAL DISPLAY UNDICATOR IP-1317()) RIGHT DIGITAL DISPLAY UNDICATOR IP-1350/A, AND ON F/A-18B THE REAR LEFT DIGITAL DISPLAY UNDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1318(), FOR MULTIPURPOSE DISPLAY GROUP, REFER TO A1-F184C-745-500.
- (8) REAR COCKPIT CAUTION LIGHTS SCHEMATIC, A1-F18AC-440-500,
- (9) COCKPIT CAUTION LIGHTS SCHMATIC, AI-FIBAC-440-500,
- 20) F/A-18A.
- ED F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 22 F/A-18B.
- (3) ENGINE START AND GROUND MAINTENANCE MODE SCHEMATIC,
- 24) 161522 AND UP.
- 25> 161353 THRU 161528 BEFORE F18 AFC 49.
- 26) 161702 AND UP, ALSO 161353 THRU 161528 AFTER F18 AFC 49.
- 27> 161353 THRU 161987.
- 28) 162394 AND UP.
- 29) 162445 AND UP.
- 30> 161353 THRU 163118, BEFORE F/A-18 AFC 90.
- [3]) 163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.
- AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-F18AC-741-500, WP005 00.

Change 7 - 1 December 2000

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC ELECTRICAL AND HYDRAULIC SYSTEMS INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM EFFECTIVITY F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Electrical And Hydraulic Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

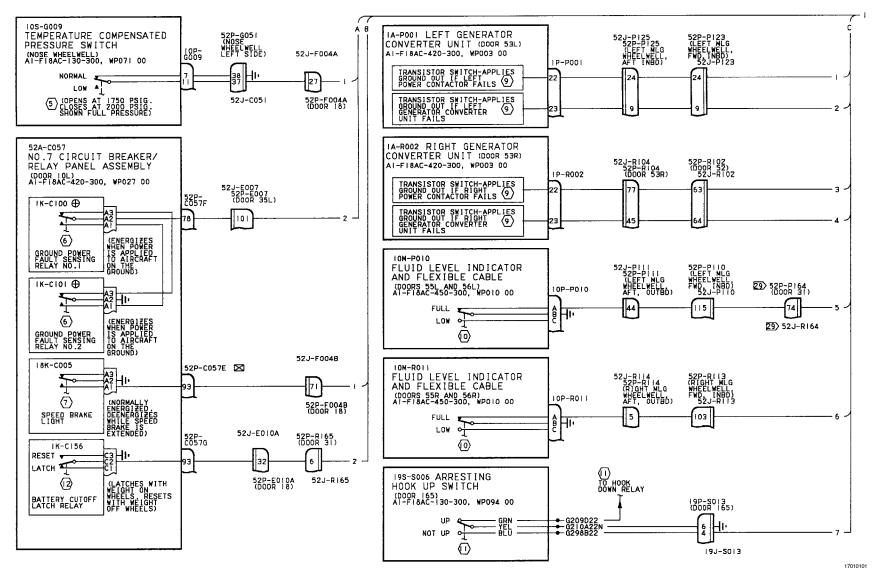


Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 1)

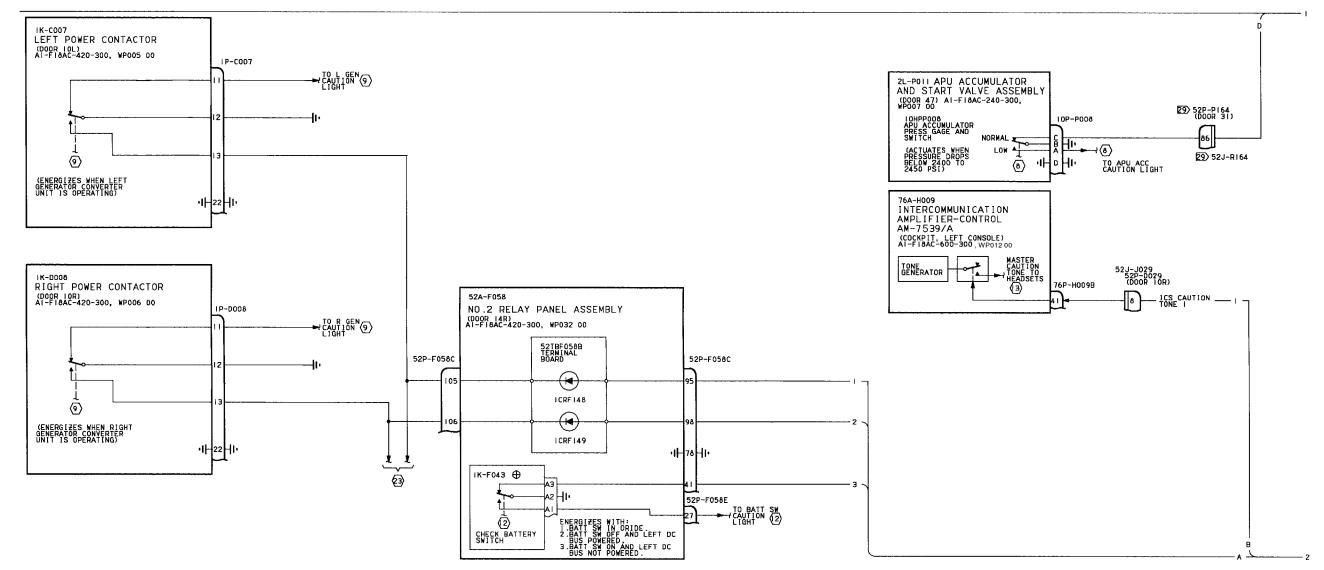
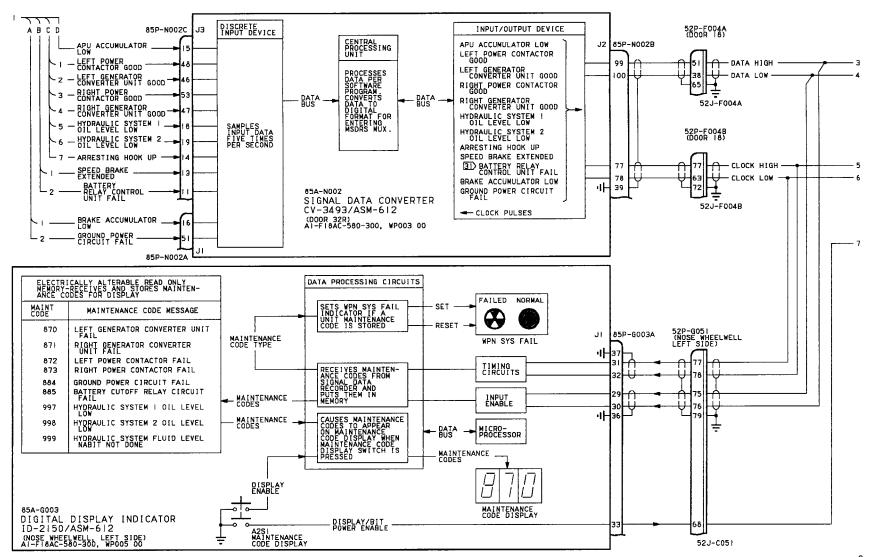


Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 2)

Figure 1.



A1-F18AC-580-500 Change 7

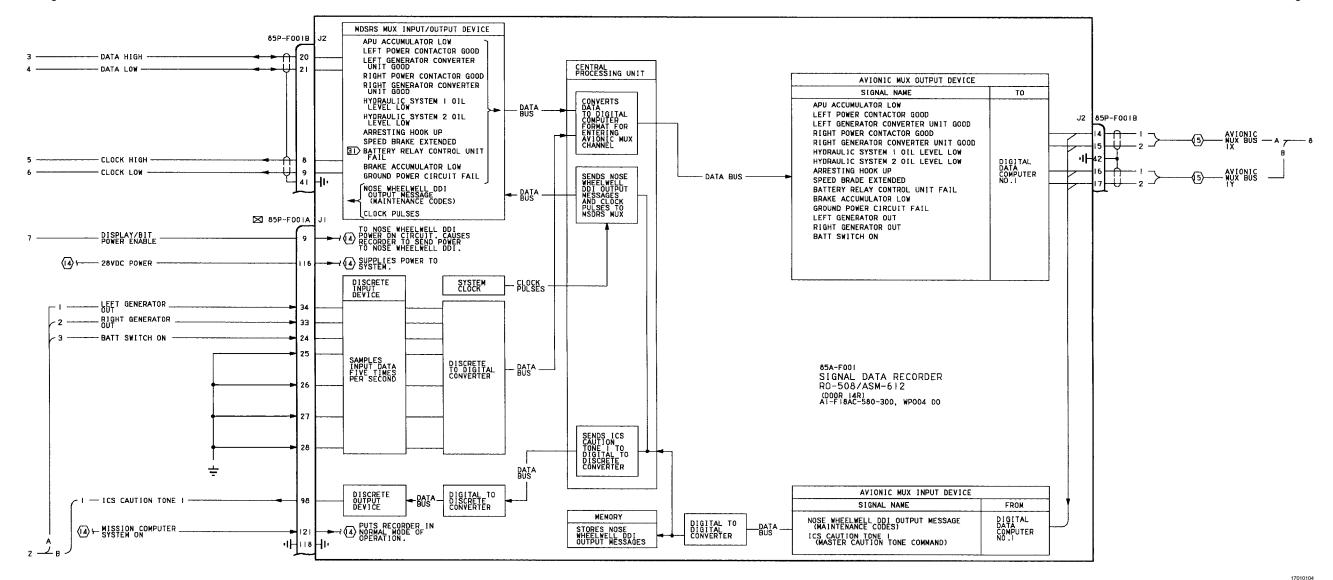
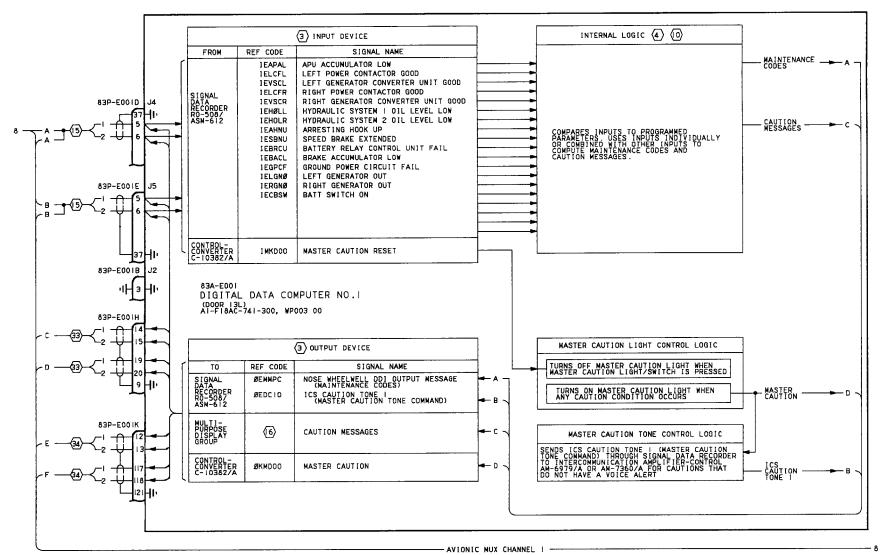


Figure 1.

Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 4)



17010105

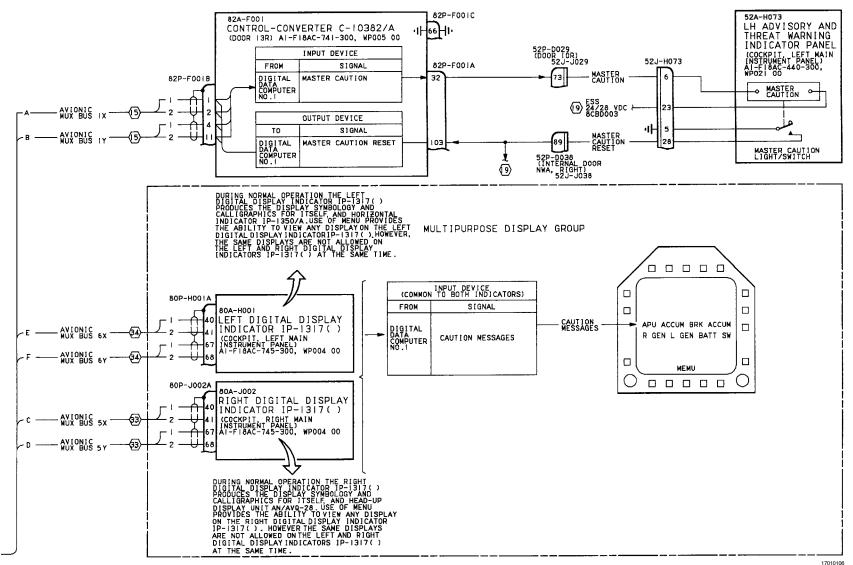
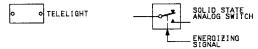


Figure 1.

Figure 1. Electrical and Hydraulic Systems Interface Schematics (Sheet 6)

LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FISA() -WDM-000.
 - B. WHEN A LOW LEYEL CURRENT SWITCHING RELAY (IDENTIFIED BY MO) CST REMOVED FOR TROUBLESHOOTING. ALL ATION: DO NOT REPLAY LOW LEYEL CURBECT SWITCHING ALL ATION: DO NOT REPLACE USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS)
 FOR CONTINUITY WITH HULTIMETER ON RXI SCALE. PIN TO PIN
 TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS
 MAY USE THE RXI SCALE.
 - D. WHEN TESTING FOR CONTINUITY. TEST FOR:
 - (1) SHORTS TO GROUND (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS. (3) SHORTS BETWEEN SHIELD AND CONDUCTORS. (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24YDC BATTERY YOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☑). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - # IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRECT.
 - \boxtimes IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.



- FOR LOGIC DIAGRAMS RELATING TO REF CODE. REFER TO AI-FIBAS ()-OLD-ODO. FOR NEWBOY INSPECT ACCESS LOCATION RELATING TO REF CODE. REFER TO AI-FIBAC-FIM-100.
- 4 ELECTRICAL SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATICS, AI-F18AC-420-500.
- (5) WHEEL BRAKE AND ANTI-SKID SYSTEM SCHEMATIC, AI-FIBAC-130-500, WP008 00.
- 6 GROUND POWER SWITCHING SCHEMATIC, AI-FIBAC-420-500, WP005 00
- 7 SPEED BRAKE SYSTEM SCHEMATIC, AI-FI8AC-570-500, WP026 00.
- (8) APU START SYSTEM SCHEMATIC, AI-FI8AC-240-500, WP003 00.
- (9) AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.
- (0) HYDRAULIC SYSTEM SCHEMATIC, A!-F!8AC-450-500, WP003 00.
- (II) ARRESTING GEAR SYSTEM SCHEMATIC, AI-FIBAC-130-500.
- (2) DC POWER SYSTEM SCHEMATIC, AI-FIBAC-420-500, WP004 00
- (3) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, A1-F18AC-741-500, WP013 00.

- (3) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC,
- (4) POWER SCHEMATIC, WP005 00.
- (5) AVIONIC MUX CHANNEL | SCHEMATIC, A1-F18AC-741-500, WP004 00.
- (6) DISPLAY REF. CODES ARE NOT TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ONLY ON ONE INDICATOR, TROUBLESHOOT BY DOING DISPLAY TEST, A1-F18AC-745-200, WP004 00.
- THE MULTIPURPOSE DISPLAY OROUP IS MADE UP OF THE LEFT DISITAL INDICATOR IP-131 (1) HEAD UP DISPLAY UNIT ANY AVG-28 HORIZONTAL INDICATOR IP-1350/A, FOR MULTIPURPOSE DISPLAY GROUP, REFER TO A 1-F18AC-745-500.
- (8) DELETED.
- (9) COCKPIT CAUTION LIGHTS SCHMATIC, AI-F18AC-440-500, WP006 00.
- 20) DELETED.
- 2I) DELETED.
- 22) DELETED.
- (3) ENGINE START AND GROUND MAINTENANCE MODE SCHEMATIC, AI-FIBAC-240-500, WP005 00.
- 24 DELETED.
- 25) DELETED.
- 26) DELETED.
- 27 DELETED.
- 28) DELETED.
- 29) 162445 AND UP.
- 30) DELETED.
- 31) DELETED
- 2 DELETED.
- (93) AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-F18AC-741-500, WP018 00.
- 4 AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00.

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - SECONDARY POWER SYSTEM INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292, AND F/A-18B

This WP supersedes WP018 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No.
Secondary Power System Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 26	-	Air Turbine Starter/Airframe Mounted Accessory Drive Design Changes (ECP MDA F18-000-68)	1 Mar 85	ECP coverage only
F/A-18 AFC 27	-	Improvement of Leading Edge Flap Design (ECP MDA-F/A-18-00044)	1 May 86	ECP coverage only

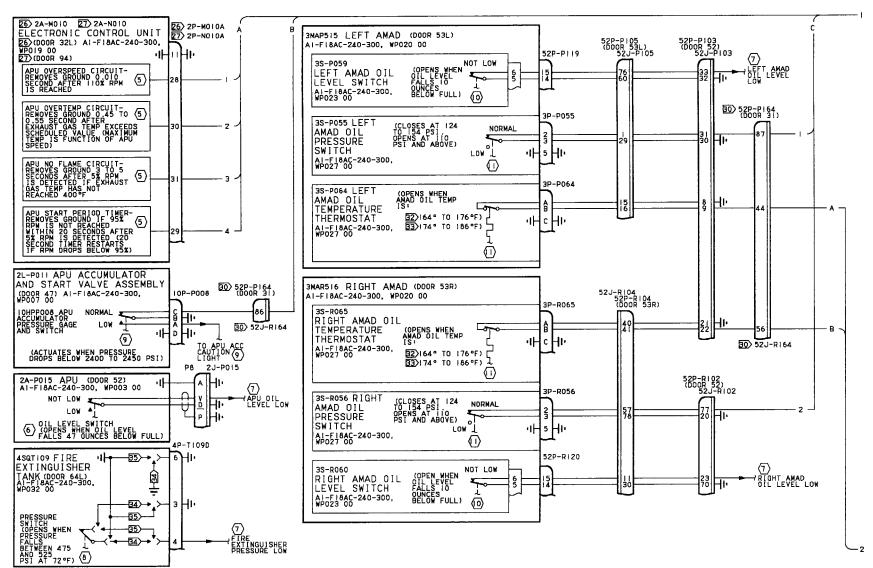
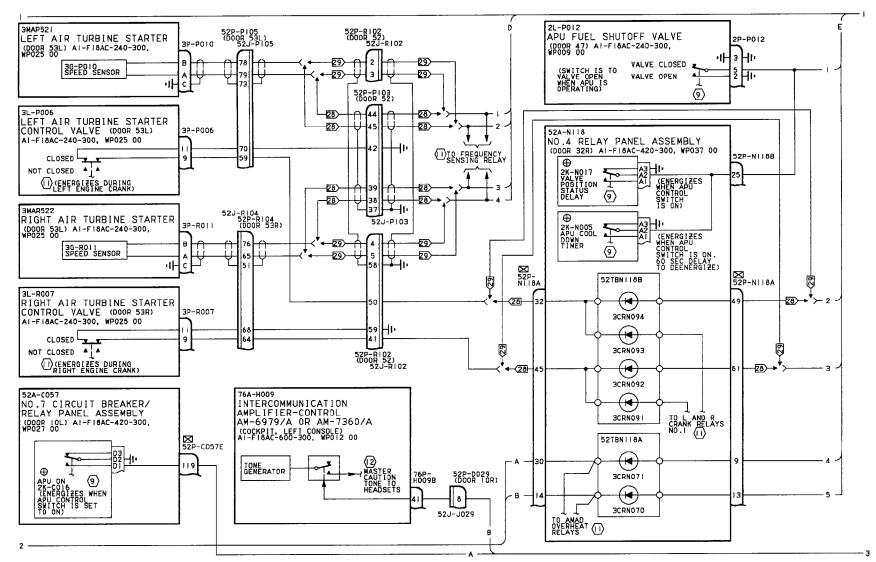


Figure 1.

Figure 1. Secondary Power System Interface Schematic (Sheet 1)

o1800101 Figure 1.



01800102

Figure 1.

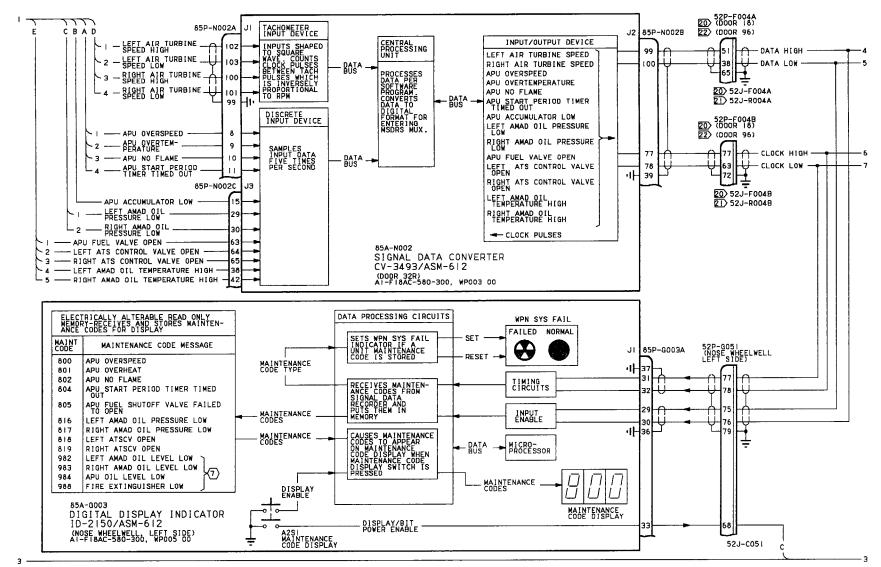


Figure 1. Secondary Power System Interface Schematic (Sheet 3)

A1-F18AC-580-500

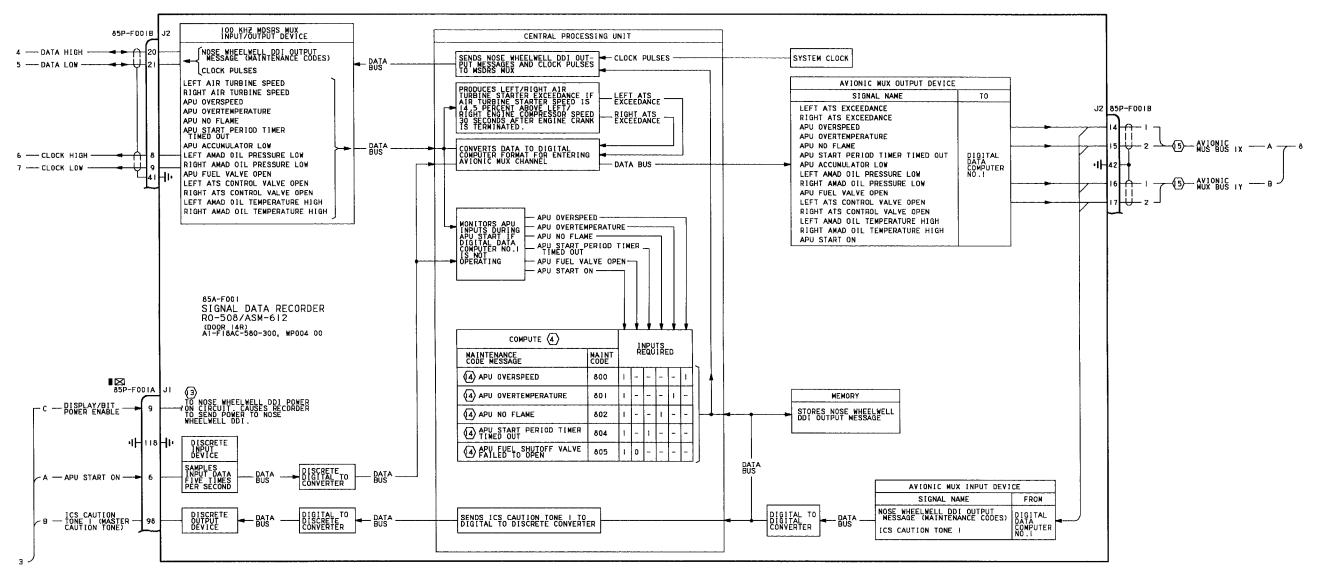
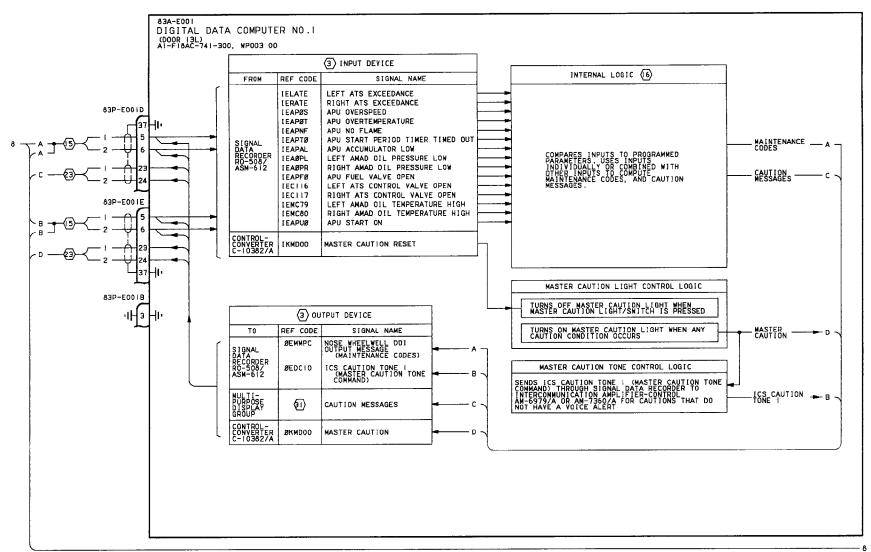


Figure 1.

Figure 1. Secondary Power System Interface Schematic (Sheet 4)



o1800105 Figure 1.

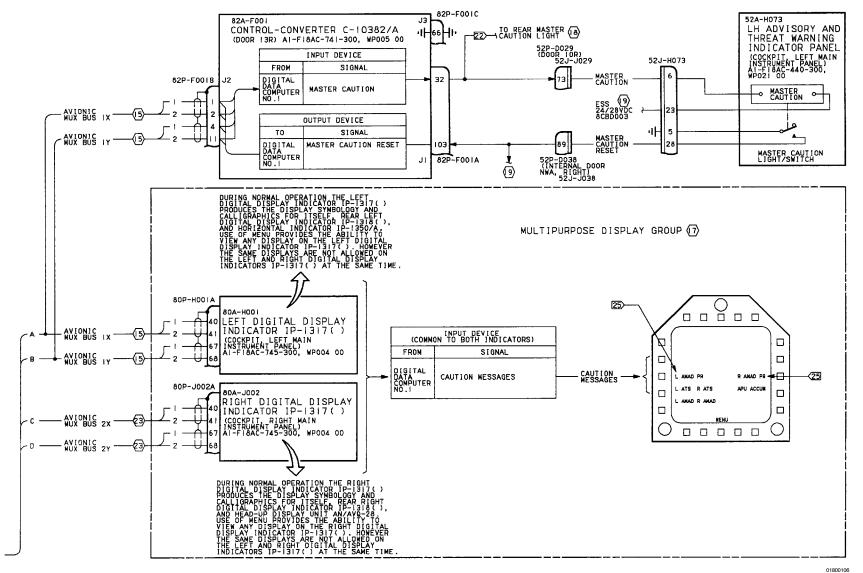


Figure 1.

Figure 1. Secondary Power System Interface Schematic (Sheet 6)

LEGEND

- I. CONTINUITY TESTS:

 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RXI SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - SHORTS TO GROUND STROUNDING PINS ON CONNECTORS. SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS. SHORTS BETWEEN SHIELD AND CONDUCTORS. SHIELD CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
 - DIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT,
 - ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE |.



- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO ALFIBRATION TO REFER TO ALFIBRAC FIN-100.
- (4) EXPLANATION OF MATRIX:
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE COMPUTED SIGNAL OUTPUT.
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED:
 - (!) ONE (!) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
 - (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.
- (5) APU CONTROL SYSTEM SCHEMATIC, AI-FIBAC-240-500, WP004 00.
- (6) APU LUBRICATION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP006 00
- (7) FLUIDS TEST SCHEMATIC, WP006 00.
- 8 APU FIRE EXTINGUISHING SYSTEM SCHEMATIC, A1-F18AC-240-500.

- APU START SYSTEM SCHEMATIC, AI-FIBAC-240-500, WP004 00.
 AMAD LUBE SYSTEM SCHEMATIC, AI-FIBAC-240-500, WP007 00.
 ANGINE AT ART AND GROUND MAINTENANCE MODE SCHEMATIC.
- INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, A1-F18AC-500-500, WPD13 00.
- POWER SCHEMATIC, WP005 00
- DURING APU START MODE, IF MISSION COMPUTER SYSTEM IS OPERATING. APU START DATA SENTENCE SOME COMPUTER SYSTEM FOR COMPUTER SYSTEM IS OPERATING SYSTEM IS OF COMPUTER SYSTEM IS OF COMPUTER SYSTEM IS NOT OPERATING SIGNAL DATA RECORDER COMPUTES APU MAINTENANCE CODES.
- (5) AYIONIC MUX CHANNEL I SCHEMATIC, AI-F18AC-741-500, WP004 00.
 (6) SECONDAY POMER SYNTEM CONTIONS AND MAINTENANCE CODES SCHEMA SECONDARY POWER SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATIC, A1-F18AC-240-500, WP005 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317() HEAD-UP DISPLAY UNIT AN AYG-28, HORIZONTAL INDICATOR IP-1350/A) AND ON F/A-188 THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1350/A) AND ON F/A-188 THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1350/A) REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1350/A) REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1350/A) FOR MULTIPURPOSE DISPLAY ROUP, REFER TO AT-186-745-50-196(). FOR MULTIPURPOSE DISPLAY ROUP, REFER TO AT-186-745-50-196().
- (18) REAR COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP007 00.
- (9) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- [7] F/A-18B PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- (23) AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FI8AC-741-500, WP005 00.
- 24) DELETED
- 25) WITH DIGITAL DATA COMPUTER NO.I CONFIG/IDENT NUMBER 84A AND UP.
- 26) 161353 THRU 161519 BEFORE F/A-18 AFC 27.
- [27] 161520 AND UP; ALSO 161353 THRU 161519 AFTER F/A-18 AFC 27.
- [28] 161353 THRU 161528 BEFORE F/A-18 AFC 26.
- 29 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 26.
- 30> 162445 AND UP
- DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS. ON MORE THAN ONE INDICATOR. IT MALFUNCTION EXISTS ONLY ON NOT INDICATOR. TROUBLESHOOT BY DOING DISPLAY TEST, AI-F18AC-745-200, WP004 00 (F/A-18A) OR WP005 00 (F/A-18B).
- 32 PART NUMBER 74B548017-105
- 33 PART NUMBER 74B548017-109
- 34) TANK PART NUMBER 33500002 AND 33500003
- 35 TANK PART NUMBER 826200-107.

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - SECONDARY POWER SYSTEM INTERFACE MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Secondary Power System Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

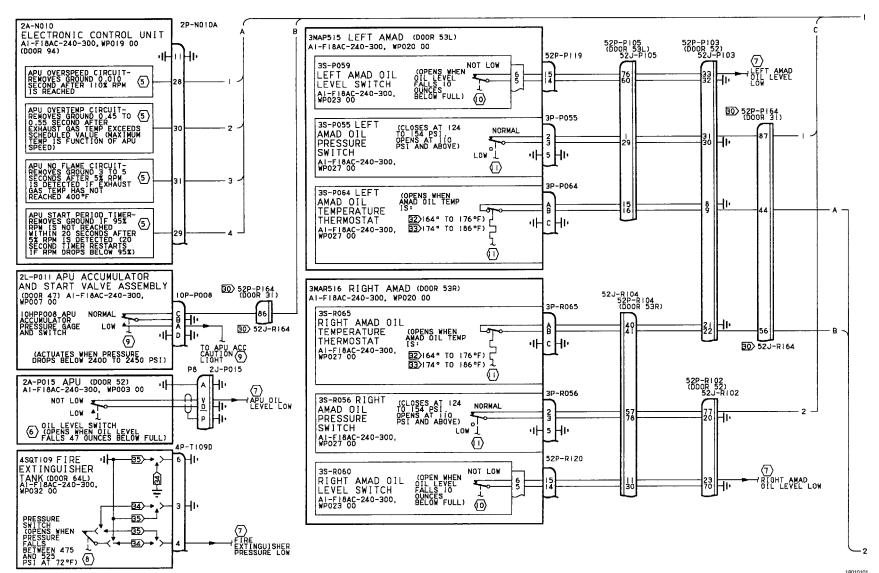
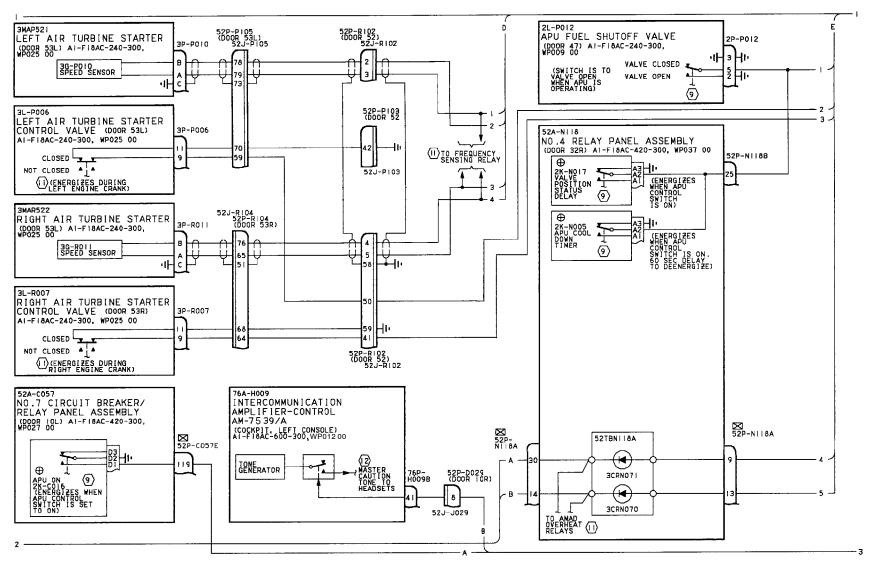


Figure 1.

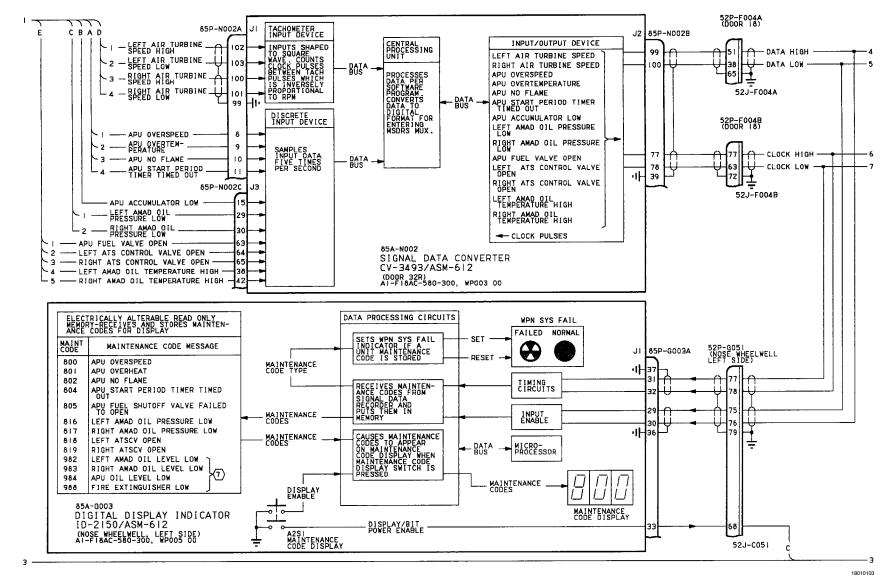
Figure 1. Secondary Power System Interface Schematic (Sheet 1)

Figure 1.



18010102 Figure 1.

Figure 1. Secondary Power System Interface Schematic (Sheet 2)



A1-F18AC-580-500

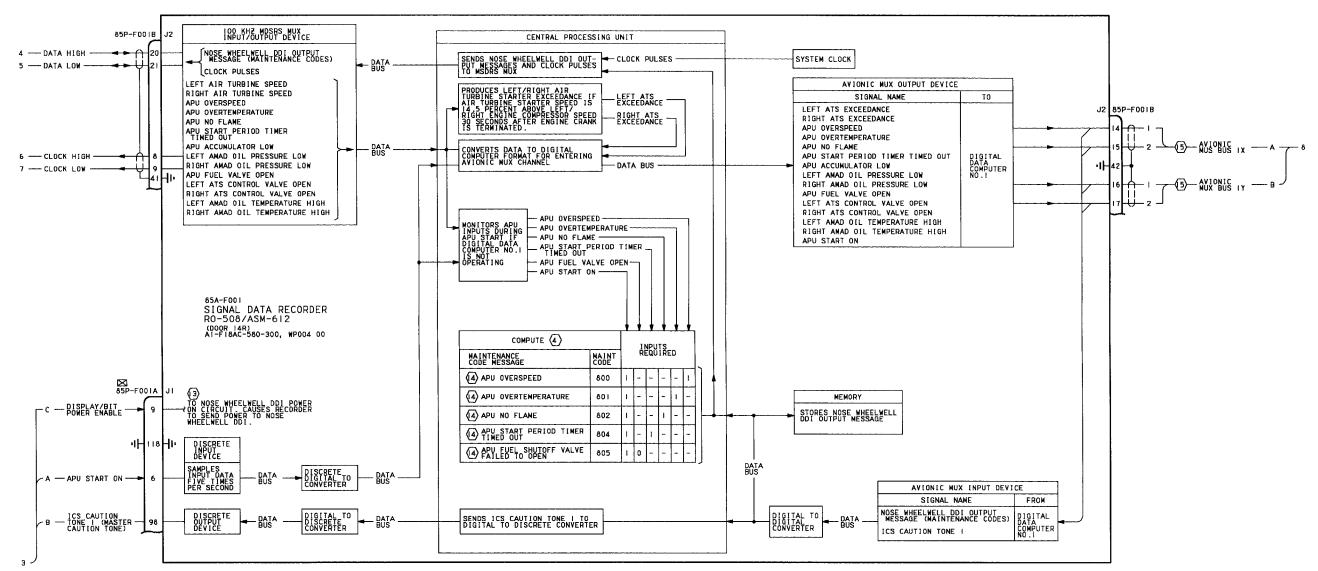


Figure 1.

Figure 1. Secondary Power System Interface Schematic (Sheet 4)

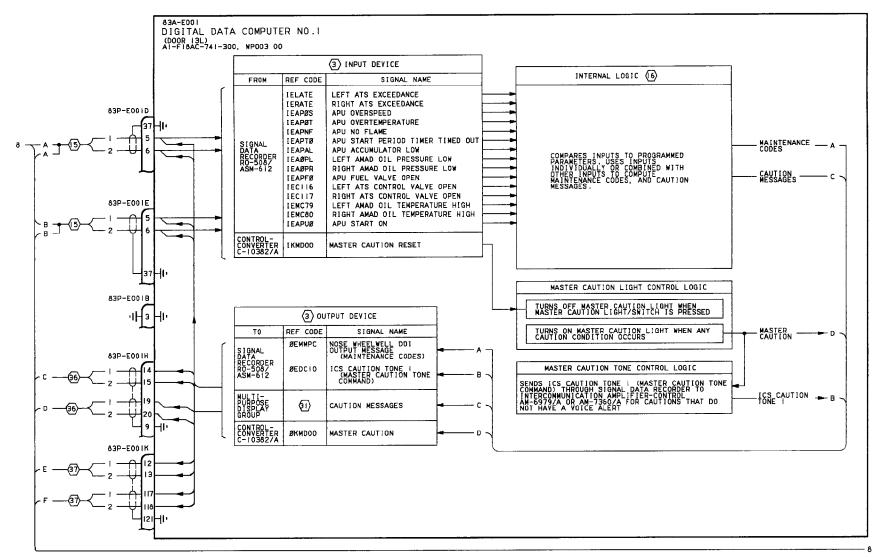


Figure 1.

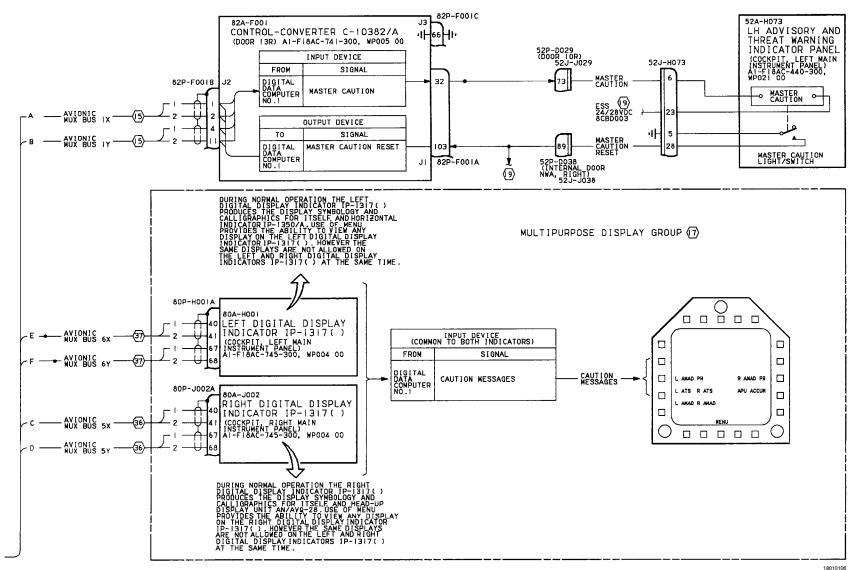


Figure 1.

Figure 1. Secondary Power System Interface Schematic (Sheet 6)

LEGEND

- I. CONTINUITY TESTS:

 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RXI SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - SHORTS TO GROUND STROUNDING PINS ON CONNECTORS. SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS. SHORTS BETWEEN SHIELD AND CONDUCTORS. SHIELD CONTINUITY.
- 2. NONSTANDARD SYMBOLS:
 - ## IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE | .
 - ☑ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE |.



- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO ALFIBRATION TO REFER TO ALFIBRAC FIN-100.
- (4) EXPLANATION OF MATRIX:
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE COMPUTED SIGNAL OUTPUT.
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED:
 - (!) ONE (!) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
 - (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.
- (5) APU CONTROL SYSTEM SCHEMATIC, AI-FIBAC-240-500, WP004 00.
- (6) APU LUBRICATION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP006 00
- (7) FLUIDS TEST SCHEMATIC, WP006 00.
- 8 APU FIRE EXTINGUISHING SYSTEM SCHEMATIC, A1-F18AC-240-500.

- APU START SYSTEM SCHEMATIC, A1-F18AC-240-500, WP004 00.
 AMAD LUBE SYSTEM SCHEMATIC, A1-F18AC-240-500, WP007 00.
 ANGINE AT AND GROUND MAINTENANCE MODE SCHEMATIC,
- INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, A1-F18AC-500-500, WPD13 00.
- POWER SCHEMATIC, WP005 00
- DURING APU START MODE, IF MISSION COMPUTER SYSTEM IS OPERATING. APU START DATA SENTENCE SOME COMPUTER SYSTEM FOR COMPUTER SYSTEM IS OPERATING SYSTEM IS OF COMPUTER SYSTEM IS OF COMPUTER SYSTEM IS NOT OPERATING SIGNAL DATA RECORDER COMPUTES APU MAINTENANCE CODES.
- (5) AYIONIC MUX CHANNEL I SCHEMATIC, AI-F18AC-741-500, WP004 00.
 (6) SECONDAY POMER SYNTEM CONTIONS AND MAINTENANCE CODES SCHEMA SECONDARY POWER SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATIC, A1-F18AC-240-500, WP005 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY UNIT ANA/MQ-28, HORIZONTAL INDICATOR IP-1350/A, FOR MULTIPURPOSE DISPLAY GROUP, REFER TO AI-F18AC-745-500.
- (8) DELETED.
- (9) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- 20) DELETED.
- DELETED.
- 22) DELETED.
- (23) DELETED.
- 24) DELETED
- 25> DELETED.
- 26) DELETED .
- 27> DELETED
- 28 DELETED .
- 29> DELETED .
- 30> 162445 AND UP
- DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS. ON AGRE THAN ONE INDICATOR TROUBLESHOOT USING MALFUNCTION EXISTS ONLY ON ONE INDICATOR TROUBLESHOOT BY DOING DISPLAY TEST. ALI-FIBAC-745-200, WP004 00.
- 32 PART NUMBER 748548017-105.
- 33 PART NUMBER 74B548017-109
- 34 TANK PART NUMBER 33500002 AND 33500003.
- 35) TANK PART NUMBER 826200-107.
- (36) AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-FIBAC-741-500, WPO18 00.
- (37) AVIONIC MUX CHANNEL 6 SCHEMATIC, A1-F18AC-741-500, WP01900.

Change 7 - 1 December 2000

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC-CANOPY, WINGFOLD, BOARDING LADDER, PITOT STATIC, GUN, ANTI-ICING, AND AIR INDUCTION SYSTEMS INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292, AND F/A-18B

This WP supersedes WP019 00, dated 1 October 1988.

Reference Material

None

Alphabetical Index

Subject	Page No
Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

None

A1-F18AC-580-500

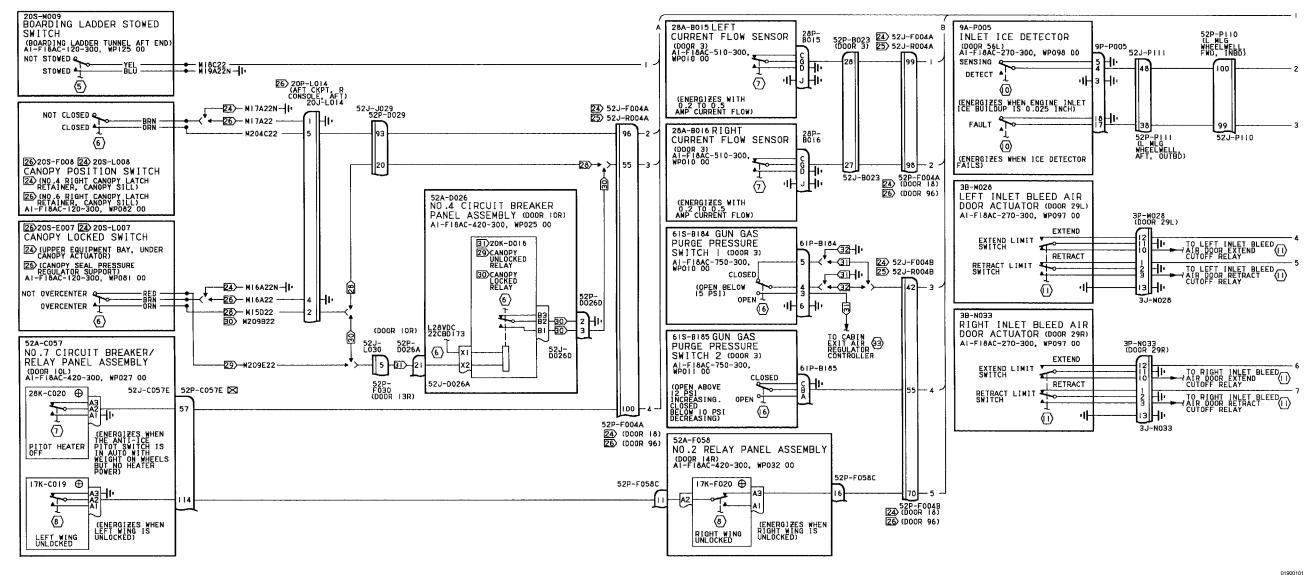
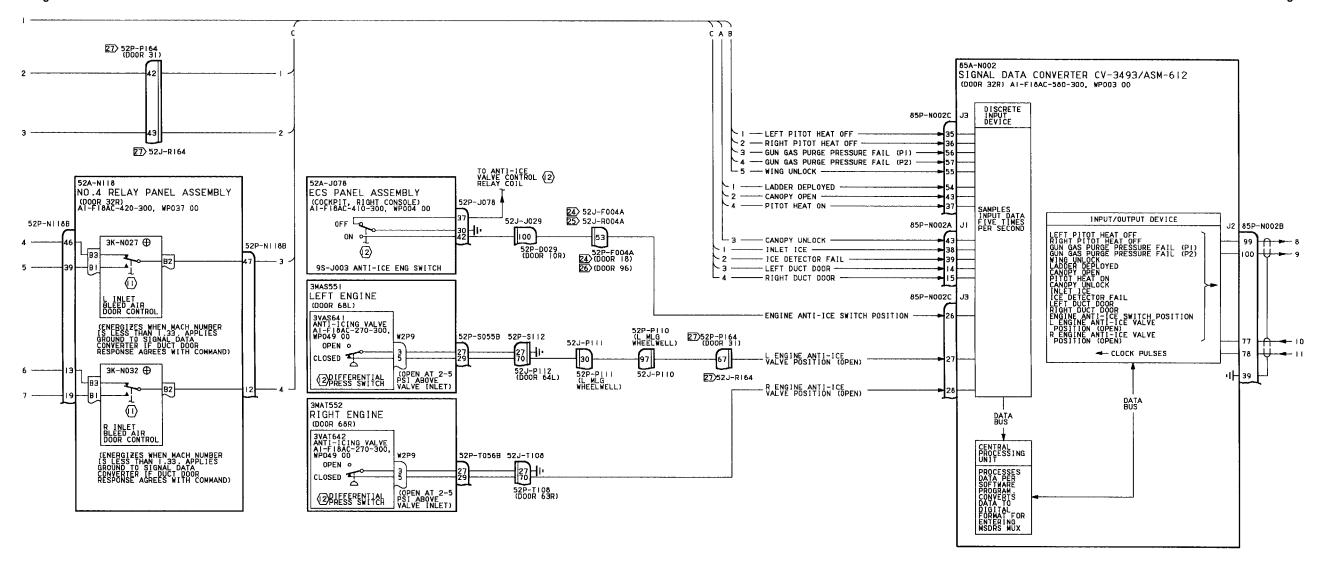


Figure 1.

Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 1)

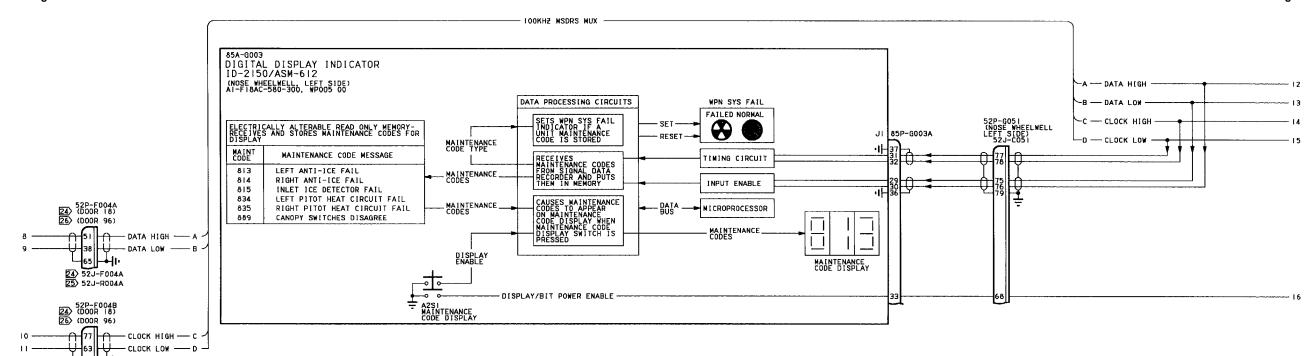
A1-F18AC-580-500 Change 7

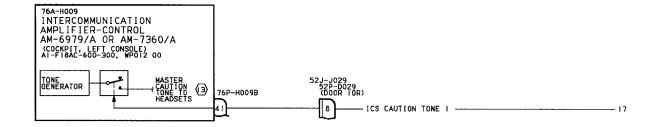


01900102 Figure 1.

Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 2)

24) 52J-F004B 25) 52J-R004B





A1-F18AC-580-500

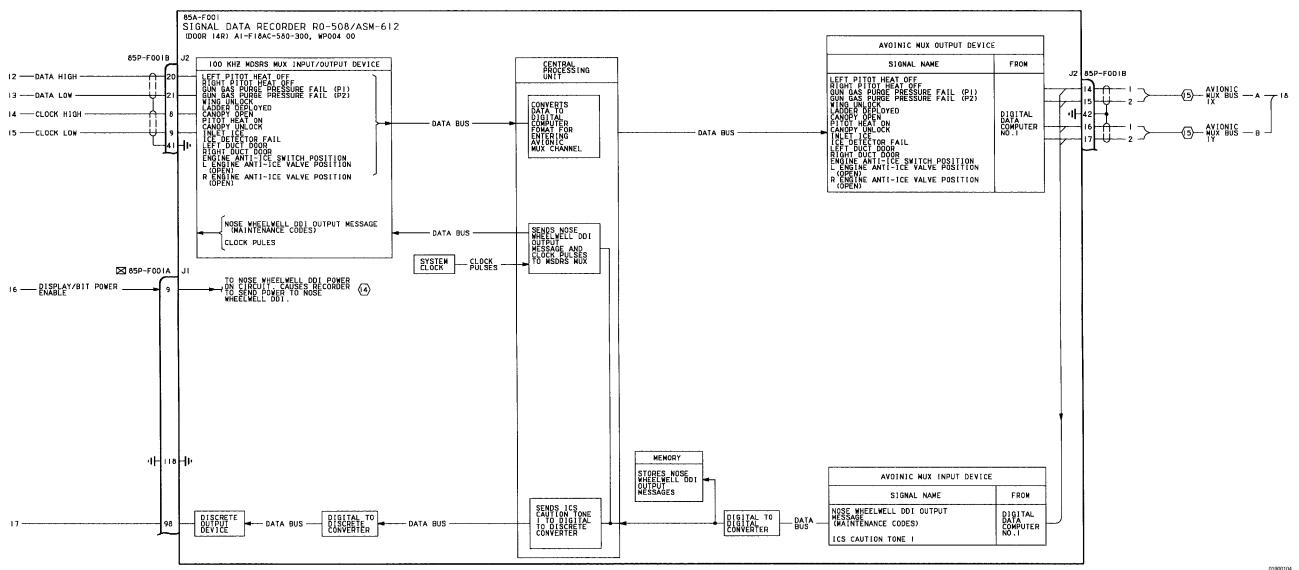
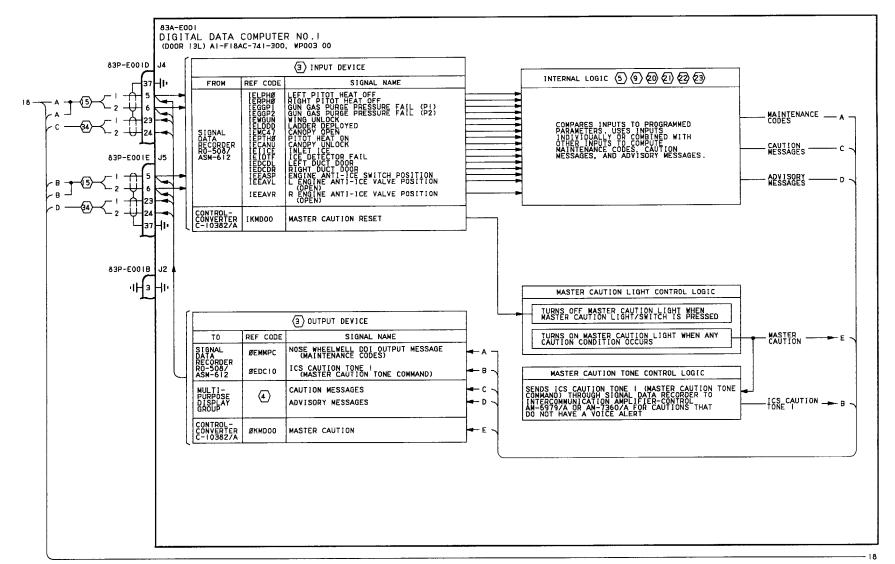


Figure 1.

Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 4)

Figure 1.



01900105

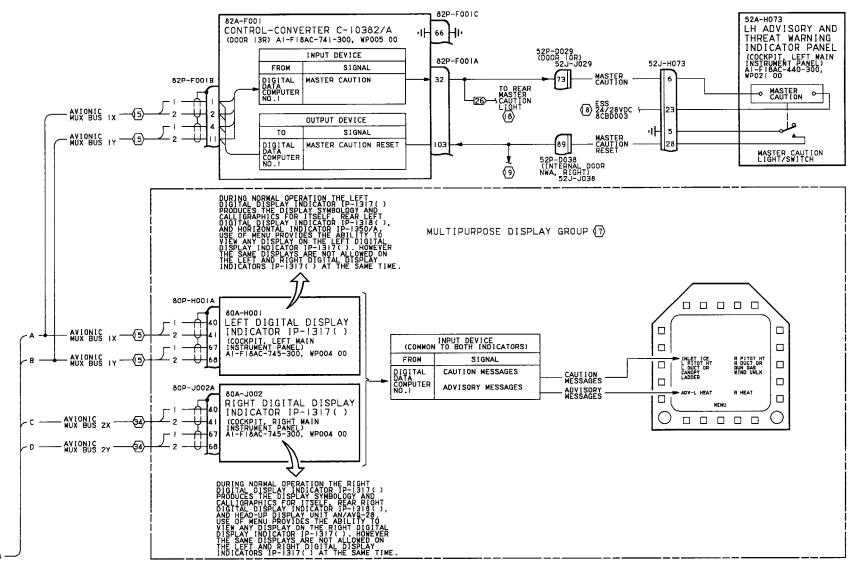
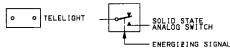


Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 6)

LEGEND

I. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ①), IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SMITCHING RELAY MITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON TAXES MAY USE IN TO PIN TEST THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX! SCALE.
- D. WHEN TESTING FOR CONTINUITY, TEST FOR:
 - (I) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, Z4VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY). MAKE SURE MULTIWETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - DIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE 1.
 - ☑IDENTIFIES 24YDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR, SEE NOTE 1.



- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AL-FL8A()-OLD-000.

 AL-FLBAC-FLM-LOG.
- 4 DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS THAN TRANSFER INDICATOR OF THE THAN TONE THAN TONE
- (5) BOARDING LADDER SYSTEM SCHEMATIC, A1-F18AC-120-500, WP009 00.
- (6) CANOPY SYSTEM SCHEMATICS, AI-FIBAC-120-500, WP006 00 AND WP007 00.
- (7) PITOT STATIC SYSTEM HEATERS SCHEMATIC, AI-F!8AC-510-500, WP003 00.
- (8) WING FOLD SYSTEM SCHEMATIC, AI-FIBAC-570-500, WP027 00
- AIR TO AIR GUN AVIONIC INTERFACE SCHEMATIC, AI-F18AC-750-500, WP005 00: OF AIR TO GROUND GUN AVIONIC INTERFACE SCHEMATIC, AI-F18AC-750-500, WP005 00.
- (10) INLET ICE DETECTOR SYSTEM SCHEMATIC, A1-F18AC-270-500, WP009 00.
- (I) INLET BLEED AIR DOOR SYSTEM SCHEMATIC, AI-FISAC-270-500, WP009 00.
- (IZ) ANTI-ICE SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP005 00.
- (3) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- (14) POWER SCHEMATIC, WP005 00.

- (5) AVIONIC MUX CHANNEL I SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- (6) GUN SYSTEM SCHEMATIC, AI-FIBAC-750-500, WP004 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(). RIGHT DIGITAL DISPLAY INDICATOR IP-1317(). AND ON F.A-180 THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1350/A. AND ON F.A-180 THE REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318() REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318() PROPERTY OF THE PROPER
- (8) REAR COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FISAC-440-500, WP007 00.
- (19) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- (20) WINGFOLD SYSTEM CAUTION SCHEMATIC, AI-FISAC-570-500, WP027 00.
- PITOT STATIC SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATIC,
- (2) INLET ICE, DUCT DOOR, AND HEAT CAUTIONS, ADVISORIES, AND MAINTENANCE CODES SCHEMATIC, AI-F18AC-270-500, WP010 00.
- (3) CANDPY SYSTEM MAINTENANCE CODES AND CAUTION SCHEMATICS, AI-FI8AC-120-500,
- 24) F/A-18A.
- 25 F/A-188 PLUG AND JACK POSITIONS ARE REVERSED FROM POSITION SHOWN.
- 26> F/A-18B.
- 27) 162445 AND UP.
- 28) F/A-18A; F/A-18B 161354 THRU 162885.
- 29>F/A-18A 163092 AND UP.
- 30 F/A-18B 163104 AND UP.
- BT> 163092 AND UP.
- 32 161353 THRU 162909.
- 3 AVIONIC COOLING SYSTEM SCHEMATIC-EXCEPT COCKPIT, AI-F18AC-410-500,
- (34) AVIONIC MUX CHANNEL 2 SCHEMATIC, A1-F18AC-741-500, WP005 00.

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC-CANOPY, WINGFOLD, BOARDING LADDER, PITOT STATIC, GUN, ANTI-ICING, AND AIR INDUCTION SYSTEMS INTERFACE

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction	
Systems Interface Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

A1-F18AC-580-500

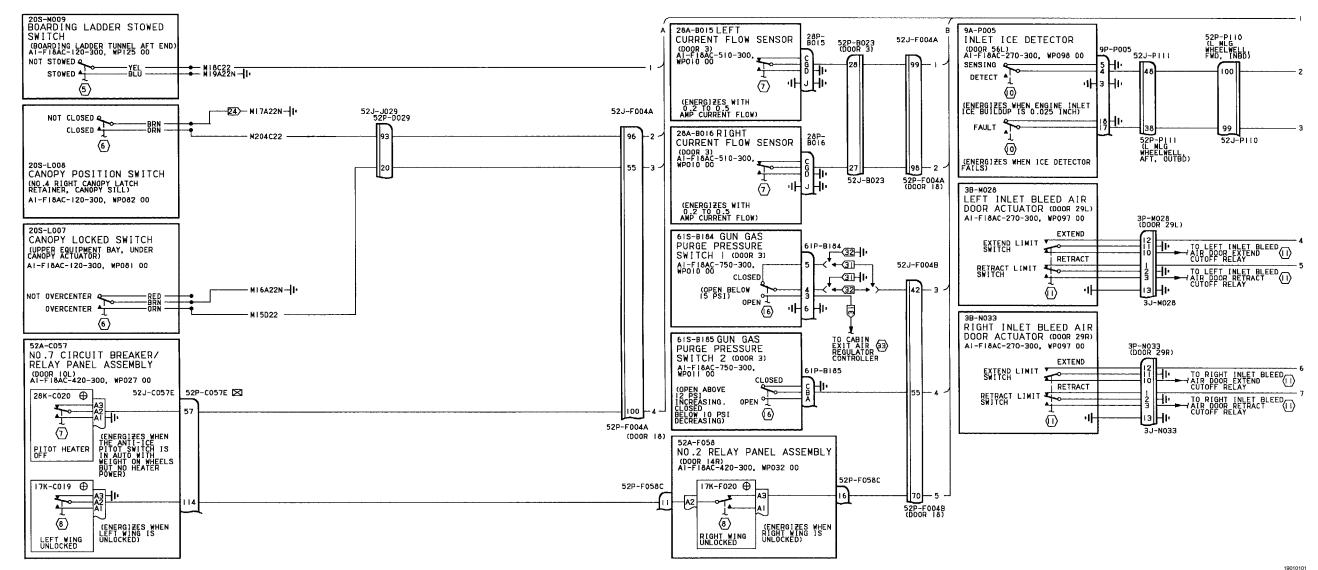


Figure 1.

Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 1)



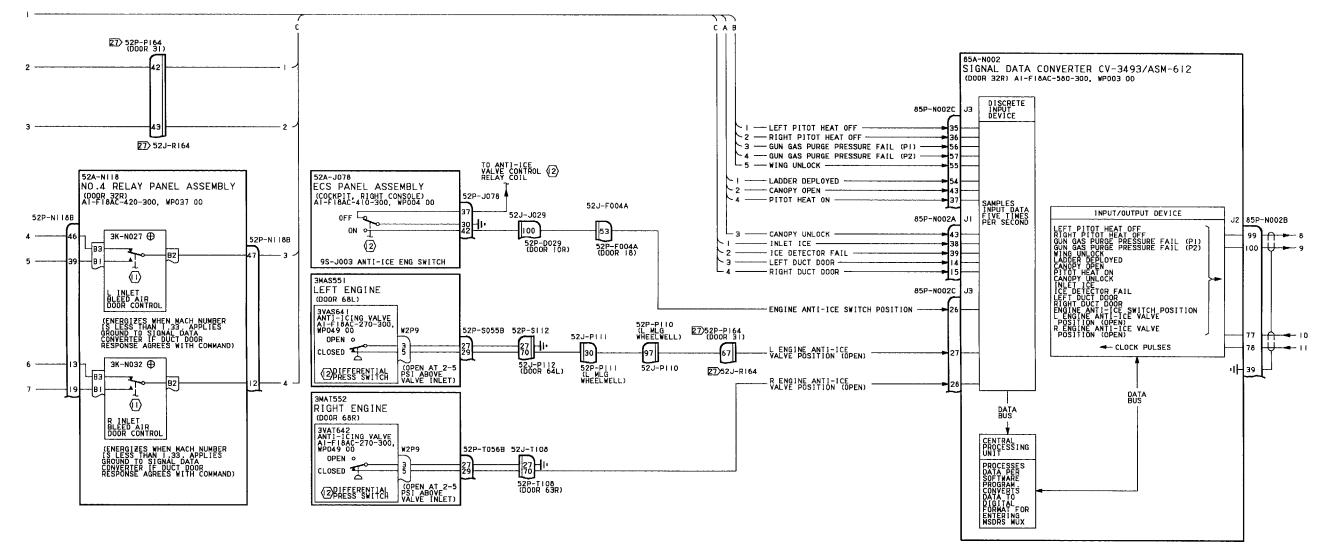
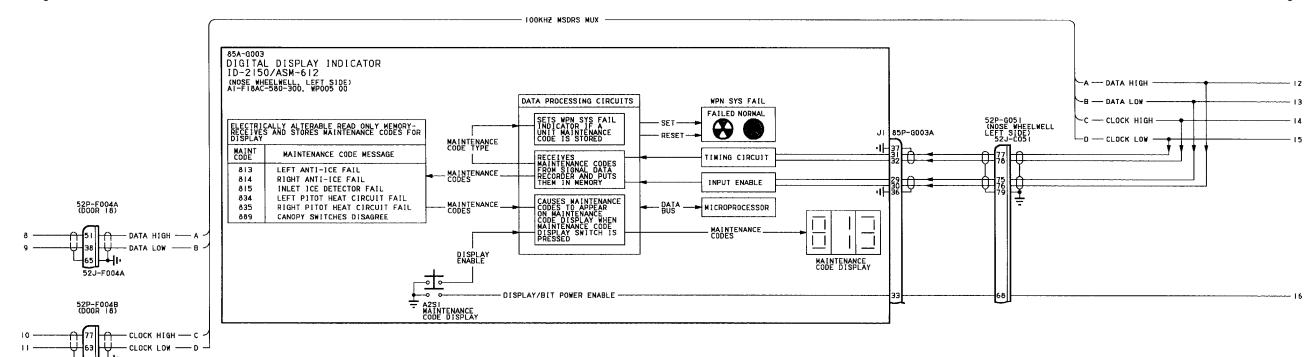
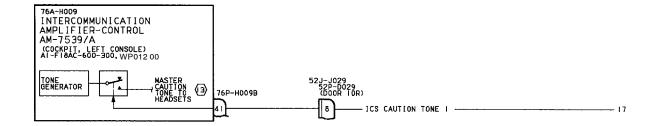


Figure 1.

Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 2)

52J-F004B





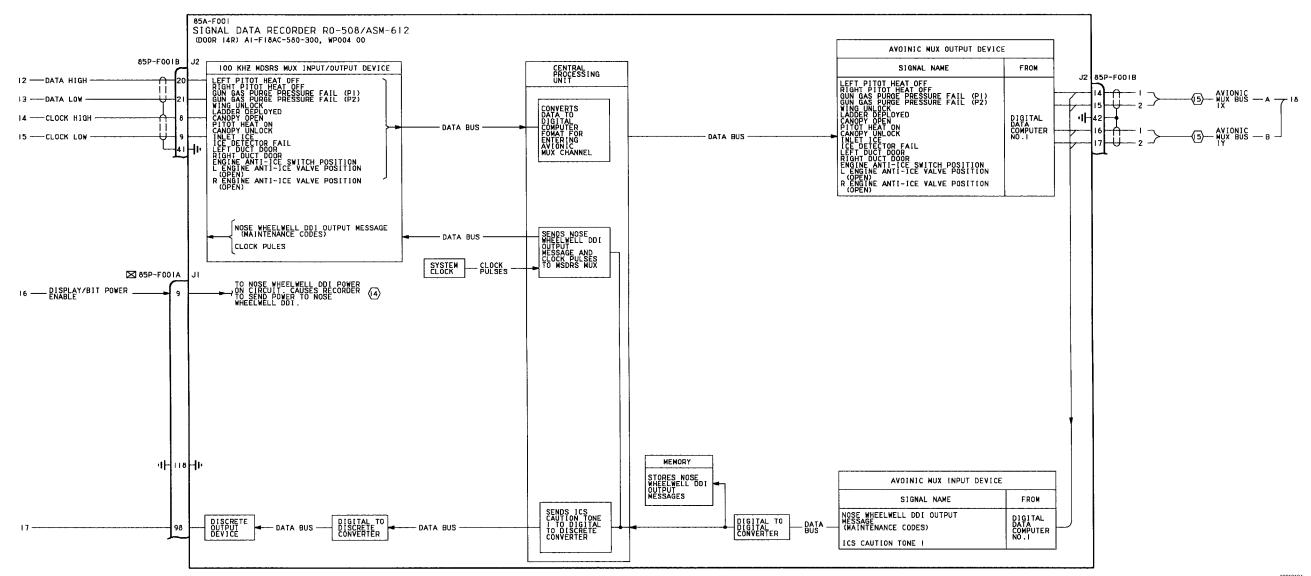
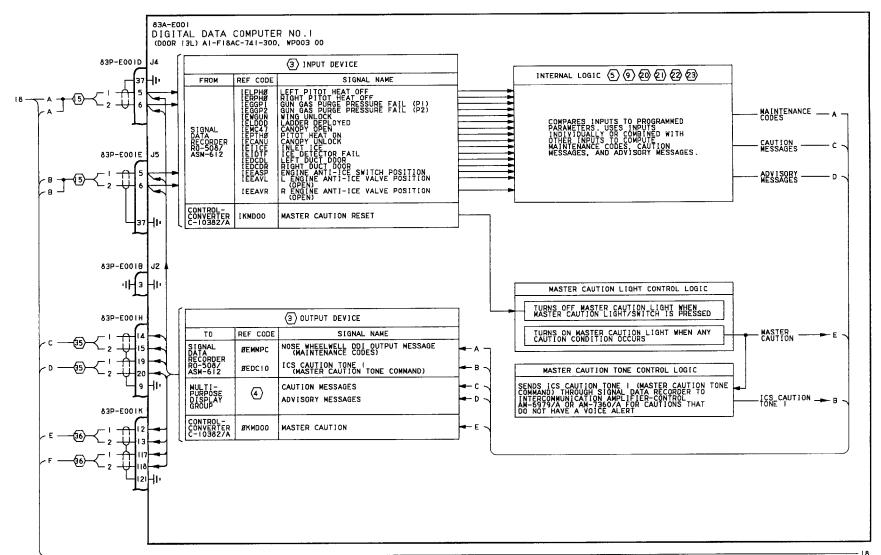
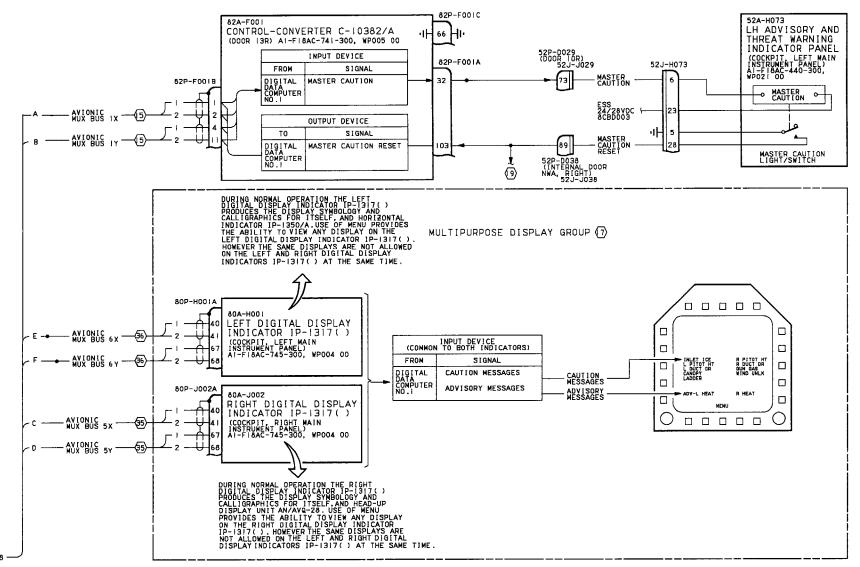


Figure 1.

Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 4)



19010105



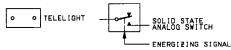
19010106

Figure 1. Figure 1. Canopy, Wingfold, Boarding Ladder, Pitot Static, Gun, Anti-Icing and Air Induction Systems Interface Schematic (Sheet 6)

LEGEND

I. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING. IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SMITCHES/RELAY CONTACTS) FOR CONTINUITY SMITCHES/RELAY CONTACTS MAY USE THE RX! SCALE.
- D. WHEN TESTING FOR CONTINUITY, TEST FOR:
 - (I) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, Z4VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY). MAKE SURE MULTIWETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS
 - DIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE 1.
 - ☑IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE | 1.



- (3) FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AL-FIBA()-OLD-OOD.

 FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO
 AL-FIBAC-FIM-100.
- DISPLAY REF ACORS ARE NOT SHOWN. IF MALFUNCTION EXISTS ATRANSFER INDICATOR TROUBLESHOOT USING AI-FIBA()-CLD-SOO INDICATOR TROUBLESHOOT USING AI-FIBA()-CLD-SOO INDICATOR TROUBLESHOOT USING AI-FIBA()-CLD-SOO INDICATOR TROUBLESHOOT USING AI-FIBA()-CLD-SOO INDICATOR TROUBLESHOOT BY COING I
- (5) BOARDING LADDER SYSTEM SCHEMATIC, AI-FIBAC-120-500, WP009 00.
- (6) CANOPY SYSTEM SCHEMATICS, AI-FIBAC-120-500, WP006 00 AND WP007 00.
- (7) PITOT STATIC SYSTEM HEATERS SCHEMATIC, AI-F!8AC-510-500, WP003 00.
- (8) WING FOLD SYSTEM SCHEMATIC, AI-FIBAC-570-500, WP027 00
- (10) INLET ICE DETECTOR SYSTEM SCHEMATIC, A1-F18AC-270-500, WP009 00.
- (I) INLET BLEED AIR DOOR SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP009 00.
- (2) ANTI-ICE SYSTEM SCHEMATIC, AI-FIBAC-270-500, WP005 00.
- (3) INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, AI-F18AC-600-500, WP013 00.
- (14) POWER SCHEMATIC, WP005 00.

- (5) AVIONIC MUX CHANNEL ! SCHEMATIC, AI-F18AC-741-500, WP004 00.
- (6) GUN SYSTEM SCHEMATIC, AI-FIBAC-750-500, WP004 00.
- THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD-UP DISPLAY UNIT AN/AVQ-28, HORIZONTAL INDICATOR IP-1350/A, FOR MULTIPURPOSE DISPLAY GROUP, REFER TO AI-F18AC-745-500.
- (8) DELETED.
- (19) COCKPIT CAUTION LIGHTS SCHEMATIC, AI-FIBAC-440-500, WP006 00.
- (a) WINGFOLD SYSTEM CAUTION SCHEMATIC, A1-F18AC-570-500, WP027 00
- PITOT STATIC SYSTEM CAUTIONS AND MAINTENANCE CODES SCHEMATIC.
- [2] INLET ICE, DUCT DOOR, AND HEAT CAUTIONS, ADVISORIES, AND MAINTENANCE CODES SCHEMATIC, AI-F18AC-270-500, WP010 00.
- (3) CANDRY SYSTEM MAINTENANCE CODES AND CAUTION SCHEMATICS, AI-FI8AC-120-500,
- 24) DELETED.
- 25 DELETED.
- 26 DELETED.
- 27 162445 AND UP.
- 28 DELETED.
- 29 DELETED.
- 30) DELETED.
- DT) 163092 AND UP.
- 32 161353 THRU 162909
- 3 AVIONIC COOLING SYSTEM SCHEMATIC-EXCEPT COCKPIT, AI-FI8AC-410-500,
- (4) DELETED.
- (35) AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-FIBAC-741-500, WPOIB 00.
- (36) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - MAINTENANCE CODE CLEAR AND INHIBIT

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: WITH DIGITAL DATA COMPUTER NO. 1 CONFIG/IDENT NUMBER 84A AND UP

This WP supersedes WP020 00, dated 1 May 1986.

Reference Material

None

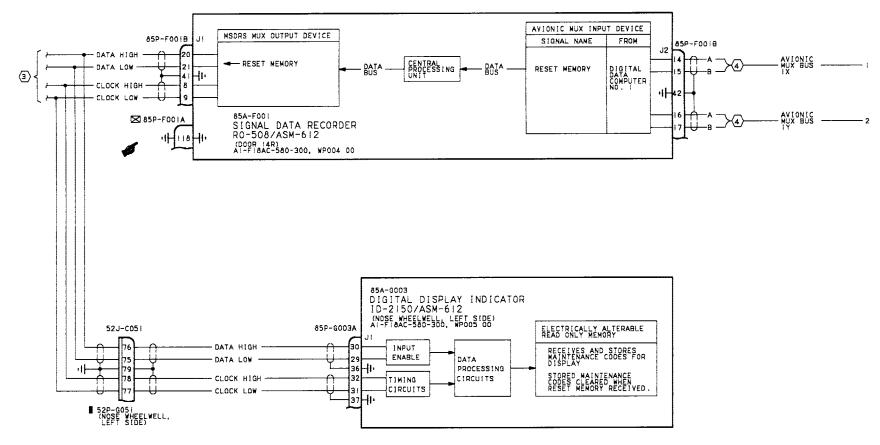
Alphabetical Index

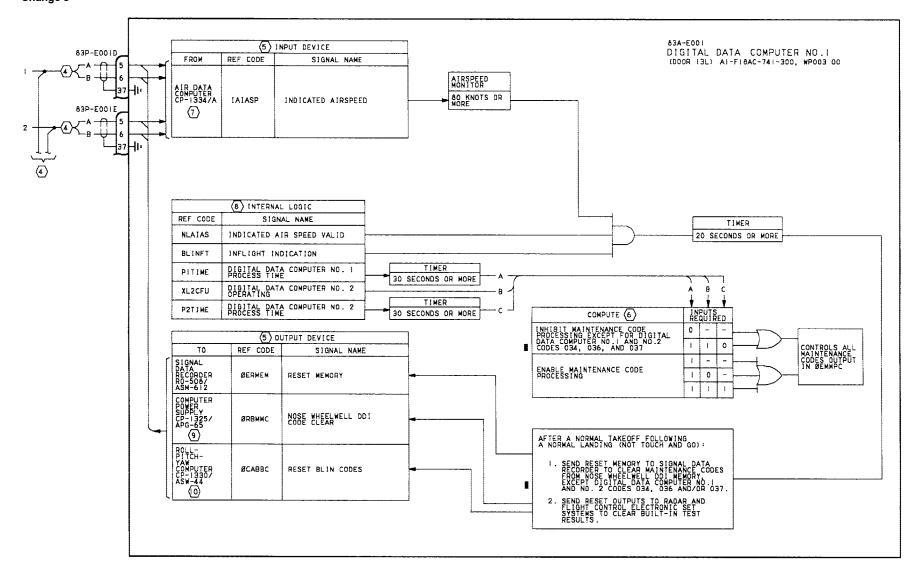
Subject	Page No
Maintenance Code Clear and Inhibit Schematic, Figure 1	2

Record of Applicable Technical Directives

None







LEGEND

- I. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-FIBA()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ①) IS REMOYED FOR TROUBLESHOOTING. IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - DO NOT TEST LOW LEYEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RYI SCALE, PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (I). SHORTS TO GROUND.
 - (2) . SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) . SHORTS BETWEEN SHIELD AND CONDUCTORS .
 - (4) . SHIELD CONTINUITY.
- E. WHEN ELECTRICAL DOWER IS OFF 24YDC BATTERY YOU TAGE EXTSTS ON SOME JUNEAR WIRES ARE INSTALLED FOR CORRECT PINS WHEN TESTING FOR CONTINUITY
- 2. NONSTANDARD SYMBOLS
 - 1 DENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE 1.
- ☑ 1DENTIFIES 24 VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE i.
- (3) POWER SCHEMATIC, WP005 00.
- 4 AVIONIC MUX CHANNEL | SCHEMATIC.AI-F18AC-741-500, WPD04 00.
- (5) FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AI-FIBAC-OLD-GOD, FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100.
- (6) EXPLANATION OF MATRIX:
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - 8. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS SHOWN:
 - (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - (2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE
 - (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.
- 7 AIR DATA COMPUTER SYSTEM SCHEMATIC, AI-FIBAC-560-500, WP004 00
- REF CODES USED FOR THESE COMPUTATIONS ARE MISSION COMPUTER INTERNAL NEW CODES IN AI-FIBAC-OLD-000, USE THE LOGIC DIAGRAMS FOR THE INPUT/OUTPUT REF CODES.
- (9) RADAR INITIATED BUILT-IN TEST FAULT INDICATION SCHEMATIC, AI-F18AC-742-500,
- (0) CAUTIONS AND BUILT-IN TESTS DISPLAYS SCHEMATIC, A1-F18AC-570-500, WP024 00.

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - MISSION DATA LOADER MISSION INITIALIZATION FUNCTIONAL

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18 AFTER F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18 AFC 225 WITH DIGITAL DATA COMPUTER CONFIG/IDENT 15C AND UP (A1-F18AC-SCM-000)

Reference Material

None

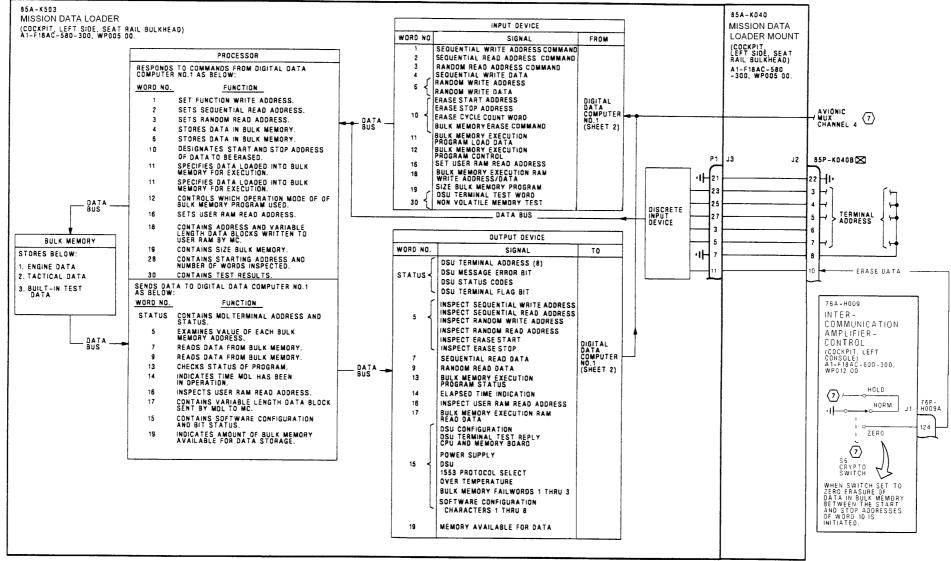
Alphabetical Index

Subject	Page No.
Mission Data Loader Mission Initialization Functional Schematic, Figure 1	2

Record of Applicable Technical Directives

		• •		
Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Avionics Multiplex Bus Upgrade, Modification of (ECP MDA-F/A-18-0529)	1 Jun 02	-
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

A1-F18AC-580-500 Change 7



A1-F18AC-580-500 Change 7

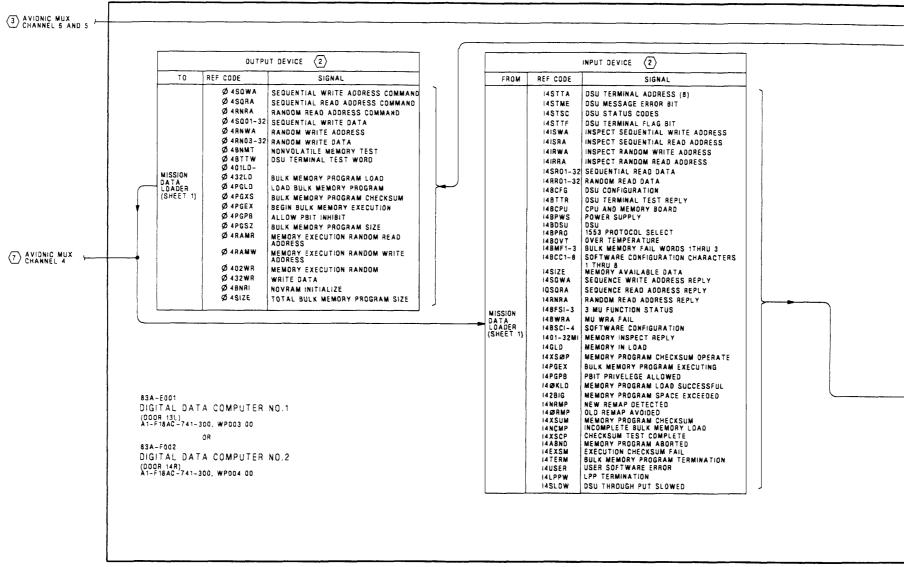


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 2)

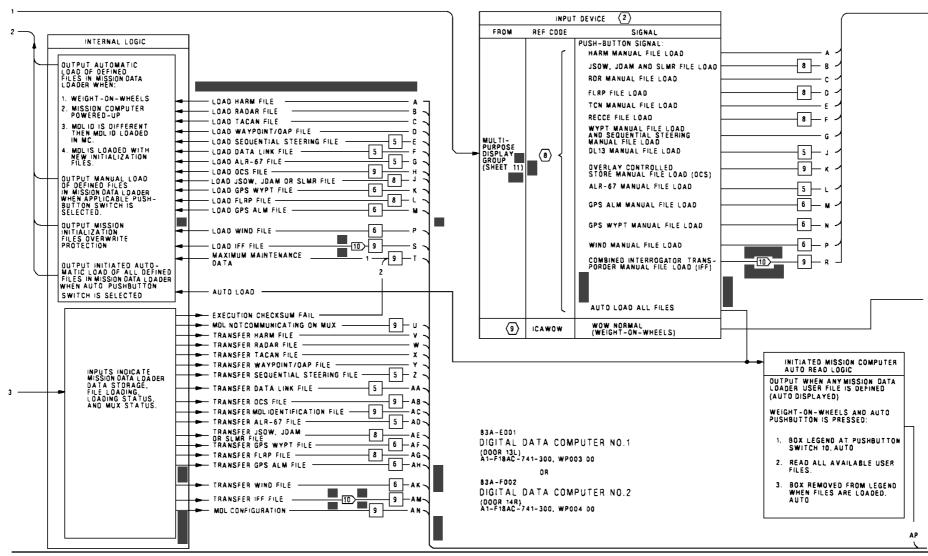


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 3)

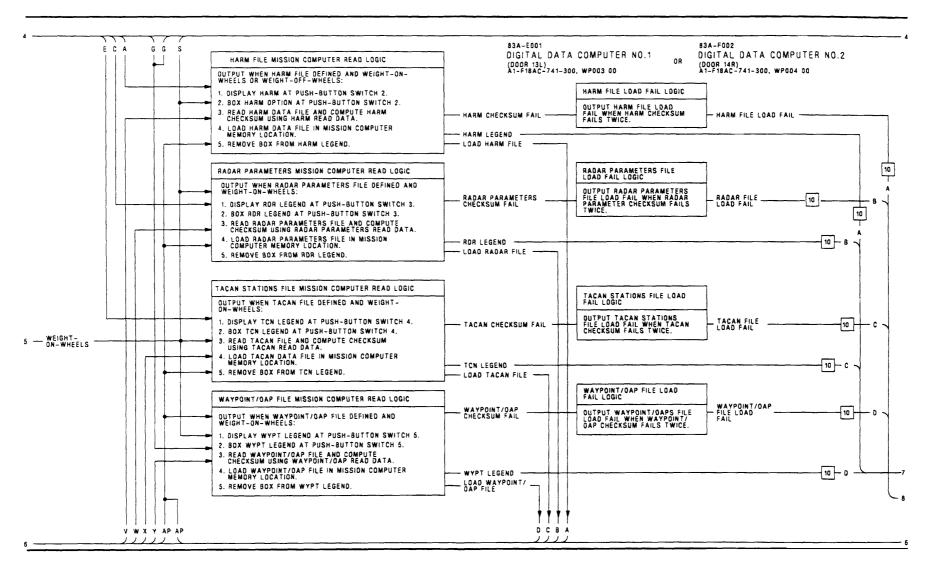


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 4)

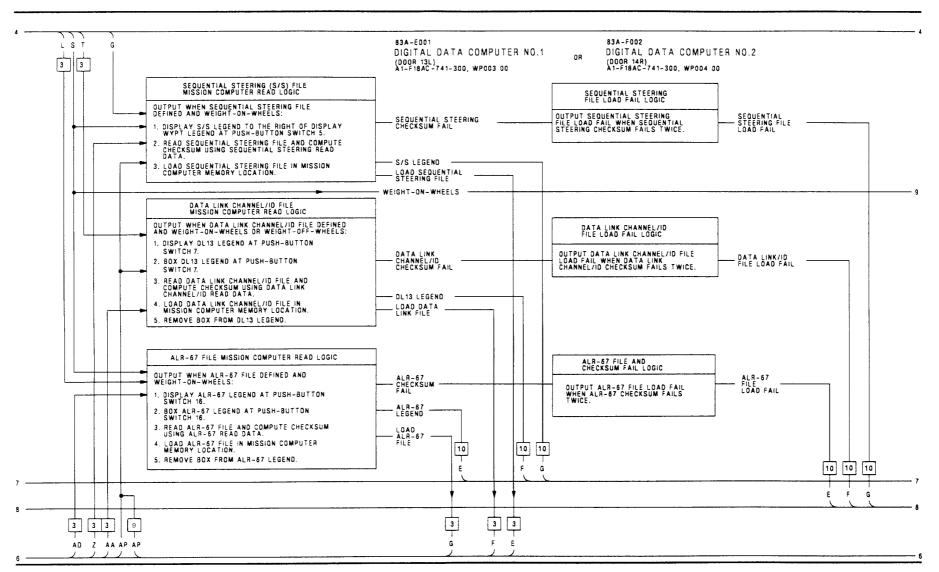


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 5)

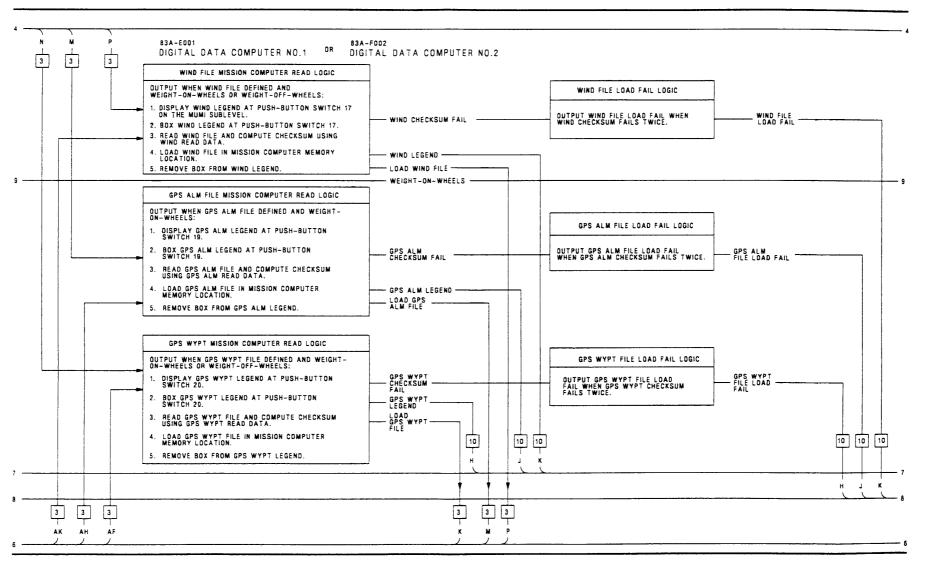


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 6)

WEIGHT-ON-WHEELS	83A-E001 DIGITAL DATA COMPUTER NO.1 OR DIGITAL DATA COMPUTER NO.2	
WEIGHT-ON-WHEELS		

Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 7)

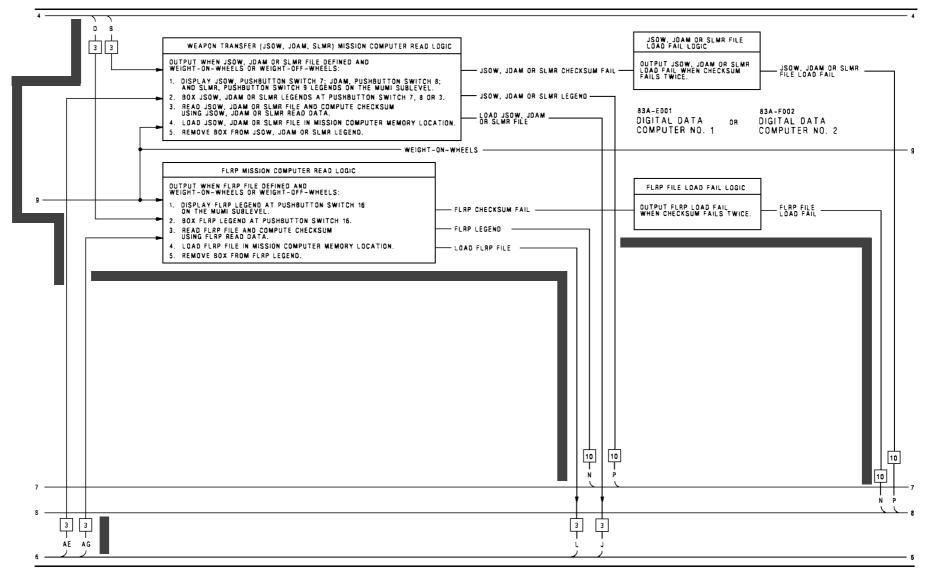


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 8)

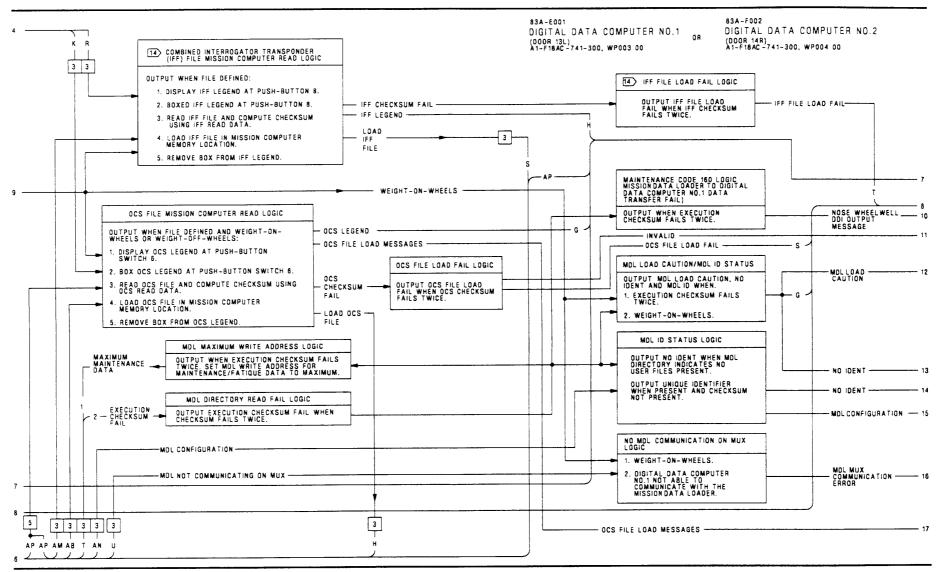


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 9)

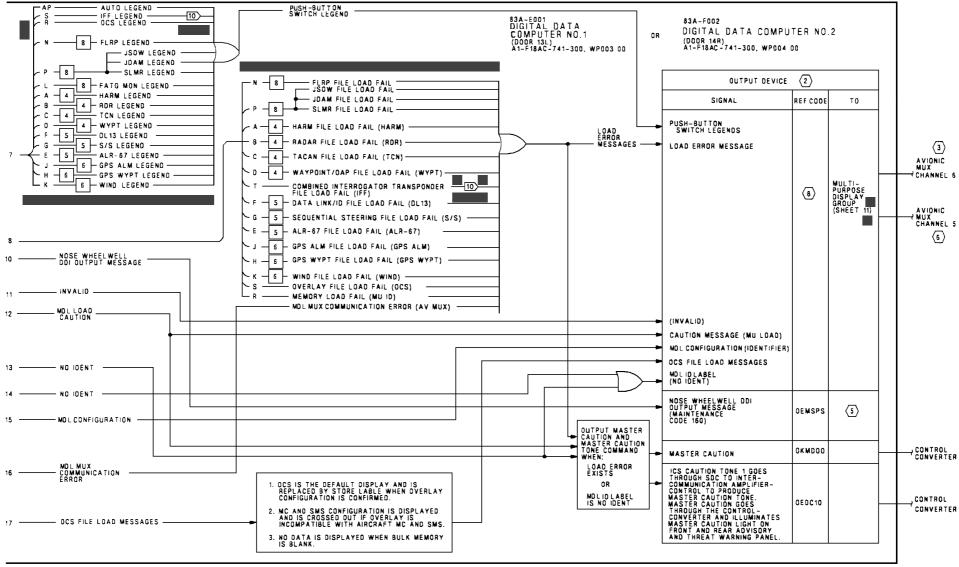


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 10)

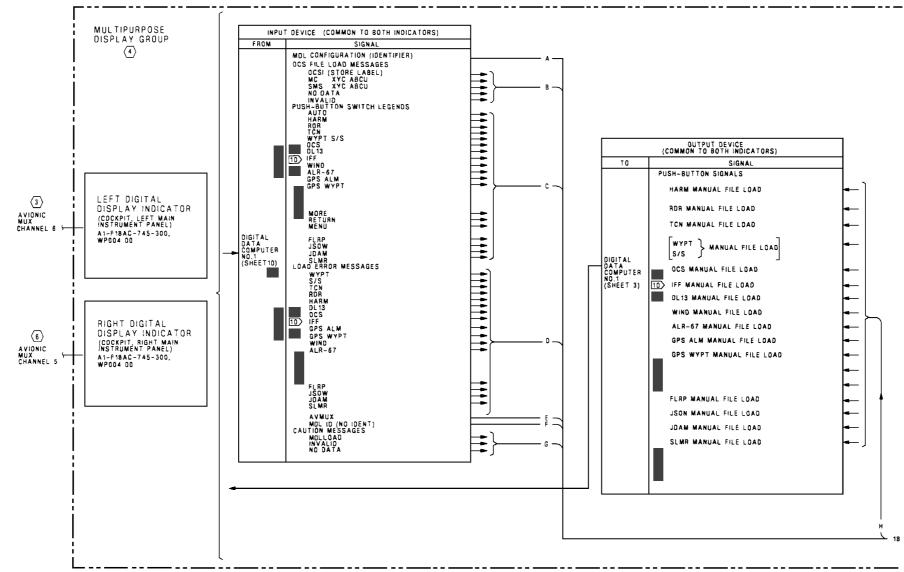


Figure 1.

Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 11)

Figure 1.

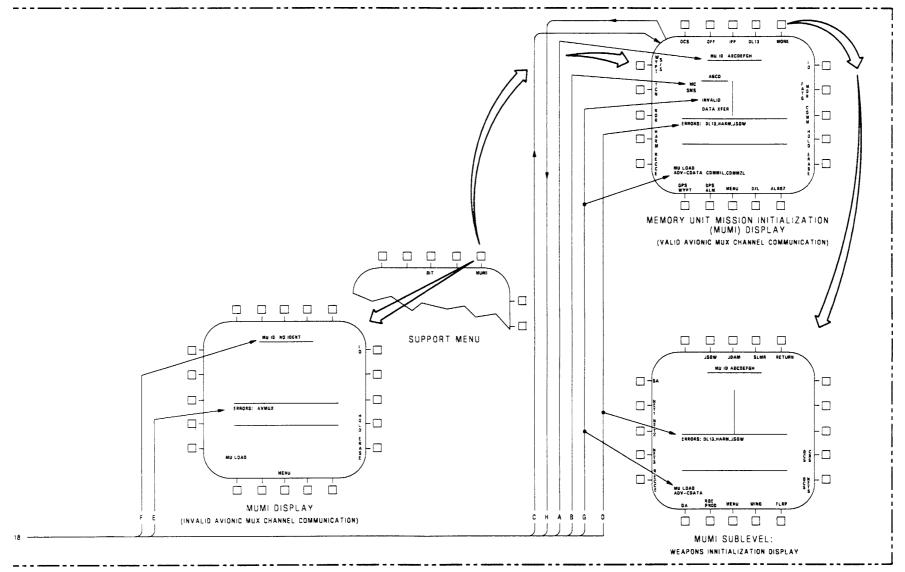


Figure 1. Mission Data Loader Mission Initialization Functional Schematic (Sheet 12)

LEGEND

CONTINUITY TESTS:

A. AIRCRAFT WIRE NUMBERS. SPLICE POINTS AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENITIFIED BY #) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. WHEN TESTING CONTINUITY TEST FOR:

(1) SHORTS TO GROUND.

(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4) SHIELD CONTINUITY.

D. WHEN ELECTRICAL POWER IS OFF. 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY 図). MAKE SURE MULTIM-ETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.

 $\binom{3}{4}$ AVIONIC MUX CHANNEL 6 SCHEMATIC, A1-F18AC-741-500, WP019 00.

MULTIPURPOSE DISPLAY GROUP INTERCONNECT SCHEMATIC, REFER TO A1-F18AC-745-500.

 $\langle 5 \rangle$ REF CODES FOR SYSTEM BUILT-IN TEST INPUTS ARE SHOWNON INDIVIDUAL SYSTEM SCHEMATICS WHEN DIGITAL DATA COMPUTER NO. 1 DETERMINES A MAINTENANCE CODE CONDITION EXISTS. PROCESSING IS THE SAME FOR ALL MAINTENANCE CODES

(6) (7) AVIONIC MUX CHANNEL 5 SCHEMATIC, A1-F18AC-741-500, WP018 00.

AVIONIC MUX CHANNEL 4 SCHEMATIC, A1-F18AC-741-500, WP017 00.

DISPLAY REF CODES ARE NOT SHOWN. IF DISPLAY MALFUNCTION EXISTS. TRANSFER DISPLAY TO ANOTHER INDICATOR. IF MALFUNCTION EXISTS ON MORE THAN ONE INDICATOR, REFER TO A1-F18AC-FRM-000. IF MALFUNCTION EXISTS ONLY ON ONE INDICATOR, TROUBLESHOOT BY DOING DISPLAY TEST A1-F18AC-745-200, WP004 00.

 $\langle 9 \rangle$ CROSS CHANNEL/MUX BUS/DISPLAYS FUNCTIONAL SCHEMATIC, A1-F18AC-570-500, WP021 01.

10 AFTER F/A-18 AFC 292.

Page 1

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - MISSION DATA LOADER BUILT-IN TEST

MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM

EFFECTIVITY: F/A-18 AFTER F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18 AFC 225

This WP supersedes WP022 00, dated 1 December 2000.

Reference Material

None

Alphabetical Index

Subject	Page No.
Mission Data Loader Built-In Test Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Avionics Multiplex Bus Upgrade, Modification of, (ECP MDA-F/A-18-0529)	1 Jun 02	-
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Dec 00	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade, Incorporation of, (ECP MDA-F/A-18-0583)	1 Dec 00	-

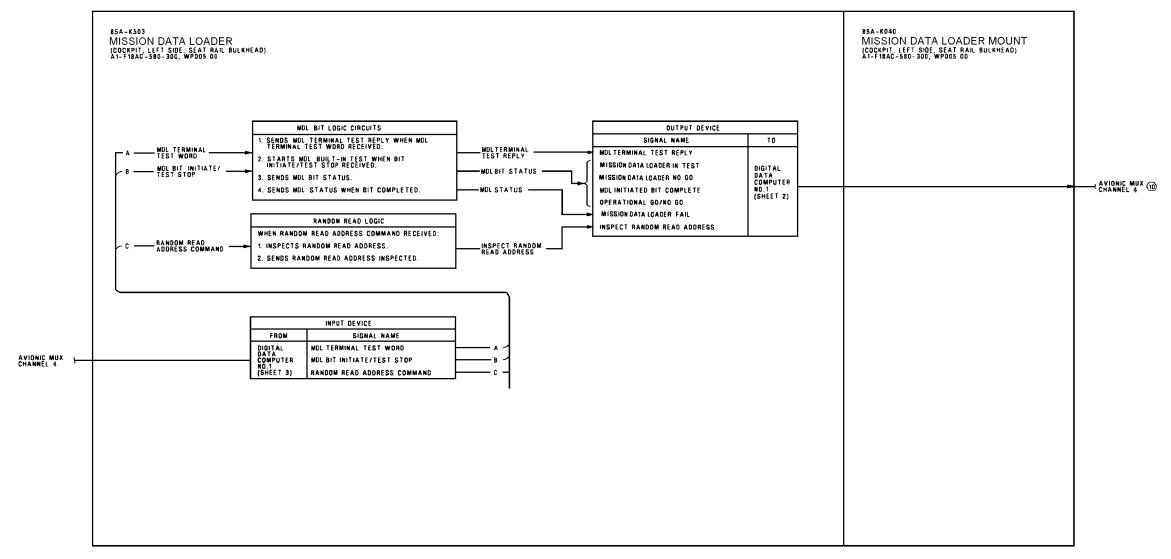
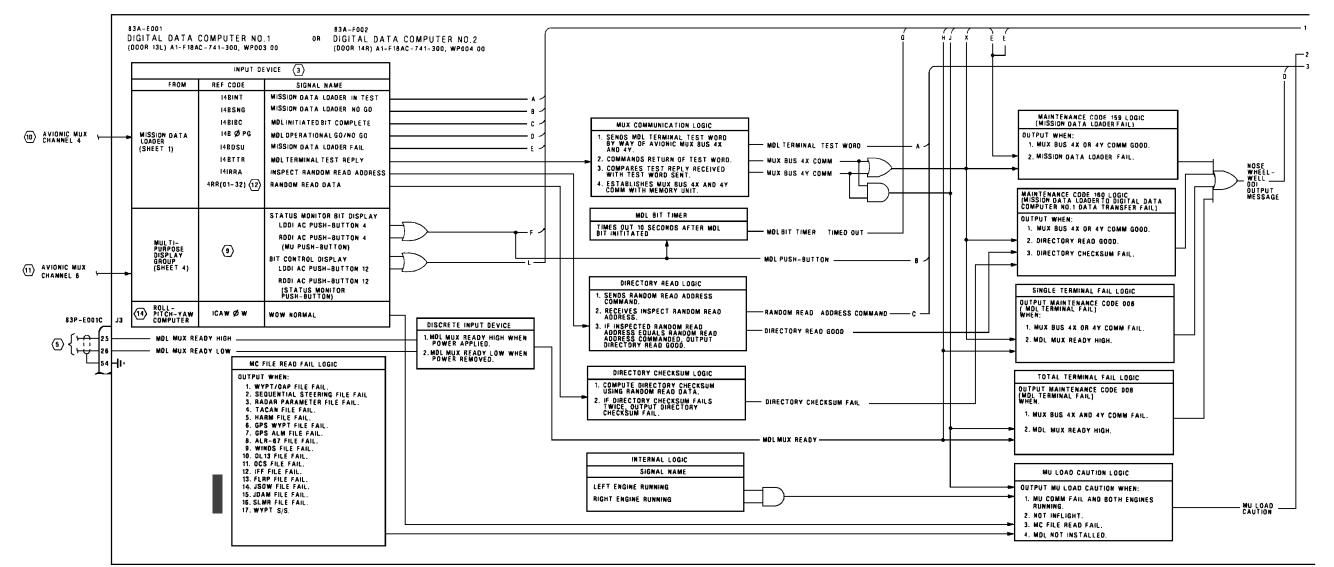


Figure 1.

Figure 1. Mission Data Loader Built-In Test Schematic (Sheet 1)



02200102

Figure 1. Mission Data Loader Built-In Test Schematic (Sheet 2)

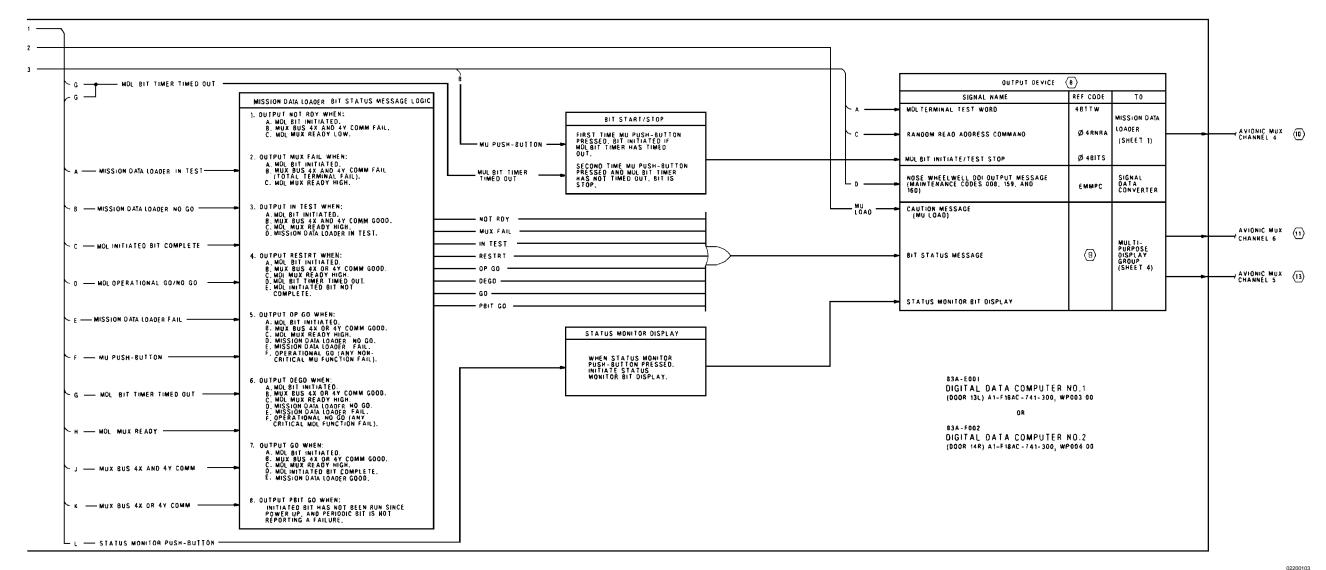


Figure 1.

Figure 1. Mission Data Loader Built-In Test Schematic (Sheet 3)

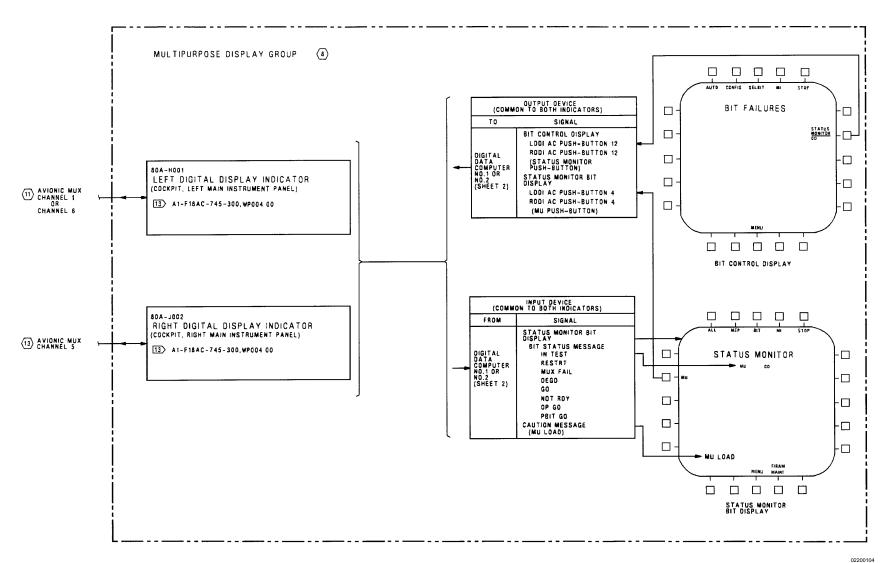
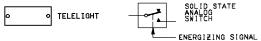


Figure 1.

LEGEND

- I. CONTINUITY TESTS:
 - ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN AI-F18A()-WDM-000.
 - WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY 6) LIS REMOVED FOR THOUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALATION. DO NOT REPLACE LOW LEVEL CYRRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUING THROUGH SWITCHES/RELAY CONTACTS MAY USE THE TRX SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (I) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY 2). MAKE SURE MULTIMETER LEADY JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. NONSTANDARD SYMBOLS

TIDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 1.



☑ IDENTIFIES 24YDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE I.

- FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO AI-FIBA()-OLD-000. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO AI-FIBAC-FIM-100.
- EXPLANATION OF MATRIX
 - A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
 - B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
 - C. SIGNAL OUTPUT IS READ HORIZONTALLY, EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
 - D. INTERPRET MATRIX TABLE AS INDICATED.
 - (I) ONE (I) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
 - (3) DASH (-) INDICATES THE DUTPUT DOES NOT DEPEND ON THIS INPUT.

- POWER SCHEMATIC, WP005 00.
- AVIONIC MUX CHANNEL | SCHEMATIC, AI-FIBAC-741-500, WP004 00.
- AVIONIC MUX CHANNEL 2 SCHEMATIC, AI-FIBAC-741-500, WP005 00.
- @ (T) (B) THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY HOLGATOR IP-1317(). HIGH THE DISPLAY INDICATOR IP-1317(). HEAD-UP-DISPLAY UNIT ANYANG-28, HORIZONTAL INDICATOR IP-1350A. FOR MULTIPURPOSE DISPLAY GROUP, REFER TO AI-FIBAC-745-500.
- DISPLAY REF CODES TARE NOT SHOWN IT DISPLAY MALFUNCTION EXISTS TRANSFER DISCLAY TO ANOTHER INDICATOR IF MALFUNCTION EXISTS ON MORE THAN ONE INDICATOR TROUBLESHOOT USING AI-FLBA()-OLD-SON INDICATOR, TROUBLESHOOT BY ON ONE INDICATOR, TROUBLESHOOT BY DOING DISPLAY TEST, AI-FLBAC-745-200, MP604 00.
- AVIONIC MUX CHANNEL 4 SCHEMATIC, AI-FIBAC-741-500, WP017 00.
- (II) AVIONIC MUX CHANNEL 6 SCHEMATIC, AI-FIBAC-741-500, WP019 00.
- MISSION DATA LOADER FUNCTIONAL SCHEMATIC, WP004 DI.
- (13) AVIONIC MUX CHANNEL 5 SCHEMATIC, AI-FIBAC-741-500, WP018 00.
- CROSS CHANNEL/MUX BUS/DISPLAYS FUNCTIONAL SCHEMATIC AI-FI8AC-570-500, WP02101.